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# THE GODDARD GENERAL ORBIT

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MAY 1968



GODDARD SPACE FLIGHT CENTER
GREENBELT, MARYLAND

# THE GODDARD GENERAL ORBIT DETERMINATION SYSTEM

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#### THE GODDARD GENERAL ORBIT DETERMINATION SYSTEM

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#### ABSTRACT

The General Orbit Determination System currently being used for Orbit Determination at the Goddard Space Flight Center has evolved over a period of approximately a decade.

It was contemplated, when the system was designed, that it would come to be used to determine orbits of satellites of various kinds, tracked by different types of tracking systems, and moving in many sorts of orbits which would be perturbed in a variety of ways by an atmosphere having variable characteristics and by a gravitational field whose higher harmonics would come into play, as well as by the moon, the sun, and radiation pressure. As the needs arose and the occasions demanded, additional orbit determination capabilities to meet the new circumstances were incorporated into the system in accordance with the basic plans. During this time the program was extended to accept new data types and, at various times, additional sets of unknowns were incorporated into it. The program was also equipped to use several general perturbation and special perturbation orbit theories and has been adapted to operate on various computers.

The purpose of this report is to set forth the formulations used in the system. In addition, flowcharts and program listings are also included.

#### THE GODDARD GENERAL ORBIT DETERMINATION SYSTEM

#### I. INTRODUCTION

The Goddard General Orbit Determination System accepts as input a set of orbital parameters at a specified epoch, some of which are to be corrected, and a set of observations of a satellite. The output of the system includes a set of corrected parameters obtained by means of a weighted least squares procedure, associated statistical information, and a satellite ephemeris. The differential correction portion of the system, which is the principal subject of this report, consists of nine basic section. These are:

- 1. Conversion of Elements, (CE)
- 2. Position of Space Vehicle, (P)
- 3. Local Station Predictions, (LSP)
- 4. Position in Ellipse, (PE)
- 5. Position Partial Derivatives, (PP)
- 6. Observation Partial Derivatives, (OP)
- 7. Equations of Condition, (EQC)
- 8. Least Squares Solution, (LSQ)
- 9. Conversion of Corrections, (CC)

In the next several sections of this report a description of each of these nine sections of the Differential Correction System, together with the mathematical formulations, will be given.

This report contains a description and documentation of one version of the Goddard General Orbit Determination System. It records the theoretical formulations most of which, aside from the indicated material obtained from references 1 and 3, were derived at the end of the last decade by J. W. Siry, who also wrote the accompanying discussion of Section 9 of Chapter III for this report. The original formulations had been prepared in a form which included condensed, general notation and minimal discussion needed to serve as the basis for programming. J. P. Murphy wrote expanded versions of formulas for certain specific cases of interest, added discussions, and carried through preparations for publication. In addition, flow charts and program listings of the version in current use were compiled for this publication by I. J. Cole.

The original flow diagram of this system is given in Figure 1.

#### II. NOTATION

The principal symbols employed in this report and their meanings are given in the following list:

- a, semi-major axis of the satellite orbit
- b, semi-minor axis of the satellite orbit

- E, eccentric anomaly
- e, eccentricity of the satellite orbit
- f, dynamical flattening of earth
- f, true anomaly
- G, universal gravitational constant
- $\underline{g}_1$  or  $\underline{\ell}$ , the local topocentric east vector
- $\underline{g}_2$  or  $\underline{m}$ , the local topocentric north vector
- $\underline{\mathtt{g}}_3$  or  $\underline{\mathtt{n}}$  , the local topocentric vertical vector
- $g_4 \text{ or } \underline{h}, \underline{\ell} \times \underline{k}$
- $\{\underline{g}_5, \ \underline{g}_6, \ \underline{g}_7\}, \{\underline{i}, \ \underline{j}, \ \underline{k}\}$
- $\{g_8, \ldots, g_{14}\}, \{\dot{g}_1, \ldots, \dot{g}_7\}$
- g, acceleration of gravity
- g, argument of perigee
- h, local hour angle
- h, right ascension of the ascending node
- I, inclincation of orbit plane to earth's equator
- i, (1,0,0)
- $J_1 = \frac{3}{2}J_2$
- $\boldsymbol{J}_{\boldsymbol{n}}\!,\! \boldsymbol{coefficient}$  of the  $\boldsymbol{n}^{\text{th}}$  zonal harmonic
- j, (0, 1, 0)
- k, (0, 0, 1)
- $\ell$ ,m, direction cosines relative to the local topocentric east-west and north-south directions, respectively
  - $\ell$ , mean anomaly
  - $\underline{\ell}$ , local topocentric east vector

- M, mean anomaly
- M, mass of the earth
- m, local topocentric north vector
- n, mean motion
- $^{n}_{p,\,q}$ , coefficients multiplying the  $p^{\,th}$  power term in the time polynominal for the value of the mean anomaly associated with the  $q^{th}$  segment of the observational arc.
  - n, local topocentric vertical vector
  - P, period of the satellite orbit
  - P<sub>n</sub>, Legendre polynomial of degree n
  - p, semi-latus rectum,  $a(1-e^2)$ .
  - p, vector directed to perigee
  - q, perigee distance
  - $\underline{\mathbf{q}}$ ,  $\underline{\alpha} \times \mathbf{p}$
  - R, mean equatorial radius of the earth
  - $\underline{R}_{i}$ , station position vector of the i<sup>th</sup> station
  - r, magnitude of radius vector of the satellite
  - r, radius vector of the satellite
  - r<sub>zi</sub>, radial distance from axis of rotation of earth to i<sup>th</sup> station
- r<sub>zsi</sub>, radial distance from axis of rotation of earth to surface point (on reference ellipsoid) directly below or above the i<sup>th</sup> station
  - t, time
  - U, potential function of the earth
  - v, magnitude of velocity vector of the satellite

- $\underline{\mathbf{v}}$ , the velocity vector of the satellite
- x, , eerrections to the unknowns
- $y_R$ , the observable quantities
- $\mathbf{z_{s\,i}}$ , distance from surface point associated with the i<sup>th</sup> station to the earth's equatorial plane
- $\alpha$ , right ascension
- $\underline{\alpha}$ , unit vector in direction of angular momentum
- $\beta$ , latitude
- $\beta$ , unit vector in a specific direction in the orbit plane
- $\underline{\gamma}$ ,  $(\underline{\alpha} \times \underline{\beta})$
- $\delta$ , declination
- $\delta_1$ , angle which the unit vector  $\underline{\beta}$  makes with the nodal ray
- $\theta$ , topocentric azimuth angle measured from the north
- $\theta_{\ell}$ , local hour angle
- $\theta_1$ , antenna x-angle
- $\theta_2$ , antenna y-angle
- $\lambda_{\mathbf{G}}$ , right ascension of Greenwich
- $\lambda_{Gi}$ . East longitude of i<sup>th</sup> station in system referred to Greenwich.
- $\lambda_i$ , inertial longitude of the  $i^{th}$  station or local sidereal time.
- $\mu$ ,  $\sqrt{\text{GM}}$
- $\nu$ , true anomaly
- $\rho$ , range
- $\phi$ , argument of latitude

- $\phi_{i}$ , topocentric elevation angle
- $\phi_{di}$ , geodetic latitude of the i<sup>th</sup> station
- $\Omega$ , longitude of the ascending node
- $\Omega$ , unit vector directed toward the ascending node
- $\omega$ , argument of perigee
- $\omega_{\mathbf{r}}$ , speed of rotation of the earth

Generally the indices and special symbols have the following meanings

- c, index to denote computed value
- e, index to denote earth
- i, index to denote ith tracking station
- j, index to denote j th unknown
- k, index to denote kth observation type
- n, index to denote nth observation time
- o, index to denote "observed" value
- o, index to denote initial value
- ', indicates long-period parameters in the Brouwer Theory
- ", indicates parameters in the Brouwer Theory which include short-period terms
- ., indicates total differentiation with respect to time
- \*, indicates unit vector (e.g.  $\underline{a}^* = \underline{a}/|\underline{a}|$ )

In certain cases, a symbol is associated with more than one definition. In each case in which such a symbol is used, the meaning will be clear from the context.

Several of the symbols in the above list are illustrated in Figure 2.

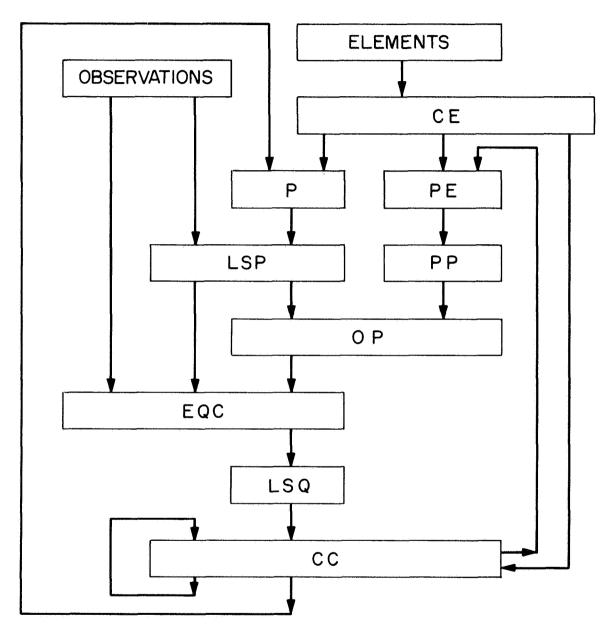


Figure 1-Differential Correction Program Flow Diagram

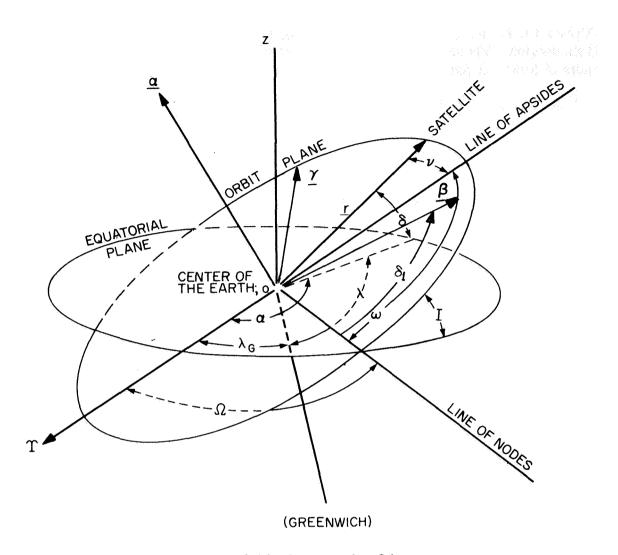


Figure 2-The Geometry of an Orbit

# III. PRINCIPAL SECTIONS OF THE SYSTEM

#### 1. Conversion of Elements

In this portion of the system a certain set of orbital elements or equivalent parameters at a particular epoch are provided as input, and two new sets of similar constants are computed. Specifically, one of the sets

$$\begin{split} & \left\{ \mathbf{x}(\mathsf{t}_0), \; \mathbf{y}(\mathsf{t}_0), \; \mathbf{z}(\mathsf{t}_0), \; \dot{\mathbf{x}}(\mathsf{t}_0) \; \dot{\mathbf{y}}(\mathsf{t}_0) \; \dot{\mathbf{z}}(\mathsf{t}_0) \right\}, \\ & \left\{ \mathbf{a}, \mathbf{e}, \; \dot{\mathbf{i}}, \Omega, \phi(\mathsf{t}_0), \nu(\mathsf{t}_0) \right\} \; \; \text{or} \; \; \left\{ \mathbf{a}, \mathbf{e}, \dot{\mathbf{i}}, \Omega, \omega, \mathsf{M}(\mathsf{t}_0) \right\} \end{split}$$

serves as input and the other two sets are computed in this section.

Values for the additional parameters, p, q, b, n, and P, are also computed in this section. These calculations are performed through the use of the following sets of formulas and others found on page 16 of this report.

A. Given 
$$\underline{r}(t_0), \underline{v}(t_0)$$

$$r_0 = |\underline{r}(t_0)|$$

$$v_0 = |\underline{v}(t_0)|$$

$$\theta_0 = -\arcsin\left[\underline{r}^*(t_0)\cdot\underline{v}^*(t_0)\right]$$

$$\underline{\alpha}(t_0) = \left[\underline{r}(t_0)\times\underline{v}(t_0)\right]^*$$

$$\underline{\alpha}(t_0) = \left[\underline{k}\times\underline{\alpha}(t_0)\right]^*$$

$$\underline{i} = \arccos\left[\underline{\alpha}(t_0)\cdot\underline{k}\right]$$

$$\Omega(t_0) = \arccos\left[\underline{\Omega}(t_0)\cdot\underline{i}\right]$$

$$\phi(t_0) = \arccos\left[\underline{r}^*(t_0)\cdot\underline{\Omega}(t_0)\right]$$

$$\omega(t_0) = \phi(t_0) - \nu(t_0)$$

$$\mathbf{r_0}$$
,  $\mathbf{v_0}$ ,  $\theta_0$ 

$$a = \frac{\mu^{2} r_{0}}{2\mu^{2} - r_{0} v_{0}^{2}}$$

$$e = \sqrt{1 - \frac{(r_{0} v_{0} \cos \theta_{0})^{2}}{a\mu^{2}}}$$

$$v_{0} = \arccos \left[\frac{a(1 - e^{2}) - r_{0}}{e r_{0}}\right]$$
(2)

C. Given a, e,  $\nu_0$ 

$$r_0 = \frac{a(1-e^2)}{1+e\cos \nu_0}$$

$$\mathbf{v_0} = \mu \sqrt{\frac{2}{\mathbf{r_0}} - \frac{1}{\mathbf{a}}}$$

$$\theta_0 = \arccos \frac{\mu \sqrt{a(1-e^2)}}{r_0 v_0}$$

$$E(t) = \operatorname{arc} \cos \left[ \frac{\cos \nu(t) + e}{1 - e \cos \nu(t)} \right] = \operatorname{arc} \sin \left[ \frac{\sqrt{1 - e^2} \sin \nu(t)}{1 + e \cos \nu(t)} \right]$$

$$\nu(t) = \arccos\left[\frac{\cos E(t) - e}{1 - e \cos E(t)}\right] = \arcsin\left[\frac{\sqrt{1 - e^2} \sin E(t)}{1 - e \cos E(t)}\right]$$

$$M(t) = E(t) - e \sin E(t) = M(t_0) + n(t-t_0)$$

# D. Additional Parameters

$$p = a(1-e^2)$$
 $q = a(1-e)$ 
 $b = a \sqrt{1-e^2}$ 
 $n = \frac{\mu}{a^{3/2}}$ 
 $P = \frac{2\pi}{a}$ 
(4)

(3)

# 2. Position of Space Vehicle

In this portion of the system the position of a space vehicle is computed by means of a special perturbation method, such as a Cowell integration method,

or by means of a general perturbation method, such as the one associated with the Brouwer theory. As an illustration of the function of this portion of the system we shall consider the case of an ephemeris of a satellite generated by means of the Brouwer theory (see References 1 and 2). The potential function\* for this theory is

$$U = \frac{\mu^2}{r} \left[ 1 - \sum_{n=2}^{5} \frac{J_n R_e^n}{r^n} P_n \left( \sin \beta \right) \right]$$
 (5)

The position and velocity of a spacecraft at any time, t, are found by substituting the time into Equations (6) through (10). The fundamental plane is the equator with the x-axis directed toward the vernal equinox. In this theory an initial set of elements a'', e'', I'',  $\ell_0''$ ,  $g_0''$ , and  $h_0''$  at an epoch time,  $t_0$ , are estimated or obtained from the previous iteration of the differential correction process. These parameters are the constants of integration of the theory and are related to the osculating Keplerian elements  $\{a, e, I, \ell, g, h\}$  through the following formulas found in References 1 and 2. The secular terms are

$$\ell'' = \ell_0'' + n_0 t \left\{ 1 - \frac{3}{4} \frac{J_2 R_e^2}{a''^2 \eta^3} (1 - 3\theta^2) + \frac{3}{128} \frac{J_2^2 R_e^4}{a''^4 \eta^7} \right\}$$

$$\cdot \left[ -15 + 16 \eta + 25 \eta^2 + (30 - 96 \eta - 90 \eta^2) \theta^2 + (105 + 144 \eta + 25 \eta^2) \theta^4 \right]$$

$$- \frac{45}{128} \frac{J_4 R_e^4}{a''^4 \eta^7} e^{i'^2} \left[ 3 - 30 \theta^2 + 35 \theta^4 \right]$$

$$= g''_0 + n_0 t \left\{ -\frac{3}{4} \frac{J_2 R_e^2}{a''^2 \eta^4} (1 - 5\theta^2) + \frac{3}{128} \frac{J_2^2 R_e^4}{a''^4 \eta^8} \right\}$$

$$\cdot \left[ -35 + 24 \eta + 25 \eta^2 + (90 - 192 \eta - 126 \eta^2) \theta^2 + (385 + 360 \eta + 45 \eta^2) \theta^4 \right]$$

$$- \frac{15}{128} \frac{J_4 R_e^4}{a''^4 \eta^8} \left[ 21 - 9 \eta^2 + (-270 + 126 \eta^2) \theta^2 + (385 - 189 \eta^2) \theta^4 \right]$$

<sup>\*</sup>Note that in Brouwer's original paper, the convention  $\mu = \mathsf{GM}_{\mathbf{e}}$  was employed.

$$\begin{split} h'' &= h''_{0} + n_{0}t \left\{ -\frac{3}{2} \frac{J_{2}R_{e}^{2}\theta}{a''^{2}\eta^{4}} + \frac{3}{32} \frac{J_{2}^{2}R_{e}^{4}}{a''^{4}\eta^{8}} \left[ (-5 + 12\eta + 9\eta^{2})\theta \right. \right. \\ &\left. + (-35 - 36\eta - 5\eta^{2})\theta^{3} \right] \left. - \frac{15}{32} \frac{J_{4}R_{e}^{4}}{a''^{4}\eta^{8}} (5 - 3\eta^{2})\theta \left. (3 - 7\theta^{2}) \right\} \,, \end{split}$$

where, 
$$\theta = \cos I''$$
,  $\eta = \sqrt{1 - e''^2}$ , and  $n_0 = \mu / \sqrt{a''^3}$ .

In cases in which the drag perturbation is not too large, it is convenient to represent its principal effect by means of quadratic and cubic terms in the mean anomaly. The coefficients  $n_{pq}$ , where p=2 or 3, refer to terms of the quadratic and cubic type respectively. Since a single pair of constants might not be accurate enough over the entire observational arc, up to twenty parameters may be employed corresponding to the epochs,  $t_q$ , where  $q=0,1,\cdots,19$ , at which the intervals of interest begin. If the arc is not subdivided, the subscript q=0 is understood but not always written.

The long-period terms for e and I are

$$\begin{split} \delta_1 e &= \left\{ \frac{1}{16} \, \frac{J_2 R_e^2 e''}{a''^2 \eta^2} \left[ 1 - 11 \theta^2 - 40 \theta^4 \, \left( 1 - 5 \theta^2 \right)^{-1} \right] \, + \\ &+ \frac{5}{16} \, \frac{J_4 R_e^2 e''}{J_2 a''^2 \eta^2} \, \left[ 1 - 3 \theta^2 - 8 \theta^4 \, \left( 1 - 5 \theta^2 \right)^{-2} \right] \right\} \cos 2 g'' \\ &+ \left\{ \frac{-J_3 R_e}{2 J_2 a''} \sin I'' - \frac{5}{32} \, \frac{J_5 R_e^3}{J_2 a''^3 \eta^4} \sin I'' \, \left( 4 + 3 e''^2 \right) \right. \\ &\cdot \left[ 1 - 9 \theta^2 - 24 \theta^4 \, \left( 1 - 5 \theta^2 \right)^{-1} \right] \right\} \sin g'' + \frac{35}{192} \, \frac{J_5 R_e^3 e''^2}{J_2 a''^3 \eta^4} \sin I' \\ &\cdot \left[ 1 - 5 \theta^2 - 16 \theta^4 \, \left( 1 - 5 \theta^2 \right)^{-1} \right] \sin 3 g'' \\ &\delta_1 I = - \, \frac{e'' \, \delta_1 e}{\eta^2 \tan I''} \end{split}$$

The secular and long-period terms for ℓ, g, and h, are, respectively,

$$\begin{split} \ell' &= \ell'' + \left\{ \frac{J_2 R_e^2}{16 a''^2 \eta} \left[ 1 - 11 \theta^2 - 40 \theta^4 (1 - 5 \theta^2)^{-1} \right] + \frac{5J_4 R_e^2}{16J_2 a''^4 \eta} \right. \\ &\cdot \left[ 1 - 3 \theta^2 - 8 \theta^4 \left( 1 - 5 \theta^2 \right)^{-1} \right] \\ &+ \left\{ \frac{J_3 R_e \eta}{2J_2 a e''} \sin I'' + \frac{5J_5 R_e^3}{32J_2 a^3 \eta^3 e''} \sin I'' \left( 4 + 9 e''^2 \right) \cdot \right. \\ &\cdot \left[ 1 - 9 \theta^2 - 24 \theta^4 \left( 1 - 5 \theta^2 \right)^{-1} \right] \right\} \cos g'' - \frac{35}{192} \frac{J_5 R_e^3 e''}{J_2 a''^3 \eta^3} \sin I'' \cdot \\ &\cdot \left[ 1 - 5 \theta^2 - 16 \theta^4 \left( 1 - 5 \theta^2 \right)^{-1} \right] \cos 3 g'' \\ g' &= g'' + \left\{ \frac{-J_2 R_e^2}{32 a''^2 \eta^4} \left[ (2 + e''^2) - 11 \left( 2 + 3 e''^2 \right) \theta^2 - 40 \left( 2 + 5 e''^2 \right) \cdot \right. \\ &\cdot \theta^4 \left( 1 - 5 \theta^2 \right)^{-1} - 400 e''^2 \theta^6 \left( 1 - 5 \theta^2 \right)^{-2} \right] \\ &- \frac{5J_4 R_e^2}{32J_2 a''^2 \eta^4} \left[ 2 + e''^2 - 3 \left( 2 + 3 e''^2 \right) \theta^2 - 8 \left( 2 + 5 e''^2 \right) \cdot \right. \\ &\cdot \theta^4 \left( 1 - 5 \theta^2 \right)^{-1} - 80 e''^2 \theta^6 \left( 1 - 5 \theta^2 \right)^{-2} \right] \right\} \sin 2 g'' \\ &+ \left\{ \frac{J_3 R_e}{2J_2 a'' \eta^3} \left( \frac{\sin I''}{e''} - \frac{e'' \theta^2}{\sin I''} \right) - \frac{5J_5 R_e^3}{32J_2 a''^3 \eta^6} \cdot \right. \\ &\cdot \left[ \left( \frac{\eta^2 \sin I''}{e''} - \frac{e'' \theta^2}{\sin I''} \right) \left( 4 + 3 e''^2 \right) + e'' \sin I'' \left( 26 + 9 e''^2 \right) \right] \cdot \right. \end{split}$$

$$\begin{split} & \cdot \left[1 - 5\theta^2 - 16\theta^4 \left(1 - 5\theta^2\right)^{-1}\right] + \frac{15J_5R_6^3\theta^3e''}{16J_2a''^3\eta^6} \sin I'' \left(4 + 3e''^2\right) \cdot \\ & \cdot \left[3 + 16\theta^2 \left(1 - 5\theta^2\right)^{-1} + 40\theta^4 \left(1 - 5\theta^2\right)^{-2}\right] \right\} \cos g'' \\ & + \left\{ \frac{35J_5R_6^3}{576J_2a''^3\eta^6} \left[ e'' \sin I'' \left(3 + 2e''^2\right) - \frac{e''^3\theta^2}{\sin I''} \right] \left[1 - 5\theta^2 - 16\theta^4 \left(1 - 5\theta^2\right)^{-1}\right] \right. \\ & - \frac{35J_5e^3\theta^2R_6^3}{288J_2a''^3\eta^6} \sin I'' \left[5 + 32\theta^2 \left(1 - 5\theta^2\right)^{-1} + 80\theta^4 \left(1 - 5\theta^2\right)^{-2}\right] \cos 3g'' \\ & h' = h'' + \left\{ -\frac{J_2R_6^2e''^2\theta}{16a''^2\eta^4} \left[11 + 80\theta^2 \left(1 - 5\theta^2\right)^{-1} + 200\theta^4 \left(1 - 5\theta^2\right)^{-2}\right] \right. \\ & \left. - \frac{5J_4e'''^2\theta R_6^2}{16J_2a''^2\eta^4} \left[3 + 16\theta^2 \left(1 - 5\theta^2\right)^{-1} \right. \\ & \left. + 40\theta^4 \left(1 - 5\theta^2\right)^{-2}\right] \right\} \sin 2g'' + \left\{ -\frac{J_3e'''\theta R_6}{2J_2a'''\eta^2\sin I''} \right. \\ & \left. - \frac{5J_5e'''\theta R_6^3}{32J_2a''^3\eta^6\sin I''} \left. \left(4 + 3e''^2\right) \left[1 - 9\theta^2 - 24\theta^4 \left(1 - 5\theta^2\right)^{-1}\right] \right. \\ & \left. - \frac{15J_5e'''\theta R_6^3}{16J_2a''^3\eta^6} \sin I'' \left(4 + 3e''^2\right) \left[3 + 16\theta^2 \left(1 - 5\theta^2\right)^{-1} + 40\theta^4 \left(1 + 5\theta^2\right)^{-2}\right] \right\} \cos g'' \\ & + \left\{ \frac{35}{576} \frac{J_5e'''^3\theta R_6^3}{J_2a''^3\eta^6} \sin I'' \left[1 - 5\theta^2 - 16\theta^4 \left(1 - 5\theta^2\right)^{-1}\right] \right. \\ & \left. + \frac{35}{288} \frac{J_5e'^3\theta R_6^3}{J_2a''^3\eta^6} \sin I'' \left[5 + 32\theta^2 \left(1 - 5\theta^2\right)^{-1} + 80\theta^4 \left(1 - 5\theta^2\right)^{-2}\right] \right\} \cos 3g'' \cdot \right. \end{aligned}$$

After adding the short-period terms, the osculating elements are obtained from

$$a = a'' \left\{ 1 + \frac{J_2 R_e^2}{2a''^2} \left[ - (1 - 3\theta^2) \left( \frac{a''^3}{r'^3} - \eta^{-3} \right) \right. \right.$$

$$+ 3 \left( 1 - \theta^2 \right) \frac{a''^3}{r'^3} \cos \left( 2g' + 2f' \right) \right] \right\}$$

$$e = e'' + \delta_1 e + \frac{\eta^2}{2e''} \left\{ \frac{J_2 R_e^2}{2a''^2} \left[ - (1 - 3\theta^2) \left( \frac{a''^3}{r'^3} - \eta^{-3} \right) \right. \right.$$

$$+ 3 \left( 1 - \theta^2 \right) \left( \frac{a''^3}{r'^3} - \eta^{-4} \right) \cos \left( 2g' + 2f' \right) \right]$$

$$- \frac{J_2 R_e^2}{2a''^2 \eta^4} \left( 1 - \theta^2 \right) \left[ 3e'' \cos \left( 2g' + f' \right) \right. \right.$$

$$+ e'' \cos \left( 2g' + 3f' \right) \right] \right\}$$

$$I = I'' + \delta_1 I + \frac{J_2 \theta R_e^2}{4a''^2 \eta^4} \sqrt{1 - \theta^2} \left[ 3\cos \left( 2g' + 2f' \right) \right. \right.$$

$$+ 3e''^2 \cos \left( 2g' + f' \right) + e'' \cos \left( 2g' + 3f' \right) \right]$$

$$\mathcal{E} = \ell' - \frac{J_2 R_e^2}{8e'' a''^2 \eta} \left\{ - 2 \left( 1 - 3\theta^2 \right) \left( \frac{a''^2}{r'^2} \eta^2 + \frac{a''}{r'} + 1 \right) \sin f' \right.$$

$$+ 3 \left( 1 - \theta^2 \right) \left[ - \left( \frac{a''^2}{r'^2} \eta^2 + \frac{a''}{r'} - 1 \right) \sin \left( 2g' + f' \right) \right.$$

$$+ \left. \left( \frac{a''^2}{r'^2} \eta^2 + \frac{a''}{r'} + \frac{1}{3} \right) \sin \left( 2g' + 3f' \right) \right] \right\}$$

$$g = g' + \frac{J_2 R_e^2}{8e''a''^2 \eta^2} \left\{ -2(1 - 3\theta^2) \left( \frac{a''^2}{r'^2} \eta^2 + \frac{a''}{r'} + 1 \right) \sin f' \right.$$

$$+ 3(1 - \theta^2) \left[ -\left( \frac{a''^2}{r'^2} \eta^2 + \frac{a''}{r'} - 1 \right) \sin (2g' + f') \right.$$

$$+ \left( \frac{a''^2}{r'^2} \eta^2 + \frac{a''}{r'} + \frac{1}{3} \right) \sin (2g' + 3f') \right] \right\}$$

$$+ \frac{J_2 R_e^2}{8a''^2 \eta^4} \left\{ -6(1 - 5\theta^2) (f' + \ell' - e'' \sin f') \right.$$

$$+ (3 - 5\theta^2) \left[ 3 \sin (2g' + 2f') \right.$$

$$+ 3e'' \sin (2g' + f') + e'' \sin (2g' + 3f') \right] \right\}$$

$$h = h' - \frac{J_2 \theta R_e^2}{4a''^2 \eta^4} \left[ 6(f' - \ell' + e'' \sin f') \right.$$

$$- 3 \sin (2g' + 2f') - 3e'' \sin (2g' + f') - e'' \sin (2g' + 3f') \right]$$

where f' and r' are computed from

 $E' - e'' \sin E' = \ell'$ 

$$\tan \frac{1}{2} f' = \sqrt{\frac{1+e''}{1-e''}} \tan \frac{1}{2} E'$$

$$\frac{a''}{r'} = (1+e'' \cos f')/(1-e''^2)$$
(8)

The position and velocity vectors in rectangular coordinates may then be obtained from

$$\begin{pmatrix} x \\ y \\ z \end{pmatrix} = \begin{pmatrix} A_x B_x 0 \\ A_y B_y 0 \\ A_z B_z 0 \end{pmatrix} \begin{pmatrix} \cos E - e \\ \sin E \\ 0 \end{pmatrix} = \underline{r}_c (t)$$

and (9)

$$\begin{pmatrix} \dot{x} \\ \dot{y} \\ \dot{z} \end{pmatrix} = \frac{a}{r} n \begin{pmatrix} A_x B_x 0 \\ A_y B_y 0 \\ A_z B_z 0 \end{pmatrix} \begin{pmatrix} -\sin E \\ \cos E \\ 0 \end{pmatrix} = \frac{\dot{r}}{c} (t)$$

where

$$E - e \sin E = \ell$$

and where

$$A_{x} = a (\cos g \cos h - \sin g \sin h \cos I)$$

$$A_{y} = a (\sin g \cos h \cos I + \cos g \sin h)$$

$$A_{z} = a \sin I \sin g$$

$$B_{x} = -a \sqrt{1 - e^{2}} (\sin g \cos h + \cos g \sin h \cos I)$$

$$B_{y} = a \sqrt{1 - e^{2}} (\cos g \cos h \cos I - \sin g \sin h)$$

$$B_{z} = a \sqrt{1 - e^{2}} \sin I \cos g$$
(10)

In the differential correction of the orbit, the constants of integration a", e", I",  $\ell_0$ ",  $g_0$ ", and  $h_0$ " are determined. One or more drag parameters are also solved for if this is appropriate.

#### 3. Local Station Predictions

In this portion of the system computed values for the various observations at the times of observation are obtained. An actual observable at time,  $t_n$ , taken at the i <sup>th</sup> station and of the k <sup>th</sup> type, is denoted by the symbol  $y_{0,k,i}$  ( $t_n$ ), while the computed value of this observable is denoted by  $y_{c,k,i}$  ( $t_n$ ). In Table 1 there appears a list of the various observation types, the symbols used to designate them, and the value of the designator k for each type.

Table 1
Observation Types - y<sub>k</sub>

k	Symbol	Name
1	ρ	range
2	l	east-west direction cosine
3	m	north-south direction cosine
4	$\theta$	azimuth
5	$\phi$	elevation
6	α	right ascension
7	δ	declination
8	h	local hour angle
9	$\dot{ ho}$	range rate
10	į	east-west direction cosine rate
11	m	north-south direction cosine rate
12	$\overset{ extbf{m}}{ heta}$	azimuth rate
13	$\dot{\phi}$	elevation rate
14	å	right ascension rate
15	α δ	declination rate
16	ĥ	local hour angle rate
17	$\theta_{1}$	antenna x-angle
18	$\theta_{2}^{1}$	antenna y-angle

Formulas for obtaining computed values of the quantities appearing in Table 1 will now be developed. Before these formulas are obtained, it is convenient to give expressions for the following quantities.

- $\underline{\mathbf{R}}_{i}$  the station position vector
- $\underline{\ell}_i$  the local topocentric east vector
- $\underline{m}_i$  the local topocentric north vector
- n; the local topocentric vertical vector
- $\underline{\mathbf{h}}_{\mathbf{i}} \quad \underline{\mathbf{\ell}}_{\mathbf{i}} \times \underline{\mathbf{k}}$

and their rates of change as functions of the coordinates of the i <sup>th</sup> tracking station. The inertial longitude or local sidereal time,  $\lambda_i$  (t), of the i <sup>th</sup> station at time t is given by

$$\lambda_{i}(t) = \lambda_{Gi} + \lambda_{G}(t_{0}) + \omega_{o}(t - t_{0})$$
(11)

where  $\lambda_{Gi}$ ,  $\lambda_{G}$  (t<sub>0</sub>), and  $\omega_{e}$  denote, respectively, the east longitude of the i <sup>th</sup> station in the system referred to Greenwich, the right ascension of the Greenwich meridian at t<sub>0</sub>, and the earth's angular velocity. Then,

$$\underline{\ell}_{i}(t) = -\sin \lambda_{i}(t) \underline{i} + \cos \lambda_{i}(t) \underline{j}$$

$$\underline{m}_{i}(t) = -\sin \phi_{di} \cos \lambda_{i}(t) \underline{i} - \sin \phi_{di} \sin \lambda_{i}(t) \underline{j} + \cos \phi_{di} \underline{k}$$

$$\underline{n}_{i}(t) = \cos \phi_{di} \cos \lambda_{i}(t) \underline{i} + \cos \phi_{di} \sin \lambda_{i}(t) \underline{j} + \sin \phi_{di} \underline{k}$$

$$\underline{h}_{i}(t) = \cos \lambda_{i}(t) \underline{i} + \sin \lambda_{i}(t) \underline{j}$$

$$\underline{\ell}_{i}(t) = -\omega_{e} \underline{h}_{i}(t)$$

$$\underline{\dot{m}}_{i}(t) = -\omega_{e} \sin \phi_{di} \underline{\ell}_{i}(t)$$

$$\underline{\dot{n}}_{i}(t) = \omega_{e} \cos \phi_{di} \underline{\ell}_{i}(t)$$

$$\underline{\dot{n}}_{i}(t) = \omega_{e} \cos \phi_{di} \underline{\ell}_{i}(t)$$

$$\underline{\dot{n}}_{i}(t) = \omega_{e} \frac{\ell}{\ell}_{i}(t)$$

where  $\phi_{\rm di}$  denotes the geodetic latitude of the i <sup>th</sup> station. Define p <sub>ei</sub> and  $\phi_{\rm ci}$  to be

$$\mathbf{p_{e\,i}}$$
 =  $(\mathbf{1}-\mathbf{f})^2$  tan  $\phi_{\mathrm{d\,i}}$  
$$\phi_{\mathrm{c\,i}} = \mathrm{arc\,tan\,p_{e\,i}}$$
 (13)

where f is the flattening. The radial distance  $r_{zsi}$  from the axis of rotation of the earth to the surface point on the reference ellipsoid below or above the i<sup>th</sup> station is given by

$$r_{zsi} = \frac{R_e (1-f)}{\sqrt{p_{ei}^2 + (1-f)^2}}$$
 (14)

where  $R_e$  is the mean equatorial radius of the earth. The radial distance  $r_{zi}$  from the axis of rotation of the earth to the i<sup>th</sup> station is given by

$$\mathbf{r}_{zi} = \mathbf{r}_{zsi} + \mathbf{H}_{i} \cos \phi_{di} \tag{15}$$

where  $H_i$  is the height of the i<sup>th</sup> station above the reference ellipsoid. The distance  $z_{si}$  from the i<sup>th</sup> station surface point on the reference ellipsoid below or above the station to the earth's equatorial plane is given by

$$z_{si} = p_{ei} r_{zsi} {.} {(16)}$$

The rectangular coordinates  $x_i$  (t),  $y_i$  (t),  $z_i$  of the i<sup>th</sup> station and their rates of change are given by

$$x_{i}(t) = r_{zi} \cos \lambda_{i}(t)$$

$$y_{i}(t) = r_{zi} \sin \lambda_{i}(t)$$

$$z_{i} = z_{si} + H_{i} \sin \phi_{di}$$

$$\dot{x}_{i}(t) = -\omega r_{zi} \sin \lambda_{i}(t)$$

$$\dot{y}_{i}(t) = \omega r_{zi} \cos \lambda_{i}(t)$$

$$\dot{z}_{i} = 0$$
(17)

Then,  $\underline{R}_{i}$  (t) and  $\underline{\dot{R}}_{i}$  (t) are given by

$$\underline{R}_{i}(t) = \mathbf{x}_{i}(t) \underline{\mathbf{i}} + \mathbf{y}_{i}(t) \underline{\mathbf{j}} + \mathbf{z}_{i} \underline{\mathbf{k}}$$

$$\underline{\dot{R}}_{i}(t) = \dot{\mathbf{x}}_{i}(t) \underline{\mathbf{i}} + \dot{\mathbf{y}}_{i}(t) \underline{\mathbf{j}}$$
(18)

The vector from the ith station to the space vehicle, and the rate of change, are given, respectively, by

$$\underline{\rho}_{i}(t) = \underline{r}_{c}(t) - \underline{R}_{i}(t)$$

$$\underline{\dot{\rho}}_{i}(t) = \underline{\dot{r}}_{c}(t) - \underline{\dot{R}}_{i}(t)$$
(19)

It is now possible to write explicit formulas for computed values of the eighteen types of observations listed in Table 1. The output of this portion of the system contains computed values of the observed quantities at each time, t<sub>n</sub>.

Formulas involving inverse tangent functions are quadrant-oriented, i.e., information needed to determine quadrants is contained within them.

$$\begin{aligned} &\mathbf{y}_{\mathbf{c},1,i} = \rho_{i} \; (\mathbf{t}) = \left| \underline{\rho}_{i} \; (\mathbf{t}) \right| \\ &\mathbf{y}_{\mathbf{c},2,i} = \mathcal{X}_{i} \; (\mathbf{t}) = \underline{\mathcal{X}}_{i} \; (\mathbf{t}) \cdot \underline{\rho}_{i}^{*} \; (\mathbf{t}) \\ &\mathbf{y}_{\mathbf{c},3,i} = \mathbf{m}_{i} \; (\mathbf{t}) = \underline{\mathbf{m}}_{i} \; (\mathbf{t}) \cdot \underline{\rho}_{i}^{*} \; (\mathbf{t}) \\ &\mathbf{y}_{\mathbf{c},4,i} = \theta_{i} \; (\mathbf{t}) = \arctan \; \left\{ \left[ \underline{\rho}_{i}^{*} \; (\mathbf{t}) \cdot \underline{\mathcal{X}}_{i} \; (\mathbf{t}) / \underline{\rho}_{i}^{*} \; (\mathbf{t}) \cdot \underline{\mathbf{m}}_{i} \; (\mathbf{t}) \right] \right\} \\ &\mathbf{y}_{\mathbf{c},5,i} = \phi_{i} \; (\mathbf{t}) = \arcsin \; \left[ \underline{\mathbf{n}}_{i} \; (\mathbf{t}) \cdot \underline{\rho}_{i}^{*} \; (\mathbf{t}) \right] \\ &\mathbf{y}_{\mathbf{c},6,i} = \alpha_{i} \; (\mathbf{t}) = \arctan \; \left\{ \left[ \underline{\rho}_{i}^{*} \; (\mathbf{t}) \cdot \underline{\mathbf{j}} / \underline{\rho}_{i}^{*} \; (\mathbf{t}) \cdot \underline{\mathbf{i}} \right] \right\} \\ &\mathbf{y}_{\mathbf{c},7,i} = \delta_{i} \; (\mathbf{t}) = \arctan \; \left\{ \left[ \underline{\rho}_{i}^{*} \; (\mathbf{t}) \cdot \underline{\mathcal{X}}_{i} / \underline{\rho}_{i}^{*} \; (\mathbf{t}) \cdot \underline{\mathbf{h}}_{i} \; (\mathbf{t}) \right] \right\} \\ &\mathbf{y}_{\mathbf{c},8,i} = \mathbf{h}_{i} \; (\mathbf{t}) = \arctan \; \left\{ \left[ \underline{\rho}_{i}^{*} \; (\mathbf{t}) \cdot \underline{\mathcal{X}}_{i} / \underline{\rho}_{i}^{*} \; (\mathbf{t}) \cdot \underline{\mathbf{h}}_{i} \; (\mathbf{t}) \right] \right\} \\ &\mathbf{y}_{\mathbf{c},9,i} = \dot{\rho}_{i} \; (\mathbf{t}) = \underline{\rho}_{i}^{*} \; (\mathbf{t}) \cdot \dot{\underline{\rho}}_{i} \; (\mathbf{t}) \end{aligned}$$

$$\mathbf{y}_{c,10,i} = \dot{\ell}_{i}(t) = \dot{\underline{\ell}}_{i}(t) \cdot \underline{\rho}_{i}^{*}(t) + \underline{\ell}_{i}(t) \cdot \frac{\mathbf{d}}{\mathbf{d}t} \underline{\rho}_{i}^{*}(t)$$

$$\mathbf{y}_{\mathbf{e},\mathbf{11},\mathbf{i}} = \dot{\mathbf{m}}_{\mathbf{i}}(\mathbf{t}) = \dot{\underline{\mathbf{m}}}_{\mathbf{i}}(\mathbf{t}) \cdot \underline{\dot{p}}_{\mathbf{i}}(\mathbf{t}) + \underline{\mathbf{m}}_{\mathbf{i}}(\mathbf{t}) \cdot \frac{\mathbf{d}}{\mathbf{d}\mathbf{t}} \underbrace{\dot{p}}_{\mathbf{i}}^{*}(\mathbf{t})$$

$$\mathbf{y}_{c,12,i} = \dot{\theta}_i(\mathbf{t}) =$$

$$\frac{\left\{ [\underline{\rho}_{i}^{*}(t) \cdot \underline{\ell}_{i}(t)] + \left[ \frac{d}{dt} \underline{\rho}_{i}^{*}(t) \cdot \underline{\ell}_{i}(t) \right] \right\} [\underline{\rho}_{i}^{*}(t) \cdot \underline{m}_{i}(t)] - \left\{ [\underline{\rho}_{i}^{*}(t) \cdot \underline{m}_{i}(t)] + \left[ \frac{d}{dt} \underline{\rho}_{i}^{*}(t) \cdot \underline{m}_{i}(t) \right] \right\} [\underline{\rho}_{i}^{*}(t) \cdot \underline{m}_{i}(t)]}{[\underline{\rho}_{i}^{*}(t) \cdot \underline{\ell}_{i}(t)]^{2} + [\underline{\rho}_{i}^{*}(t) \cdot \underline{m}_{i}(t)]^{2}}$$

$$\mathbf{y}_{\mathbf{c},\mathbf{13},\mathbf{i}} = \dot{\phi}_{\mathbf{i}}(\mathbf{t}) = \left\{ \left[ \frac{\mathrm{d}}{\mathrm{d}\mathbf{t}} \underline{\rho}_{\mathbf{i}}^{*}(\mathbf{t}) \cdot \underline{\mathbf{n}}_{\mathbf{i}}(\mathbf{t}) \right] + \left[ \underline{\rho}_{\mathbf{i}}^{*}(\mathbf{t}) \cdot \underline{\mathbf{n}}_{\mathbf{i}}(\mathbf{t}) \right] \right\} / \sqrt{1 - \left[ \underline{\rho}_{\mathbf{i}}^{*}(\mathbf{t}) \cdot \underline{\mathbf{n}}_{\mathbf{i}}(\mathbf{t}) \right]^{2}}$$

$$y_{c,14,i} = \dot{a}_{i}(t) =$$

$$\left\{ \left[ \frac{d}{dt} \underline{\rho}_{i}^{*}(t) \cdot \underline{\mathbf{j}} \right] \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\mathbf{i}} \right] - \left[ \frac{d}{dt} \underline{\rho}_{i}^{*}(t) \cdot \underline{\mathbf{i}} \right] \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\mathbf{j}} \right] \right\} / \left\{ \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\mathbf{i}} \right]^{2} + \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\mathbf{j}} \right]^{2} \right\}$$

$$\mathbf{y}_{c,15,i} = \dot{\delta}_{i}(t) = \left\{ \left[ \frac{d}{dt} \frac{\rho_{i}^{*}(t) \cdot \underline{\mathbf{k}}}{\rho_{i}^{*}(t) \cdot \underline{\mathbf{k}}} \right] / \sqrt{1 - \left[ \frac{\rho_{i}^{*}(t) \cdot \underline{\mathbf{k}}}{\rho_{i}^{*}(t) \cdot \underline{\mathbf{k}}} \right]} \right\}$$

$$y_{c,16,i} = \dot{h}_{i}(t) =$$

$$\frac{\left\{ \begin{array}{c} \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{\ell}}_{i}(t) \right] + \left[ \frac{d}{dt} \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{\ell}}_{i}(t) \right] \right\} \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{h}_{i}(t) \right] - \left\{ \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{h}}_{i}(t) \right] + \left[ \frac{d}{dt} \underline{\rho}_{i}^{*}(t) \cdot \underline{h}_{i}(t) \right] \right\} \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{\ell}}_{i}(t) \right] - \left\{ \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{h}}_{i}(t) \right] + \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{h}_{i}(t) \right] \right\} \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{\ell}}_{i}(t) \right] - \left\{ \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{h}}_{i}(t) \right] + \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{h}}_{i}(t) \right] \right\} \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{h}}_{i}(t) \right] \right\} \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{\ell}}_{i}(t) \right] + \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{h}}_{i}(t) \right] \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{h}}_{i}(t) \right] + \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{h}}_{i}(t) \right] \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{h}}_{i}(t) \right] \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{h}}_{i}(t) \right] + \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{h}}_{i}(t) \right] \left[ \underline{\dot{h}}_{i}^{*}(t) \cdot \underline{\dot{h}}_{i}(t) \right] \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{h}}$$

$$\mathbf{y}_{\mathtt{c},17,\,i} \!=\! \theta_{1,\,i}(\mathtt{t}) \!=\! \mathtt{arc} \, \tan \, \left\{ [\underline{\rho}_{i}^{*} \!\cdot\! \underline{\ell}_{i}(\mathtt{t})] / [\underline{\rho}_{i}^{*}(\mathtt{t}) \!\cdot\! \underline{n}_{i}(\mathtt{t})] \right\}$$

$$\mathbf{y}_{c,18,i} = \theta_{2,i}(t) = \arcsin \left[ \underline{\rho}_{i}^{*}(t) \cdot \underline{\mathbf{m}}_{i}(t) \right]$$
 (20)

where

$$\frac{\mathrm{d}}{\mathrm{d}t} \underline{\rho}_{i}^{*} = \frac{1}{\rho_{i}(t)} \underline{\dot{\rho}}_{i}(t) - \frac{1}{\rho_{i}(t)} [\underline{\rho}_{i}^{*}(t) \cdot \underline{\dot{\rho}}_{i}(t)] \underline{\rho}_{i}^{*}(t)$$

The expressions,  $y_{.c,k,i}$  (t), then are used to calculate the computed value of an observation of the  $k^{th}$  type, relative to the  $i^{th}$  station, at the time t.

In addition, the times at which

$$y_{c,2,i} = 0$$
 $y_{c,3,i} = 0$ 
 $y_{c,5,i} = 0$ 
 $y_{c,9,i} = 0$ 
 $y_{c,13,i} = 0$ 
 $y_{c,7,i} = \frac{\pi}{2}$ 
 $y_{c,7,i} = -\phi_i$ 

are provided in the output.

# 4. Position in Ellipse

The input to this portion of the system is the set of parameters  $\{a, e, i, \Omega, \omega, M(t_0)\}$  obtained from either the Conversion of Elements Section or the Conversion of Corrections Section. The output contains the vector quantities  $\underline{\alpha}(t)$ ,  $\underline{\beta}(t)$ ,  $\underline{\gamma}(t)$ ,  $\underline{q}(t)$ ,  $\underline{r}(t)$ ,  $\underline{\dot{r}}(t)$  obtained from the formulas listed below. Approximate values of  $\Omega$  and  $\omega$  can be obtained from

$$\Omega(t) = \Omega(t_0) + \dot{\Omega}(t - t_0)$$

$$\omega(t) = \omega(t_0) + \dot{\omega}(t - t_0)$$
(21)

where

$$\dot{\Omega} = -\frac{2\pi \cos I}{Pp^2} R_e^2 J$$

$$\dot{\omega} = \frac{4\pi \left(1 - \frac{5}{4} \sin^2 I\right)}{Pp^2} R_e^2 J,$$

then

$$\underline{\Omega}(t) = \underline{\mathbf{i}} \cos \Omega(t) + \underline{\mathbf{j}} \sin \Omega(t)$$

$$\underline{\alpha}(t) = \underline{\mathbf{k}} \cos \mathbf{I} + [\underline{\Omega}(t) \times \underline{\mathbf{k}}] \sin \mathbf{I}$$

$$\underline{\beta}(t) = \underline{\Omega}(t) \cos \delta_1 + [\underline{\alpha}(t) \times \underline{\Omega}(t)] \sin \delta_1$$

$$\underline{\gamma}(t) = \underline{\alpha}(t) \times \underline{\beta}(t)$$

$$\underline{p}(t) = \underline{\beta}(t) \cos [\omega(t) - \delta_1] + \underline{\gamma}(t) \sin [\omega(t) - \delta_1]$$

$$\underline{q}(t) = \underline{\beta}(t) \sin [\omega(t) - \delta_1] + \underline{\gamma}(t) \cos [\omega(t) - \delta_1]$$

$$\underline{r}(t) = \underline{\alpha}(t) [\cos \mathbf{E}(t) - \underline{\alpha}] + \underline{\gamma}(t) \sin \mathbf{E}(t)$$

$$\underline{\dot{r}}(t) = \underline{\alpha}(t) [\cos \mathbf{E}(t) - \underline{\alpha}] + \underline{\alpha}(t) \sin \mathbf{E}(t)$$

$$\underline{\dot{r}}(t) = \underline{\alpha}(t) [\cos \mathbf{E}(t) - \underline{\alpha}] + \underline{\alpha}(t) \sin \mathbf{E}(t)$$

where E(t) is obtained through Kepler's equation, and  $\delta_1$  here is the angle which the unit vector  $\beta$  makes with the nodal ray. This angle can be set equal to  $\omega_0$ ,  $\omega_0 + \nu_0$ ,  $\omega + \nu$ , or some other specified angle.

#### 5. Position Partial Derivatives

This portion of the system is used to evaluate partial derivatives of the form

$$\frac{\partial \underline{\mathbf{r}}(\mathbf{t})}{\partial \mathbf{x_i}}$$
,

and

$$\frac{\partial \underline{\dot{\mathbf{r}}}(\mathbf{t})}{\partial \mathbf{x_{j}}}$$
,

for  $j = 1, 2, \ldots, n$ , where quantities,  $x_j$ ,  $j = 1, 2, \ldots, n$ , are the unknowns.

A discussion of the position partial derivatives, the unknowns, and the formulas which give corrections to the elements in terms of the unknowns is contained in the section entitled Conversion of Corrections. (Cf. reference 3 in connection with relations such as (24-1) and (24-3) and related portions of this section and the Conversion of Corrections Section.)

Let

$$r(t) = a [1 - e \cos E(t)],$$

$$C(t) = (1 - e^{2}) \cos E(t),$$

$$S(t) = -\sqrt{1 - e^{2}} \sin E(t),$$
(23)

and

$$K = 2 \cos E(t_0) + e \sin^2 E(t_0)$$
.

Then

$$\frac{\partial \underline{\underline{r}}(t)}{\partial x_1} = \underline{\underline{r}}(t) - \frac{3}{2}(t - t_0) \underline{\dot{\underline{r}}}(t), \qquad (24-1)$$

$$\frac{\partial \dot{\underline{r}}(t)}{\partial x_1} = -\frac{\dot{\underline{r}}(t)}{2} + \left[ \frac{3\mu^2}{2r^3(t)} (t - t_0) \right] \underline{\underline{r}}(t)$$
 (24-2)

$$\frac{\partial \underline{r}(t)}{\partial x_2} = -\frac{\cos E(t) + e}{1 - e^2} \underline{r}(t) + \left[ \frac{2}{1 - e^2} - e \frac{\cos E(t) + e}{1 - e^2} \right] \frac{\sin E(t)}{n} \dot{\underline{r}}(t) \quad (24-3)$$

$$\frac{\partial \underline{\dot{r}}(t)}{\partial x_2} = \frac{\cos a^2}{r^2(t)(1-e^2)} \left\{ \frac{a}{r(t)} S(t) \underline{r}(t) + \cos E(t) [\underline{\alpha}(t) \times \underline{r}(t)] \right\}$$
(24-4)

$$\frac{\partial \underline{r}(t)}{\partial x_3} = \frac{a^2}{r(t)} \left[ \frac{\sin E(t)}{\sqrt{1 - e^2}} \left\{ K - 2e - C(t) \right\} \underline{p}(t) + \left\{ 1 - \left[ K - \cos E(t) \right] \cos E(t) \right\} \underline{q}(t) \right]$$
(24-5)

$$\frac{\partial \dot{\mathbf{r}}(t)}{\partial x_{3}} = \frac{na^{4}}{r^{3}(t)\sqrt{1-e^{2}}} \left[ \left[ \{1 + 2\cos E(t) \left[\cos E(t_{0}) - \cos E(t)\right] - \cos E(t) \right] \right] \\
- e \left\{ \cos E(t) \left[ \sin^{2} E(t) + \cos^{2} E(t_{0}) \right] + 2\cos E(t_{0}) \right\} \\
+ e^{2} \left[ 2\cos^{2} E(t) + \cos^{2} E(t_{0}) \right] - e^{3}\cos^{3} E(t) \right\} \underline{p}(t) \\
+ S(t) \left\{ \left[\cos E(t_{0}) - \cos E(t) \right] \left[ e \left[\cos E(t) + \cos E(t_{0}) \right] - 2 \right] \right\} \underline{q}(t) \right] (24-6)$$

$$\frac{\partial \underline{r}(t)}{\partial x_4} = \underline{\alpha}(t) \times \underline{r}(t)$$
 (24-7)

$$\frac{\partial \dot{\underline{\mathbf{r}}}(t)}{\partial \mathbf{x}_4} = \underline{\alpha}(t) \times \underline{\dot{\mathbf{r}}}(t)$$
 (24-8)

$$\frac{\partial \underline{r}(t)}{\partial x_5} = \underline{\beta}(t) x \underline{r}(t)$$
 (24-9)

$$\frac{\partial \dot{\underline{r}}(t)}{\partial x_5} = \underline{\beta}(t) \times \underline{\dot{r}}(t)$$
 (24-10)

$$\frac{\partial \underline{r}(t)}{\partial x_6} = \chi(t) x \underline{r}(t)$$
 (24-11)

$$\frac{\partial \dot{\underline{\mathbf{r}}}(t)}{\partial \mathbf{x}_{6}} = \chi(t) \mathbf{x} \dot{\underline{\mathbf{r}}}(t) \tag{24-12}$$

$$\frac{\partial \underline{r}(t)}{\partial x_7} = 2 \frac{\partial \underline{r}(t)}{\partial x_1} + C(t_0) \frac{\partial \underline{r}(t)}{\partial x_2} + S(t_0) \frac{\partial \underline{r}(t)}{\partial x_3}$$
(24-13)

$$\frac{\partial \dot{\underline{r}}(t)}{\partial x_7} = 2 \frac{\partial \dot{\underline{r}}(t)}{\partial x_1} + C(t_0) \frac{\partial \dot{\underline{r}}(t)}{\partial x_2} + S(t_0) \frac{\partial \dot{\underline{r}}(t)}{\partial x_3}$$
(24-14)

$$\frac{\partial \underline{\underline{r}}(t)}{\partial x_8} = [1 + e \cos E(t_0)] \frac{\partial \underline{\underline{r}}(t)}{\partial x_1} + C(t_0) \frac{\partial \underline{\underline{r}}(t)}{\partial x_2} + S(t_0) \frac{\partial \underline{\underline{r}}(t)}{\partial x_3}$$
(24-15)

$$\frac{\partial \dot{\underline{r}}(t)}{\partial x_8} = [1 + e \cos E(t_0)] \frac{\partial \dot{\underline{r}}(t)}{\partial x_1} + C(t_0) \frac{\partial \dot{\underline{r}}(t)}{\partial x_2} + S(t_0) \frac{\partial \dot{\underline{r}}(t)}{\partial x_3}$$
(24-16)

$$\frac{\partial \underline{\mathbf{r}}(t)}{\partial \mathbf{x}_{9}} = \mathbf{S}(t_{0}) \frac{\partial \underline{\mathbf{r}}(t)}{\partial \mathbf{x}_{2}} - \left[\mathbf{e} + \cos \mathbf{E}(t_{0})\right] \frac{\partial \underline{\mathbf{r}}(t)}{\partial \mathbf{x}_{3}}$$
(24-17)

$$\frac{\partial \dot{\underline{r}}(t)}{\partial x_9} = S(t_0) \frac{\partial \dot{\underline{r}}(t)}{\partial x_2} - \left[e + \cos E(t_0)\right] \frac{\partial \dot{\underline{r}}(t)}{\partial x_3}$$
 (24-18)

$$\frac{\partial \underline{\mathbf{r}}(t)}{\partial \mathbf{x}_{10}} = n \, \dot{\underline{\mathbf{r}}}(t) \, (t - t_0)^2 \tag{24-19}$$

$$\frac{\partial \dot{\underline{r}}(t)}{\partial x_{10}} = -\frac{(t - t_0)^2}{\sqrt{a} r^3} \left[ (\cos E - e) \underline{p} + \sqrt{1 - e^2} \sin E \underline{q} \right]$$
 (24-20)

$$\frac{\partial \underline{\mathbf{r}}(\mathbf{t})}{\partial \mathbf{x}_{11}} = \mathbf{n} (\mathbf{t} - \mathbf{t}_0) \frac{\partial \underline{\mathbf{r}}(\mathbf{t})}{\partial \mathbf{x}_{10}}$$
 (24-21)

$$\frac{\partial \dot{\mathbf{r}}(t)}{\partial \mathbf{x}_{11}} = \mathbf{n}(t - t_0) \frac{\partial \dot{\mathbf{r}}(t)}{\partial \mathbf{x}_{10}}$$
 (24-22)

$$\frac{\partial \underline{\mathbf{r}}(t)}{\partial \mathbf{x}_{19}} = \frac{\mathbf{a} (1 - \cos \mathbf{E}_0)^2}{(1 - \mathbf{e}^2) (1 - \mathbf{e} \cos \mathbf{E})} [\mathbf{S}(t) \underline{\mathbf{p}} + \mathbf{C}(t) \underline{\mathbf{q}}]$$
(24-23)

$$\frac{\partial \dot{\underline{r}}(t)}{\partial x_{19}} = \frac{a^4 n (1 - e \cos E_0)^2}{r^3 \sqrt{1 - e^2}} \left[ (e - \cos E) \underline{p} + S(t) \underline{q} \right]. \tag{24-24}$$

The partial derivatives of  $\underline{r}(t)$  and  $\underline{\dot{r}}(t)$  with respect to the unknowns  $x_{20+q}$  and  $x_{40+q}$  are the same as the partial derivatives with respect to  $x_{10}$  and  $x_{11}$ , respectively, when q = 0. For other values of the index, in the range  $q = 1, 2, \ldots, 19$ , the partial derivatives of  $\underline{r}(t)$  and  $\underline{\dot{r}}(t)$  with respect to

 $x_{20+q}$  and  $x_{40+q}$  are the same, respectively, as the partial derivatives with respect to  $x_{20}$  and  $x_{40}$  with the exception that the former partials are referred to the epoch  $t_q$ , while the latter partials are referred to the epoch  $t_0$ .

This portion of the program uses, as input, the specification of the set of unknown parameters,  $\mathbf{x}_j$ , and the times  $\mathbf{t}_n$ ,  $n=1,2,\ldots$ , at which the appropriate position partial derivatives are to be calculated. This portion of the program also uses as input either the output of the Position in Ellipse Section of the program or corresponding information from the portion of the program which is used to calculate the satellite position.

#### 6. Observation Partial Derivatives

The inputs to this portion of the system are the outputs of the Local Station Predictions Section and the Position Partial Derivatives Section. The output contains the values of the partial derivatives of the observation of type k from the i<sup>th</sup> tracking station at time  $t_n$  with respect to the unknowns,  $x_j$ , for various values of the indices. It will be understood that the quantities such as  $\underline{m}_i(t)$ ,  $\theta_i(t)$ ,  $\underline{\rho}_i^*(t)$ , etc. are functions of the time, t, and the i<sup>th</sup> tracking station, and therefore, for convenience, we shall, at times, write  $\underline{m}$ ,  $\theta$ ,  $\rho$ \*, etc. The partial derivatives are now given.

$$\frac{\partial \rho}{\partial \mathbf{x}_{j}} = \underline{\rho}^{*} \cdot \frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{x}_{j}}$$

$$\frac{\partial \ell}{\partial \mathbf{x}_{j}} = \underline{\mathbf{c}^{j} \cdot \underline{\ell}}$$

$$\frac{\partial \mathbf{m}}{\partial \mathbf{x}_{j}} = \underline{\mathbf{c}^{j} \cdot \underline{\mathbf{m}}}$$

$$\frac{\partial \theta}{\partial \mathbf{x}_{j}} = \frac{(\underline{\rho}^{*} \cdot \underline{\mathbf{m}}) \left(\frac{\partial \ell}{\partial \mathbf{x}_{j}}\right) - (\underline{\rho}^{*} \cdot \underline{\ell}) \left(\frac{\partial \mathbf{m}}{\partial \mathbf{x}_{j}}\right)}{(\underline{\rho}^{*} \cdot \underline{\mathbf{m}})^{2} + (\underline{\rho}^{*} \cdot \underline{\ell})^{2}}$$

$$\frac{\partial \phi}{\partial \mathbf{x}_{j}} = \frac{(\underline{\mathbf{c}^{j} \cdot \underline{\mathbf{m}}})}{\sqrt{1 - (\rho^{*} \cdot \underline{\mathbf{n}})^{2}}}$$

$$\frac{\partial a}{\partial \mathbf{x}_{i}} = \frac{(\underline{\rho}^{*} \cdot \underline{\mathbf{i}}) \cdot (\underline{c}^{i} \cdot \underline{\mathbf{j}}) - (\underline{\rho}^{*} \cdot \underline{\mathbf{j}})^{2}}{(\underline{\rho}^{*} \cdot \underline{\mathbf{i}})^{2} + (\underline{\rho}^{*} \cdot \underline{\mathbf{j}})^{2}}$$

$$\frac{\partial \delta}{\partial \mathbf{x}_{i}} = \frac{(\underline{c}^{j} \cdot \underline{\mathbf{k}})}{\sqrt{1 - (\underline{\rho}^{*} \cdot \underline{\mathbf{k}})^{2}}}$$

$$\frac{\partial h}{\partial \mathbf{x}_{i}} = -\frac{\partial a}{\partial \mathbf{x}_{i}}$$

$$\frac{\partial \theta_{1}}{\partial \mathbf{x}_{i}} = \frac{(\underline{\rho}^{*} \cdot \underline{\mathbf{n}}) \cdot (\underline{c}^{j} \cdot \underline{\ell}) - (\underline{\rho}^{*} \cdot \underline{\ell}) \cdot (\underline{c}^{j} \cdot \underline{\mathbf{n}})}{(\underline{\rho}^{*} \cdot \underline{\mathbf{n}})^{2} + (\underline{\rho}^{*} \cdot \underline{\ell})^{2}}$$

$$\frac{\partial \dot{\rho}}{\partial \mathbf{x}_{i}} = \frac{\partial \dot{\rho}}{\partial \mathbf{x}_{i}} = \frac{(\underline{c}^{j} \cdot \underline{\mathbf{m}})}{\sqrt{1 - (\underline{\rho}^{*} \cdot \underline{\mathbf{m}})^{2}}}$$

$$\frac{\partial \dot{\rho}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\ell})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\mathbf{m}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\mathbf{m}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\mathbf{m}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\mathbf{m}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\mathbf{m}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\mathbf{m}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\mathbf{m}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{c}}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{c}^{j} \cdot \underline{\dot{\mu}}) + (\underline{\dot{\mu}}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{\dot{\theta}}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{\dot{\theta}}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{\dot{\theta}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{\dot{\theta}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{\dot{\theta}^{j} \cdot \underline{\dot{\mu}})$$

$$\frac{\partial \dot{\theta}}{\partial \mathbf{x}_{i}} = (\underline{\dot{\theta}^{j}$$

$$\frac{\partial \dot{\phi}}{\partial \mathbf{x}_{j}} = \frac{(\dot{\mathbf{c}}^{j} \cdot \underline{\mathbf{n}}) + (\mathbf{c}^{j} \cdot \dot{\underline{\mathbf{n}}})}{\sqrt{1 - (\underline{\rho}^{*} \cdot \underline{\mathbf{n}})^{2}}}$$

•

$$+\frac{(\underline{c}^{j} \cdot \underline{n}) \ (\underline{\rho}^{*} \cdot \underline{n}) \ [(\underline{d} \cdot \underline{n}) + (\underline{\rho}^{*} \cdot \underline{n})]}{[1 - (\underline{\rho}^{*} \cdot \underline{n})^{2}]^{3/2}}$$

$$\frac{\partial \dot{\alpha}}{\partial \mathbf{x}^{i}} = \frac{(\underline{\partial}^{k}) (\underline{\mathbf{q}} \cdot \underline{\mathbf{i}}) - (\underline{\partial}^{m}) (\underline{\mathbf{q}} \cdot \underline{\mathbf{i}})^{2}}{(\underline{\rho}^{k} \cdot \underline{\mathbf{i}})^{2} + (\underline{\rho}^{k} \cdot \underline{\mathbf{i}})^{2}}$$

$$+\frac{\left(\frac{\partial \dot{\ell}}{\partial \mathbf{x}_{\mathbf{j}}}\right)(\underline{\rho}^{*}\cdot\underline{\mathbf{i}})-\left(\frac{\partial \dot{\mathbf{m}}}{\partial \mathbf{x}_{\mathbf{j}}}\right)(\underline{\rho}^{*}\cdot\underline{\mathbf{j}})}{(\underline{\rho}^{*}\cdot\underline{\mathbf{i}})^{2}+(\underline{\rho}^{*}\cdot\underline{\mathbf{j}})^{2}}$$

$$-2\left(\frac{\partial a}{\partial \mathbf{x}_{\mathbf{j}}}\right)\left[\frac{(\underline{\rho}^{*}\cdot\underline{\mathbf{i}})(\underline{\mathbf{d}}\cdot\underline{\mathbf{i}})+(\underline{\rho}^{*}\cdot\underline{\mathbf{j}})(\underline{\mathbf{d}}\cdot\underline{\mathbf{j}})}{(\underline{\rho}^{*}\cdot\underline{\mathbf{i}})^{2}+(\underline{\rho}^{*}\cdot\underline{\mathbf{j}})^{2}}\right]$$

$$\frac{\partial \dot{\delta}}{\partial \mathbf{x}_{j}} = \frac{(\dot{\underline{c}}^{j} \cdot \underline{\mathbf{k}})}{\sqrt{1 - (\underline{\rho}^{*} \cdot \underline{\mathbf{k}})^{2}}} + \frac{(\underline{c}^{j} \cdot \underline{\mathbf{k}}) \ (\underline{\rho}^{*} \cdot \underline{\mathbf{k}}) \ (\underline{d} \cdot \underline{\mathbf{k}})}{[1 - (\underline{\rho}^{*} \cdot \underline{\mathbf{k}})^{2}]^{3/2}}$$

where

$$\frac{\partial \mathbf{h}}{\partial \mathbf{x}_{j}} = -\frac{\partial \dot{\alpha}}{\partial \mathbf{x}_{j}}$$

$$\underline{\mathbf{c}}^{j} = \frac{\partial \underline{\rho}^{*}}{\partial \mathbf{x}_{j}} = \frac{1}{\rho} \left( \frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{x}_{j}} - \underline{\rho}^{*} \frac{\partial \rho}{\partial \mathbf{x}_{j}} \right); \ \underline{\mathbf{d}} = \frac{\mathbf{d}\underline{\rho}^{*}}{\mathbf{d}\mathbf{t}} = \frac{1}{\rho} \left( \underline{\dot{\rho}} - \underline{\rho}^{*} \dot{\rho} \right)$$

and

$$\underline{\dot{\mathbf{c}}}^{\mathbf{j}} = \frac{1}{\rho} \left( \frac{\partial \underline{\dot{\mathbf{r}}}}{\partial \mathbf{x}_{\mathbf{j}}} - \underline{\rho}^* \frac{\partial \dot{\rho}}{\partial \mathbf{x}_{\mathbf{j}}} - \underline{\mathbf{d}} \frac{\partial \rho}{\partial \mathbf{x}_{\mathbf{j}}} - \underline{\mathbf{c}}^{\mathbf{j}} \dot{\rho} \right)$$

# 7. Equations of Condition

The input to this portion of the system includes the values of the various  $y_{0,k,i}$   $(t_n)$ , values for weights  $w_{k,i}$   $(t_n)$ , and the outputs of the Observation Partial Derivatives Section and the Local Station Predictions Section. The output of this portion of the system contains the equations of condition and the normal equations.

The equations of condition are

$$\sum \frac{\partial y_{k,i}(t_n)}{\partial x_i} x_j = y_{0,k,i}(t_n) - y_{c,k,i}(t_n).$$
 (26)

The normal equations are

$$\sum_{i} \left[ \sum_{j} w_{k,i} \frac{\partial y_{i}}{\partial x_{j}} \frac{\partial y_{i}}{\partial x_{\lambda}} x_{j} - w_{k,i} \frac{\partial y_{i}}{\partial x_{\lambda}} (y_{0,k,i} - y_{c,k,i}) \right] = 0 \quad (\lambda = 1, 2, \cdots) \quad (27)$$

where the weights,  $w_{k,i}$ , are functions of the observation type, estimated or observed uncertainties, and factors associated with the geometry of a particular pass at a particular station. Weights, including null weights, can be assigned to subsets of observations selected in terms of time, station, type, or residual characteristics.

#### 8. Least Squares Solution

The input to this portion of the system is the output of the Equations of Condition Section. The output contains values for the various quantities,  $\mathbf{x}_{j}$ , and statistical information associated with the fit to the observations.

#### 9. Conversion of Corrections

The input to this portion of the system consists of the output of the Least Squares Solution Section, the Conversion of Elements Section and the Conversion of Corrections Section associated with the current iteration. The output includes the original value, the previous value, the new value of each of the parameters, and the standard deviation of fit, and its increment.

Corrections to the previous values or the original values of the elements and associated parameters are obtained in the manner indicated in the following discussion.

The partial derivatives and the corresponding expressions for the differential changes in the elements as functions of the unknowns are employed in sets. For example, when the partial derivatives

$$\frac{\partial \mathbf{r}}{\partial \mathbf{x} \mathbf{j}}$$
, (28)

j = 1, 2, ..., 6, i.e., the expressions (24-1), (24-3), (24-5), (24-7), (24-9), and (24-11), are employed, the use of formulas for obtaining differential changes in the elements as functions of the unknowns can be indicated in the following way.

The expression

$$\delta \mathbf{a} = \mathbf{a} \mathbf{x}_1 , \qquad (29)$$

gives the differential change,  $\delta\,a$  , in the semi-major axis, a , as a function of the unknown,  $x_1$  .

Similarly, the expression

$$\delta e = x_2, \qquad (30)$$

gives the differential change,  $\delta e$ , in the eccentricity, e, as a function of the unknown,  $x_2$ .

The differential change,  $\delta i$ , in the inclination, i, is given by the relation

$$\delta i = \delta \beta \cos \delta_1 - \delta \gamma \sin \delta_1, \qquad (31)$$

where the quantities  $\delta \beta$  and  $\delta \gamma$  are obtained from the relations

$$\delta \beta = \mathbf{x}_5, \qquad (32)$$

and

$$\delta \gamma = x_6 . ag{33}$$

Hence, the expression for  $\delta i$  directly in terms of the unknowns is

$$\delta i = x_5 \cos \delta_1 - x_6 \sin \delta_1. \tag{34}$$

In similar fashion, the expression obtained for the differential change,  $\delta \Omega$ , in the right ascension of the ascending node,  $\Omega$ , directly in terms of the unknowns is

$$\delta \Omega = \csc i \left( x_5 \sin \delta_1 + x_6 \cos \delta_1 \right), \tag{35}$$

the expression obtained for the differential change,  $\delta M_0$ , in the mean anomaly at the epoch,  $M_0$ , directly in terms of the unknown is

$$\delta M_0 = \frac{(1 - e \cos E_0)^2}{\sqrt{1 - e^2}} \frac{1}{e} x_3, \qquad (36)$$

and the expression obtained for the differential change,  $\delta \omega$ , in the argument of perigee,  $\omega$ , directly in terms of the unknowns is

$$\delta \omega = x_4 - \frac{x_3}{e} - \operatorname{ctn} i \left( x_5 \sin \delta_1 + x_6 \cos \delta_1 \right). \tag{37}$$

The relationships between these two sets of formulas for computation, i.e., those for the partial derivatives, and those for the corresponding differential changes in the elements as functions of the unknowns, can be indicated in the following way.

In view of the relation (29) we can write

$$\frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{x}_1} = \mathbf{a} \frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{a}},\tag{38}$$

from which expression (24-1) can be derived. (Cf. reference 3 in connection with this relation and portions of the following discussion.) In view of the relation (30) we can write

$$\frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{x_2}} = \frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{e}} , \qquad (39)$$

from which expression (24-3) follows.

The quantities  $\delta\beta$  and  $\delta\gamma$  denote, respectively, differential rotations of the orbit plane about the  $\beta$  and  $\gamma$  axes, as can be seen, respectively, from the pairs of expressions (24-9) and (32), and (24-11) and (33). We can also write the expression (37) in the following form:

$$\delta \omega = \delta \omega_{\alpha} - \operatorname{ctn} i \left( \delta \beta \sin \delta_{1} + \delta \gamma \cos \delta_{1} \right), \tag{40}$$

where the terms involving  $\delta\beta$  and  $\delta\gamma$  correspond to the differential changes in  $\omega$  due to differential rotations of the orbit plane about the  $\underline{\beta}$  and  $\underline{\gamma}$  axes, respectively, and the term  $\delta\omega_a$  corresponds to differential changes in  $\omega$  due to differential changes within the orbit plane, i.e., to differential rotations about the  $\underline{\alpha}$  axis, i.e.,

$$\frac{\partial \underline{\mathbf{r}}}{\partial \beta} = \underline{\beta} \times \underline{\mathbf{r}} , \qquad (41)$$

$$\frac{\partial \mathbf{r}}{\partial \gamma} = \gamma \times \mathbf{r} , \qquad (42)$$

and

$$\frac{\partial \underline{\mathbf{r}}}{\partial \omega_a} = \underline{\alpha} \times \underline{\mathbf{r}} , \qquad (43)$$

and, from (37) and (40),

$$\delta \omega_{\alpha} = x_4 - \frac{1}{e} x_3. \tag{44}$$

In view of (36) and (44), we have

$$\frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{x}_{3}} = \frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{M}_{0}} \left[ \frac{(1 - e \cos \mathbf{E}_{0})^{2}}{(\sqrt{1 - e^{2}})e} \right] + \frac{\partial \underline{\mathbf{r}}}{\partial \omega_{\alpha}} \left[ -\frac{1}{e} \right], \tag{45}$$

and

$$\frac{\partial \mathbf{r}}{\partial \mathbf{x_4}} = \frac{\partial \mathbf{r}}{\partial \omega_a}, \tag{46}$$

from which expressions (24-5) and (24-7), respectively, can be derived.

The partial derivatives

$$\frac{\partial \underline{r}}{\partial \mathbf{x_j}}$$
, (47)

$$j = 1, 2, 3, 5, 6, 19,$$

i.e., the expressions (24-1), (24-3), (24-9), (24-9), (24-11), and (24-23), are frequently employed. The use of formulas for obtaining differential changes in the elements as functions of the unknowns in this case can be indicated in the following way.

The expressions (29), (30), (34), and (35) occur as in the case discussed previously. The differential changes in the mean anomaly at the epoch and the argument of perigee obtained in this case will be denoted, respectively, by the symbols

$$\delta M_{01}$$
, and  $\delta \omega_1$ ,

in order to distinguish them from the corresponding quantities which were discussed above.

The expression obtained for  $\,\delta\,M_{01}^{}$  , directly in terms of the unknowns is, in this case,

$$\delta M_{01} = \frac{(1 - e \cos E_0)^2}{\sqrt{1 - e^2}} \left[ \frac{1}{e} x_3 + x_{19} \right]$$
 (48)

The expression obtained for  $\delta \omega_1$ , directly in terms of the unknowns is, in this case,

$$\delta \omega_1 = -\frac{x_3}{e} - \operatorname{ctn} i \left( x_5 \sin \delta_1 + x_6 \cos \delta_1 \right). \tag{49}$$

We can also write the expression (49) in the following form:

$$\delta \omega_1 = \delta \omega_{1a} - \cot i \, (\delta \beta \sin \delta_1 + \delta \gamma \cos \delta_1), \tag{50}$$

where, here, the terms involving  $\delta\beta$  and  $\delta\gamma$  correspond to the differential changes in the argument of perigee due to differential rotations of the orbit plane about the  $\beta$  and  $\gamma$  axes, respectively, and the term  $\delta\omega_{1\alpha}$  corresponds to differential changes in the argument of perigee due to differential changes within the orbit plane, i.e., to differential rotations about the  $\alpha$  axis, i.e., we have the relations (41) and (42) and

$$\frac{\partial \underline{\mathbf{r}}}{\partial \omega_{\mathbf{1}\alpha}} = \underline{\alpha} \times \underline{\mathbf{r}} \,, \tag{51}$$

and, from (49) and (50),

$$\delta \,\omega_{1\alpha} = -\frac{x_3}{e} \,. \tag{52}$$

In view of (48) and (52) we have

$$\frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{x}_{3}} = \frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{M}_{01}} \left[ \frac{(1 - e \cos \mathbf{E}_{0})^{2}}{(\sqrt{1 - e^{2}}) e} \right] + \frac{\partial \underline{\mathbf{r}}}{\partial \omega_{1\alpha}} \left[ -\frac{1}{e} \right], \tag{53}$$

and

$$\frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{x}_{19}} = \frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{M}_{01}} \begin{bmatrix} (1 - e \cos \mathbf{E}_0)^2 \\ \sqrt{1 - e^2} \end{bmatrix}, \tag{54}$$

from which expressions (24-5) and (24-23), respectively, can be derived.

Corrections to previous or original values of other parameters of interest are obtained with the aid of formulas (55) through (67).

$$\delta \mathbf{r}_{0} = \frac{\mathbf{r}_{0}^{2}}{\mathbf{a}} \mathbf{x}_{7} , \qquad (55)$$

$$\delta v_0 = \frac{na}{2} \sqrt{1 - e^2 \cos^2 E_0} x_8, \qquad (56)$$

$$\delta \theta_0 = \mathbf{x}_0 \,, \tag{57}$$

$$\delta n_2 = n^2 x_{10}, (58)$$

$$\delta n_3 = n^3 x_{11}$$
, (59)

The corrections  $\delta n_{2,0}$  and  $\delta n_{3,0}$  bear the same relationship to the unknowns  $x_{20}$  and  $x_{40}$ , respectively, as do the corrections  $\delta n_2$  and  $\delta n_3$  to the unknowns  $x_{10}$  and  $x_{11}$ , respectively.

The ratio of  $\delta n_{2,q}$  to  $x_{20+q}$  is the same as the ratio of  $\delta n_2$  to  $x_{10}$ , for  $q=0,1,2,\cdots,19$ , and similarly, the ratio of  $\delta n_{3,q}$  to  $X_{40+q}$  is the same as the ratio of  $\delta n_3$  to  $x_{11}$ , for  $q=0,1,2,\cdots,19$ .

$$\delta \underline{\mathbf{r}} = \sum_{i=1}^{n} \frac{\partial \underline{\mathbf{r}}}{\partial \mathbf{x}_{i}} \mathbf{x}_{i}, \qquad (60)$$

$$\delta \, \, \underline{\dot{\mathbf{r}}} = \sum_{i=1}^{n} \, \frac{\partial \, \underline{\dot{\mathbf{r}}}}{\partial \, \mathbf{x}_{i}} \, \, \mathbf{x}_{i} \,, \tag{61}$$

$$\delta \mathbf{x} = \delta \mathbf{\underline{r}} \cdot \mathbf{\underline{i}}, \tag{62}$$

$$\delta \mathbf{y} = \delta \mathbf{\underline{r}} \cdot \mathbf{\underline{j}}, \tag{63}$$

$$\delta z = \delta r \cdot \underline{k}, \tag{64}$$

$$\delta \dot{\mathbf{x}} = \delta \dot{\mathbf{r}} \cdot \dot{\mathbf{i}}, \tag{65}$$

$$\delta \dot{\mathbf{y}} = \delta \, \underline{\dot{\mathbf{r}}} \, \cdot \, \mathbf{j} \,, \tag{66}$$

and

$$\delta \dot{\mathbf{z}} = \delta \dot{\mathbf{r}} \cdot \mathbf{\underline{k}} . \tag{67}$$

These latter relations are used, for example, when the spacecraft position is specified by means of a special perturbation numerical integration method referred to Cartesian coordinates.

This completes the outline of the contents of the nine basic portions of the Goddard General Orbit Determination System.

## IV. SUMMARY

In the previous pages of this report the mathematical description of the Goddard General Orbit Determination System is presented. Appendix A contains the schematic flowchart for the differential correction portion of the system. In Appendices B and C, respectively, the program flowchart and program listing are to be found. It is hoped that the formulation together with the program information will be of assistance to the many users of the Goddard General Orbit Determination System as well as to the recipients of the orbital information which it is used to generate.

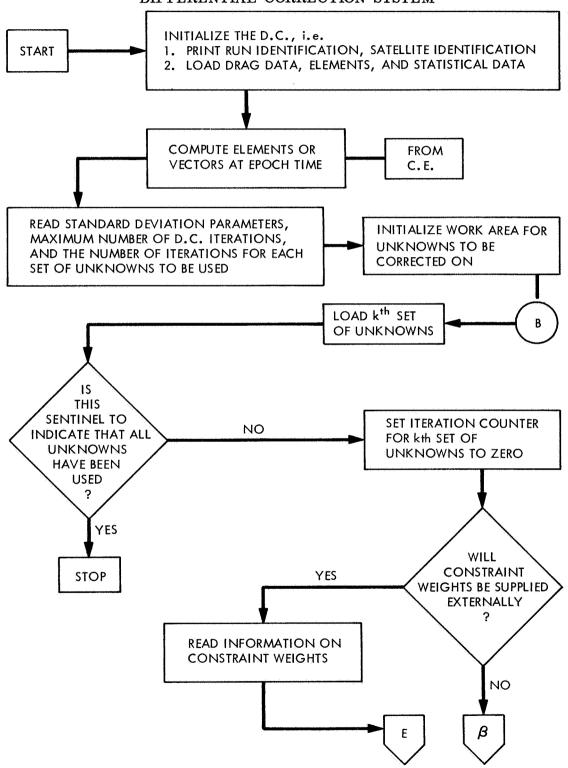
# V. ACKNOWLEDGMENT

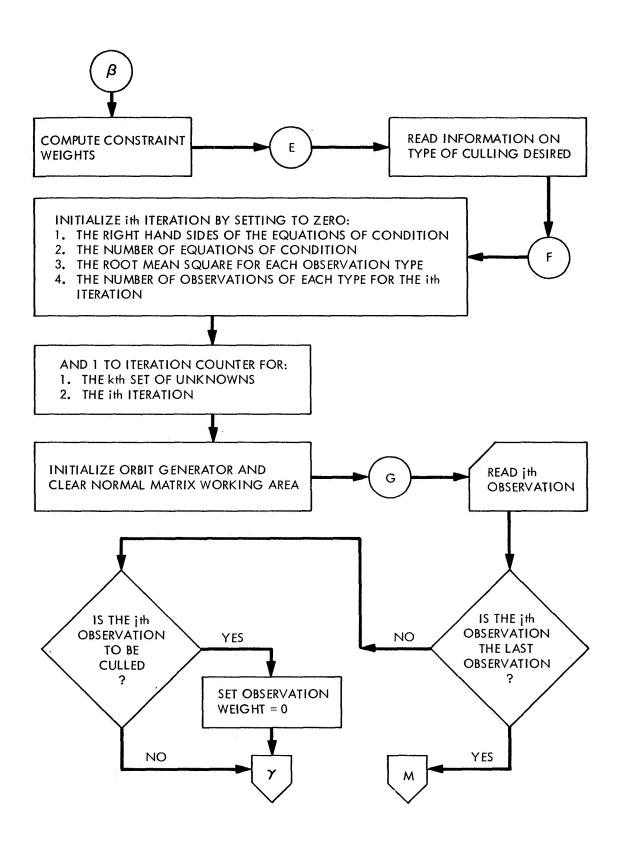
The Goddard General Orbit Determination System was programmed by a team of Goddard staff members under the direction of Thomas P. Gorman, who served for a number of years as Head of the Data Systems Division's Advanced Orbital Programming Branch, and Melba Mouton, who succeeded him in that position and is currently serving as Head of the Mission and Trajectory Analysis Division's Program Systems Branch. The authors also wish to express their thanks to Dr. Hans G. Hertz for his careful reading of the manuscript.

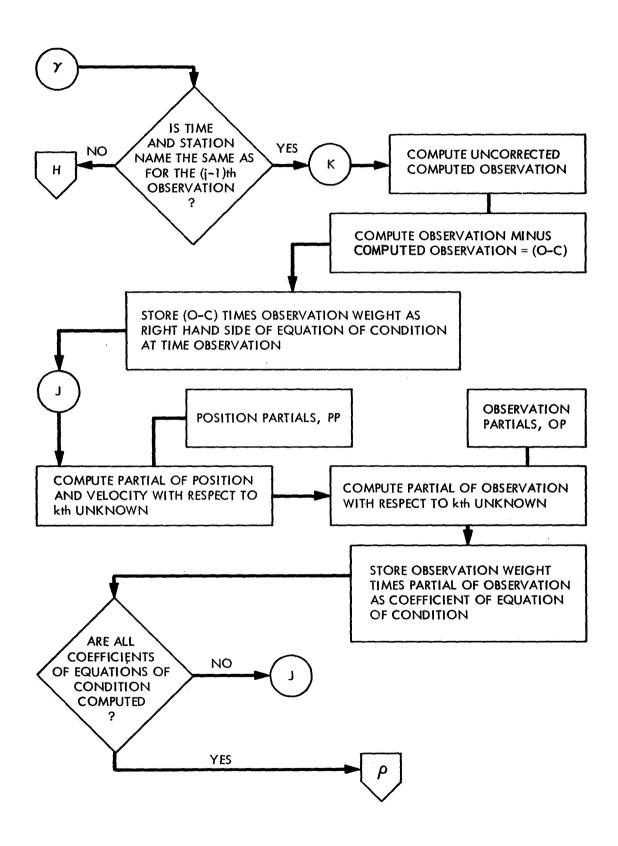
## VI. REFERENCES

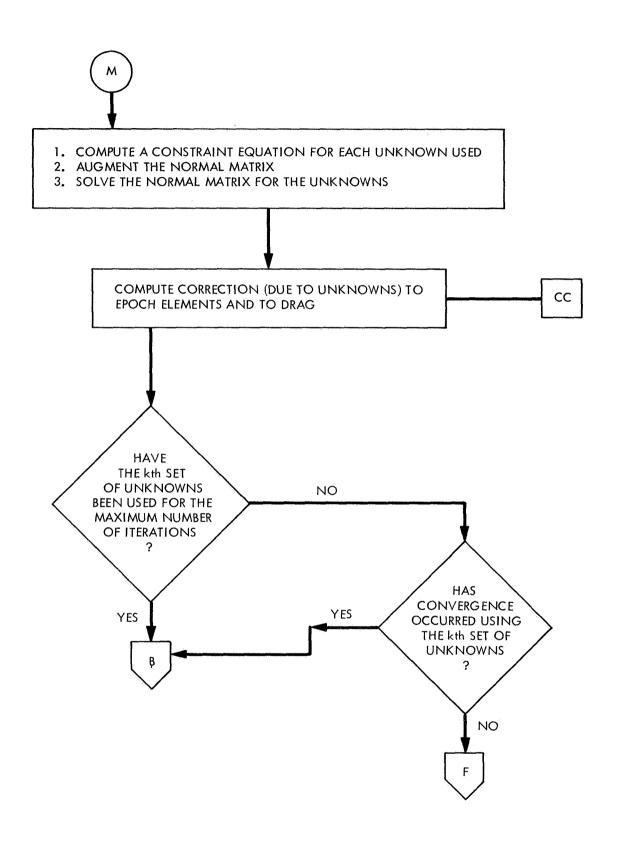
- 1. Brouwer, D., 1959, A. J. (64).
- 2. Brouwer, D. and Clemence, G. M., <u>Methods of Celestial Mechanics</u>, Academic Press, New York, 1961.
- 3. Eckert, W. J. and Brouwer D., 1937, A. J., (46).

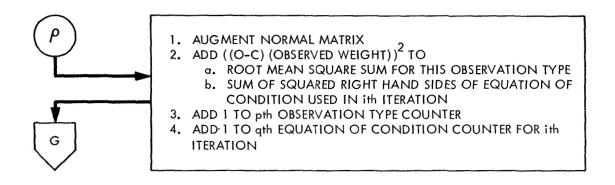
# APPENDIX A SCHEMATIC FLOW CHART OF THE DIFFERENTIAL CORRECTION SYSTEM

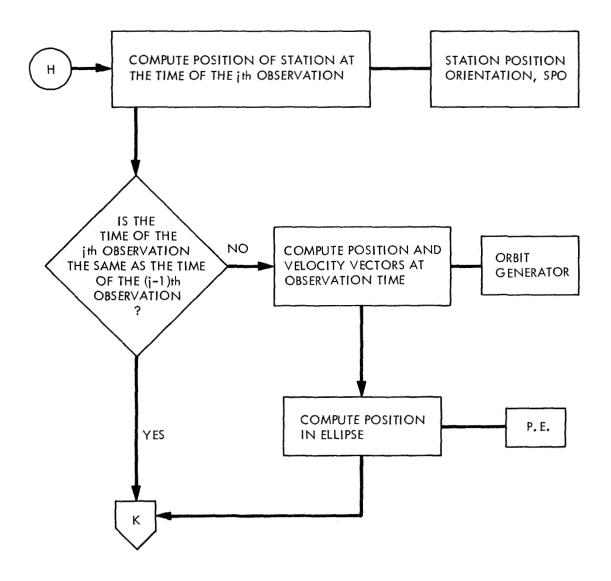










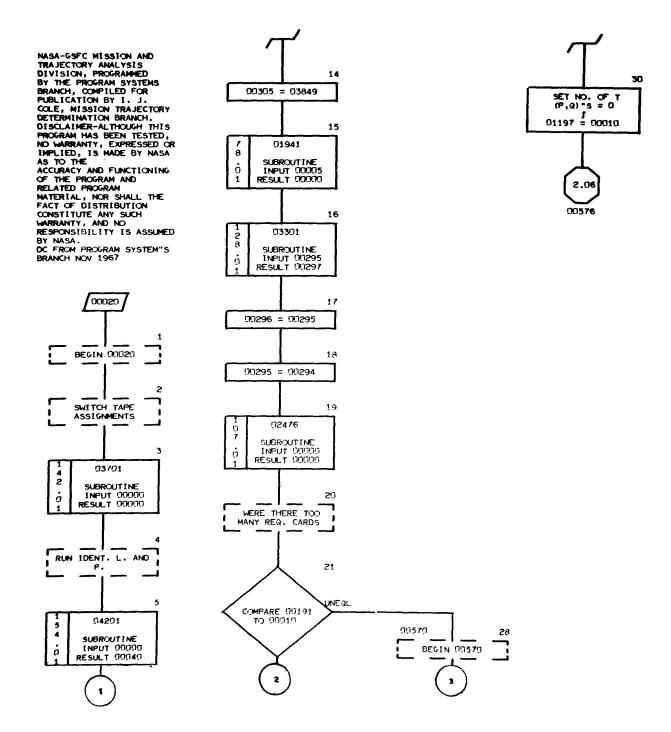


# APPENDIX B

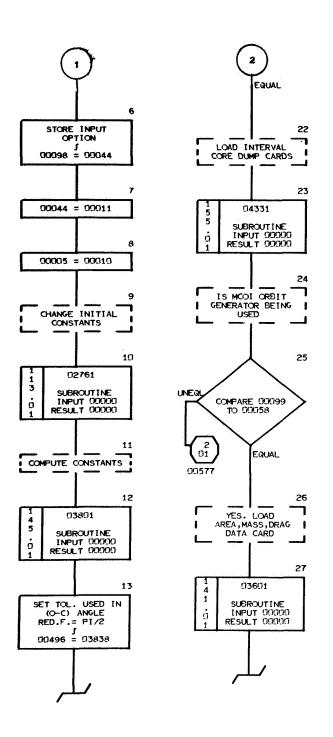
Differential Correction System Flowchart

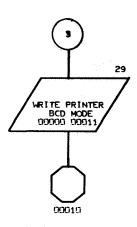
The Differential Correction program is designed so that the basic computations are performed by several subroutines. This portion of this paper presents the logic of these subroutines and describes their relationships to each other and to the main or executive program. It is intended to be an aid in relating the actual computations to the formulae given earlier and to show the logical flow of these computations.

The location statements (Q statements) and constant statements (V statements) for each subroutine are presented along with a Table of Contents and Cross-Reference Listing for the subroutine. The Cross-Reference Listings show for each label in the subroutine the page and symbol number for the label itself and for all points in the flow charts at which transfers to it are found. These latter points are referred to as references in the Cross Reference Listings. These page numbers refer to the flow charts and are at the right uppermost corner of each flow chart page. They are not to be confused with the page numbers of this paper itself. An asterisk, \*, indicates that there are transfers to a label other than the one shown.

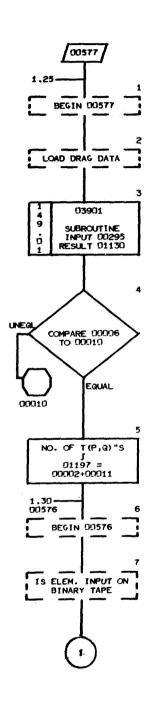


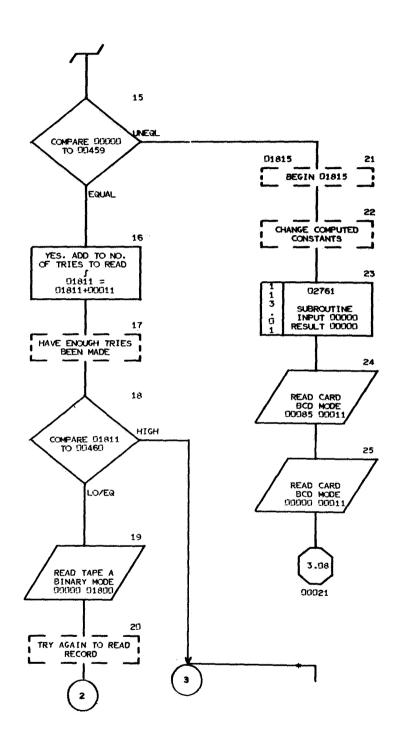
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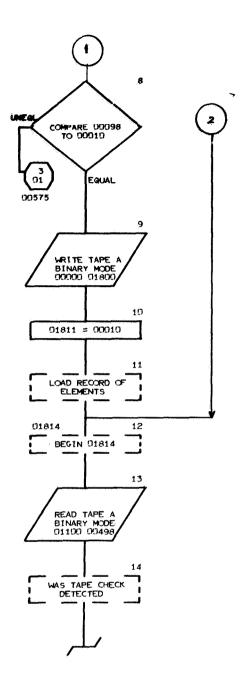


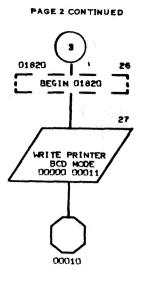


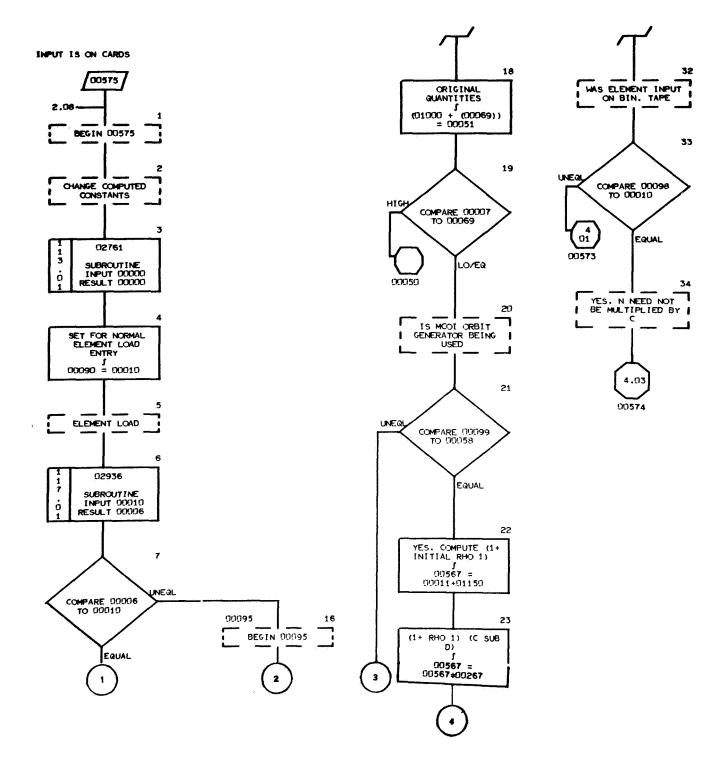
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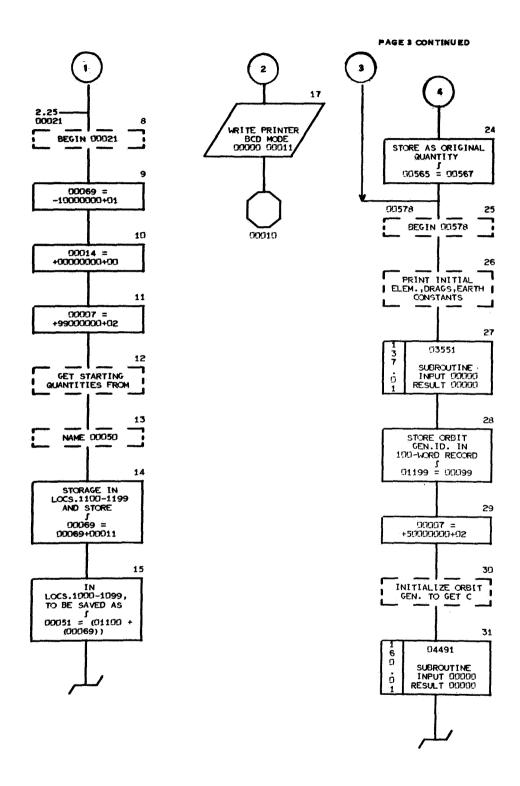


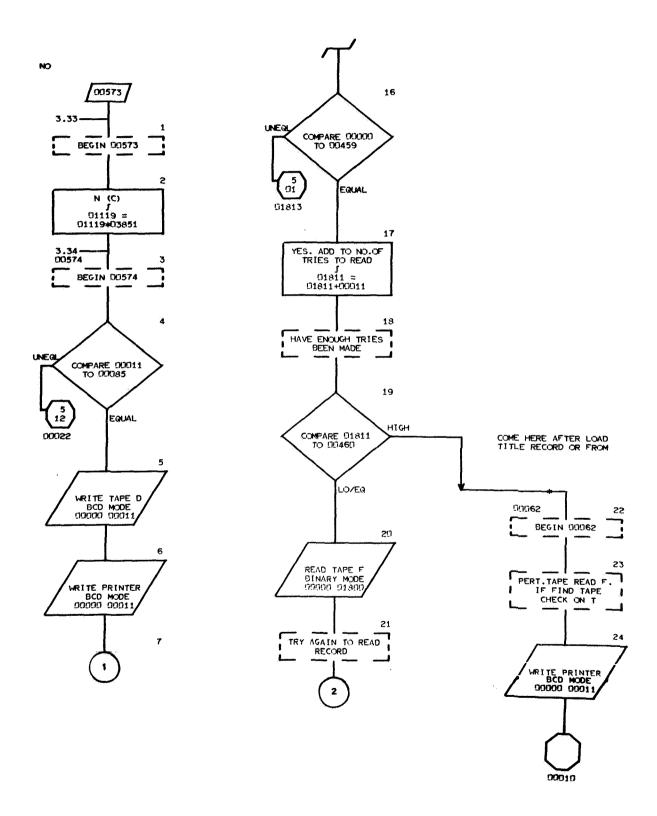


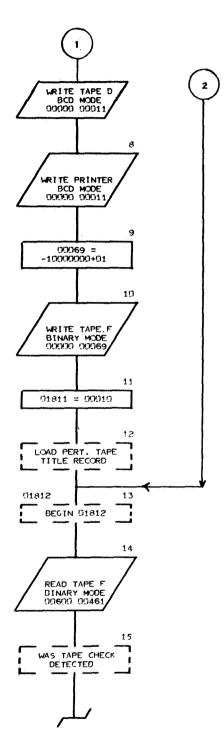


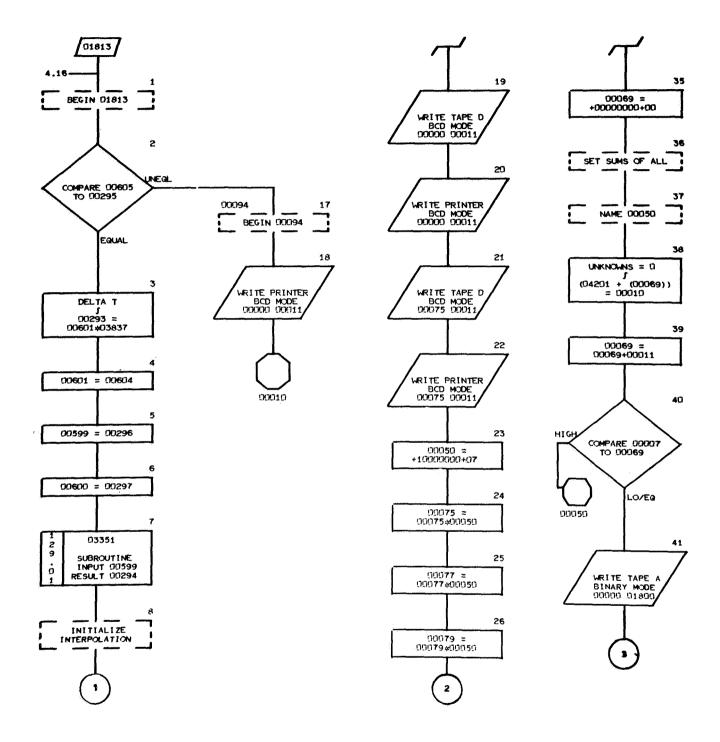




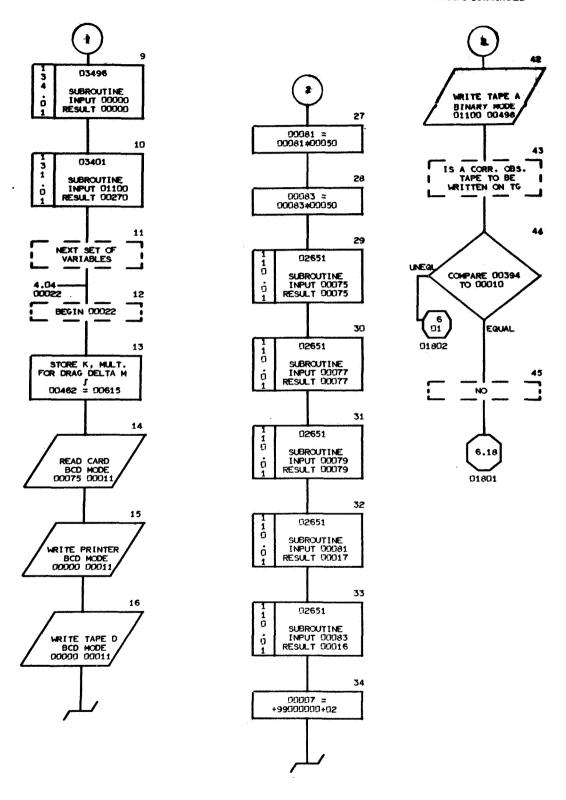


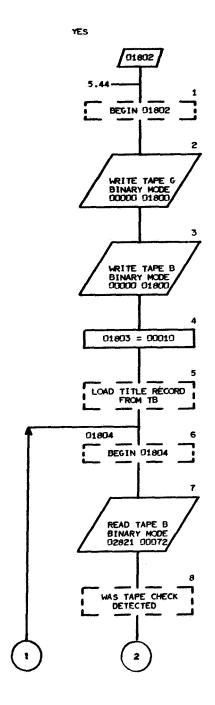


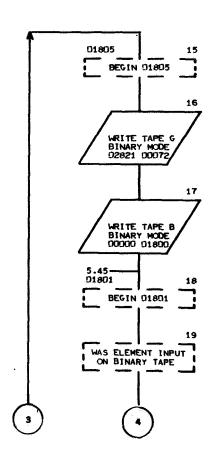


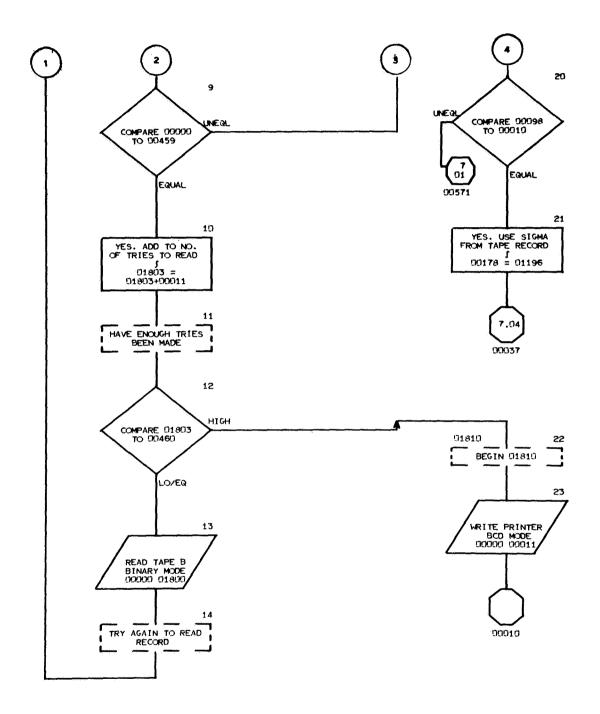


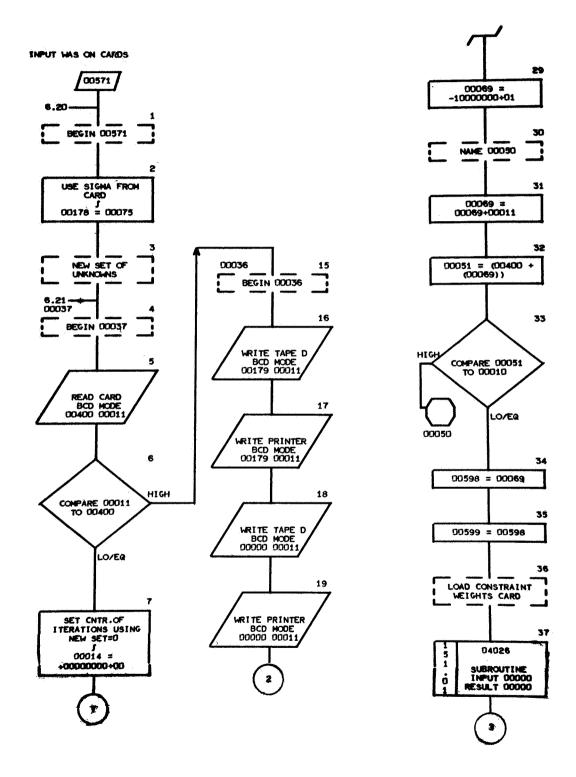
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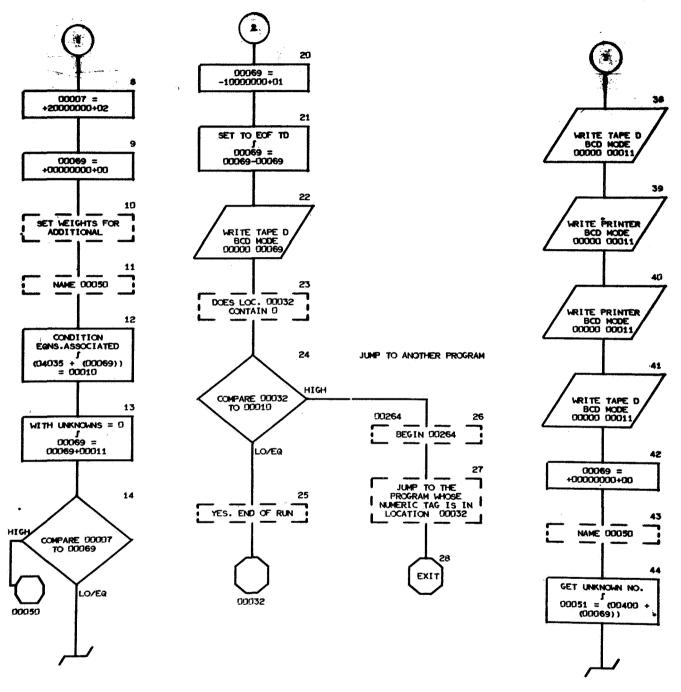


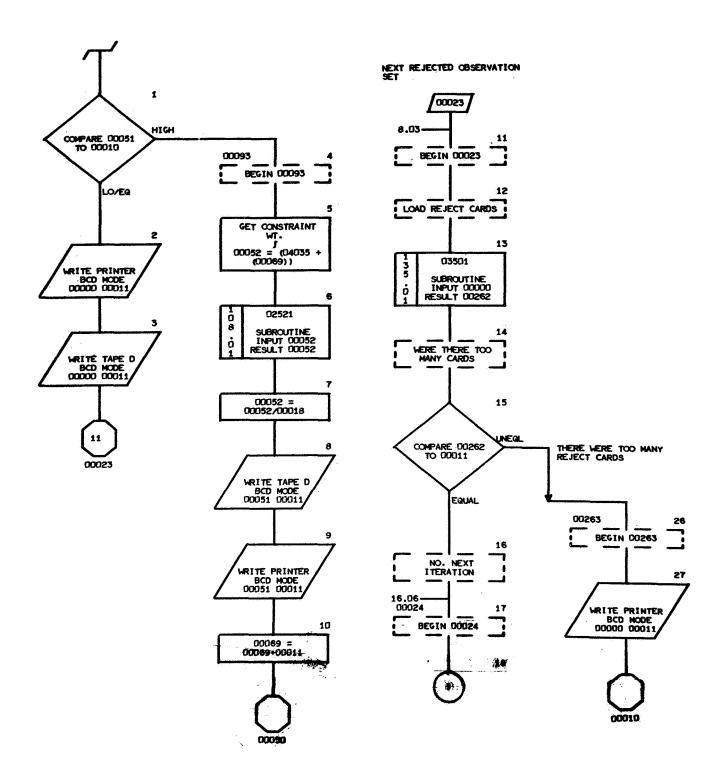


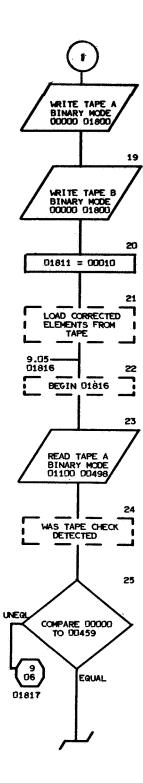


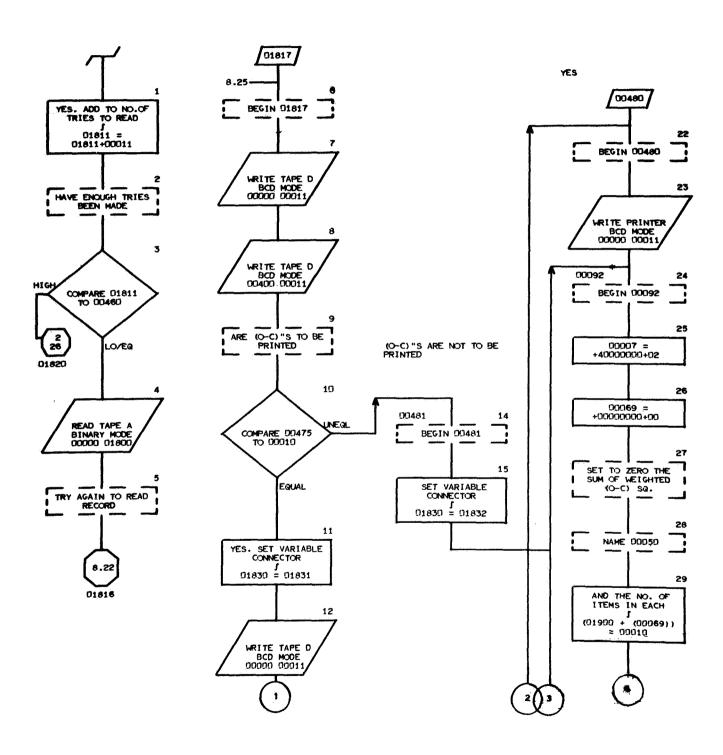


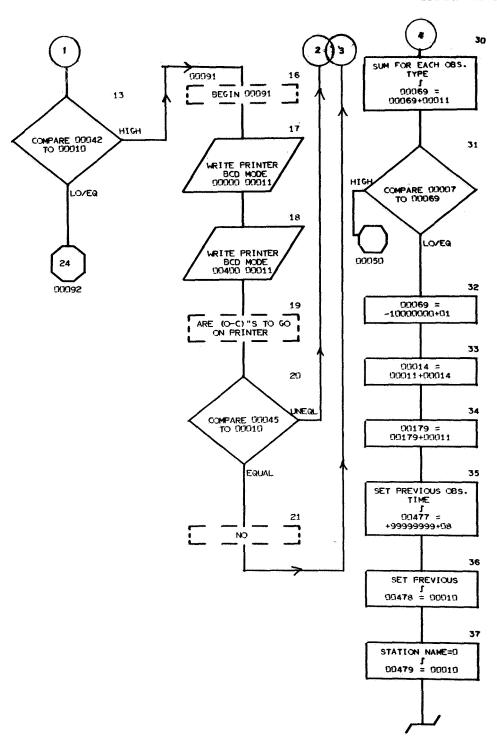
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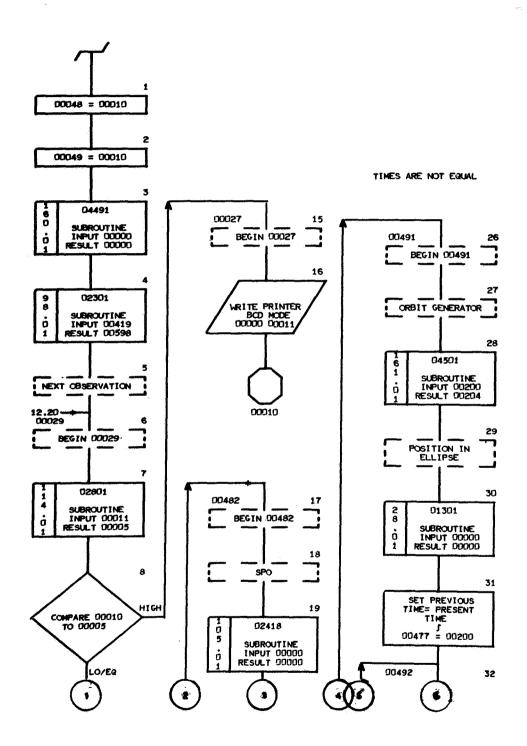


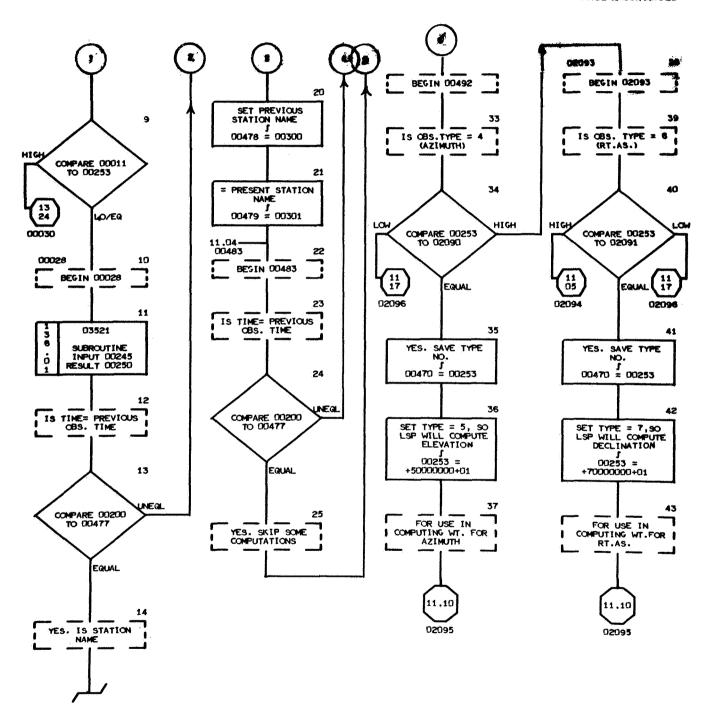


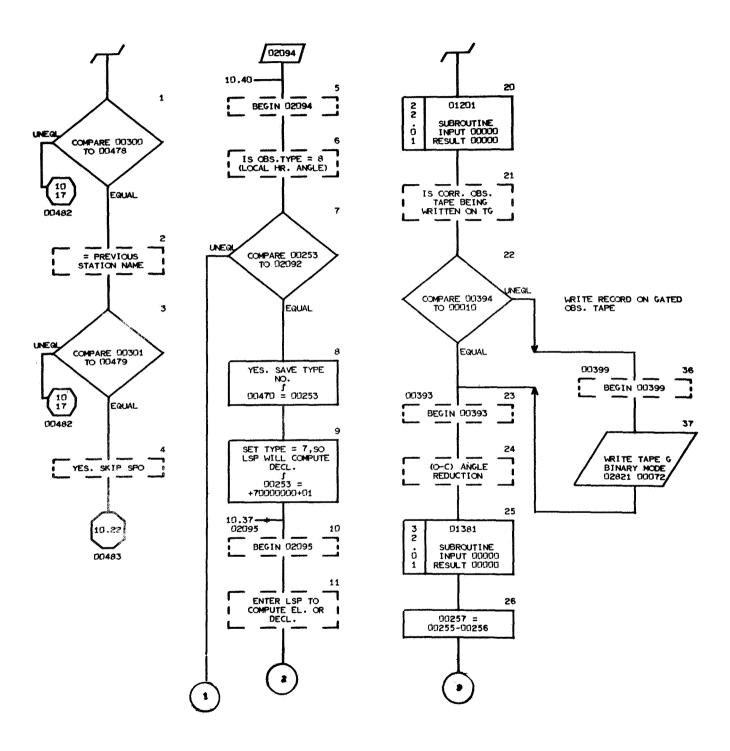


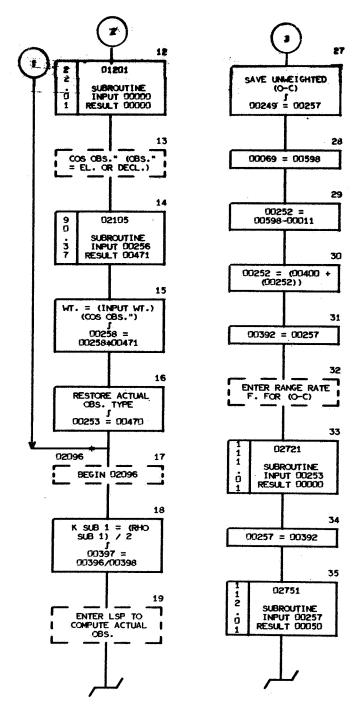


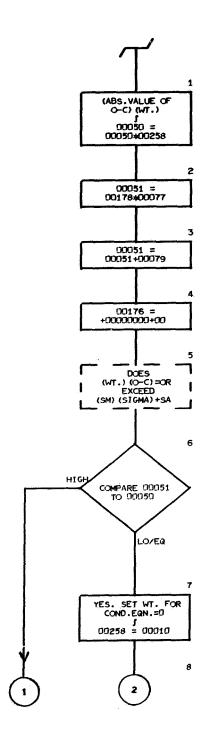


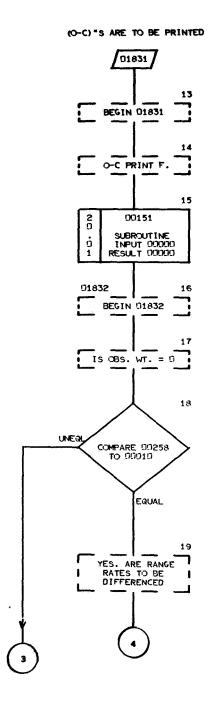




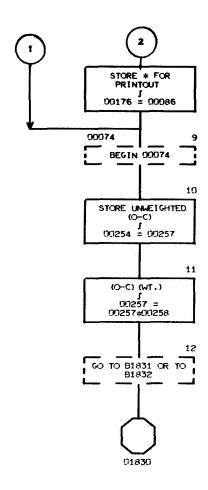


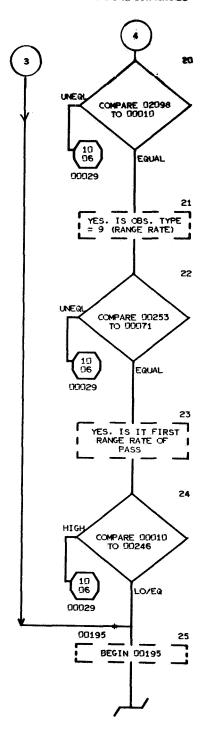


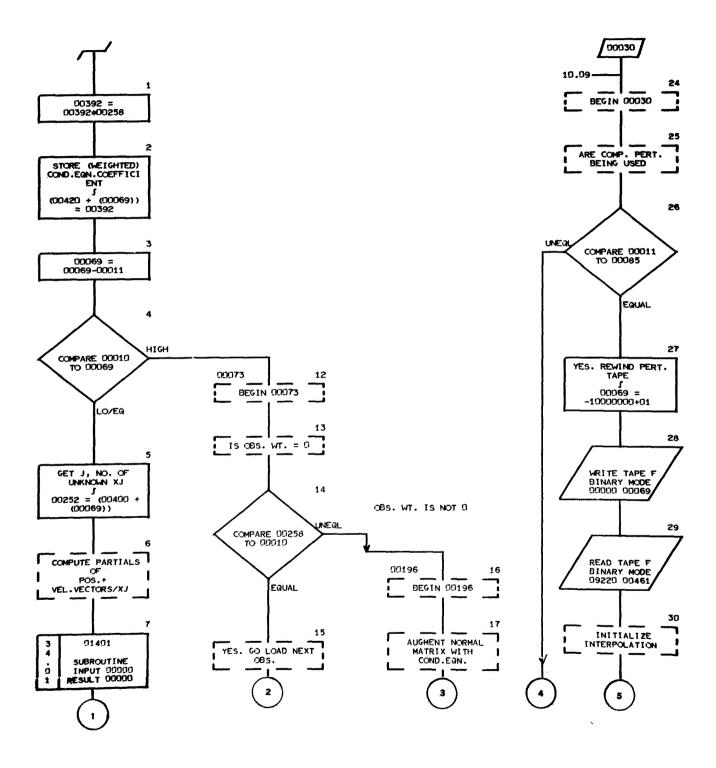




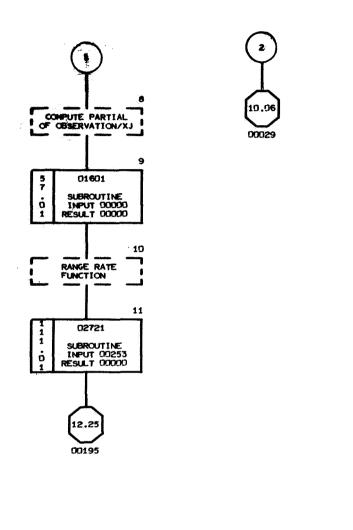
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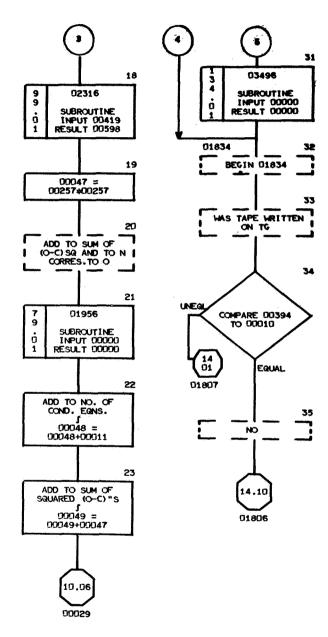


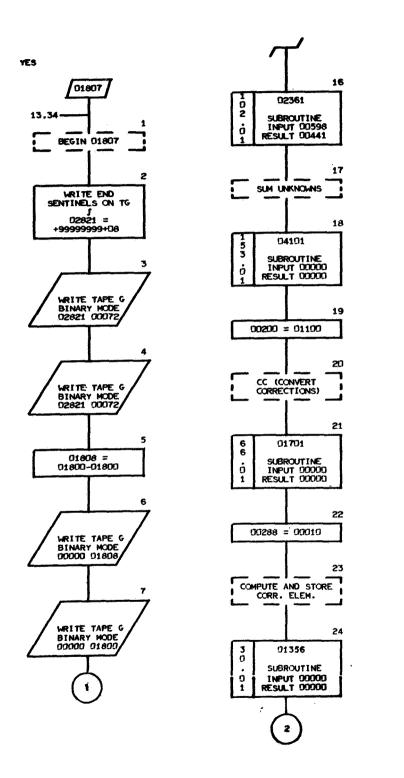


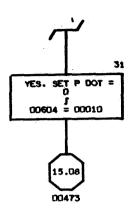


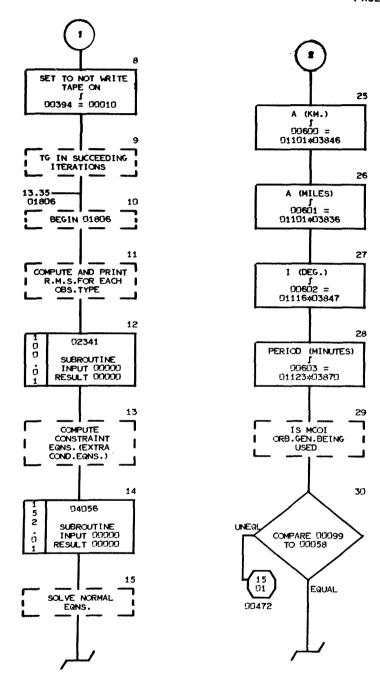
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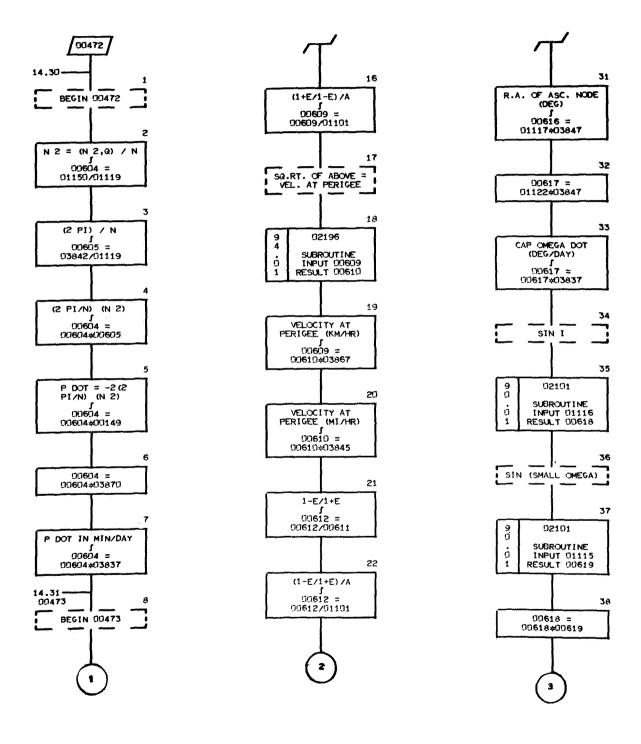




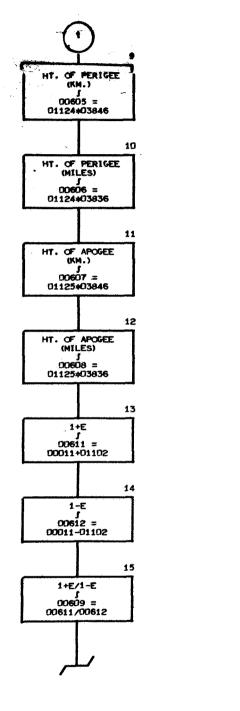


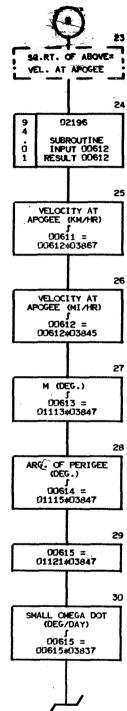


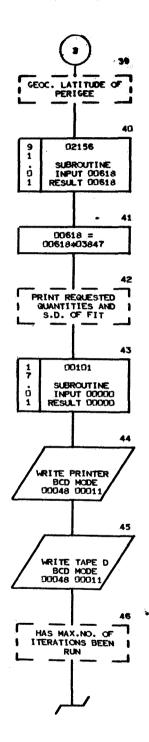


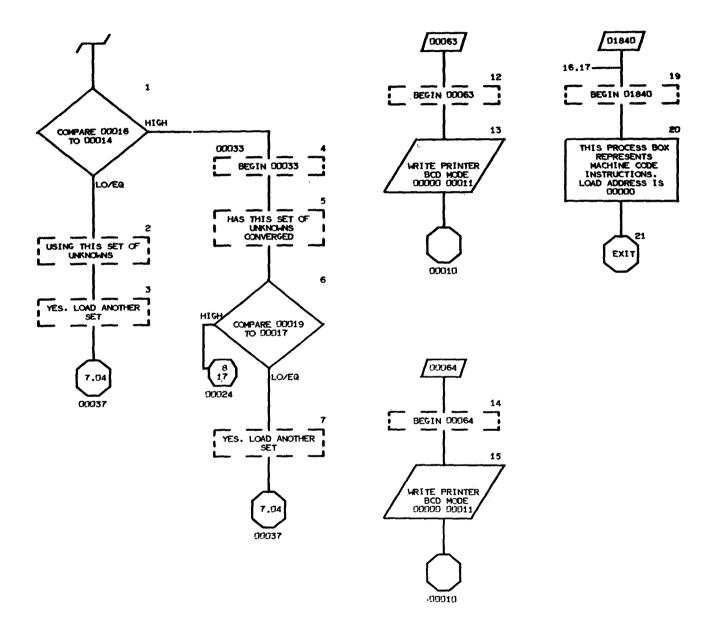


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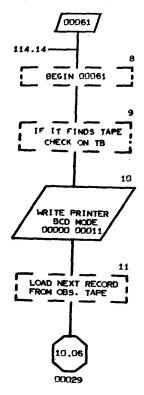




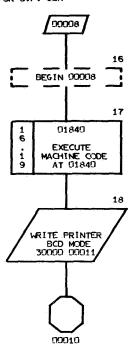




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OVER-UNDERFLOW TRANSFER FOR S.P.-32K



PAGE I

MAGA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI CROSS-REFERENCE LISTING

CROSS-REFERENCE LISTING					
PAGE BOX	LABEL	RE	FERENCES		
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1.28	00570	1.21			
2.01	90577	1.25			
2.08	00576	1.30			
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7.15	00036	7.06			
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9.14	00481	9.10			
9.16	00091	9.13			
9.22	00480	9.20			

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10.17	00482	10.13	11.01	11.03	¥g.±.	
10.22	00463	11.04		*	# () #	
10.26	00491	10.24				
10.32	00492	10.25				
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11.05	02094	10,40				
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16.14	00064					
16.16	00008					
16.19	01840	16.17*				

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Q9004103551 PRINT INITIAL ELEM., DRAGS, EARTH CON Q9004303601 LOAD AREA, MASS, DRAG DATA CARD FOR MC

Q9004503801 COMPUTE CONSTANTS FUNCTION

99004602341 COMPUTE AND PRINT R.M.S. FOR EACH OB

Q9006604401 COMPUTE EFFECTIVE DRAG

Q9004701956 ADD TO SUM OF (O-C) SQ. AND TO N COR Q9004904056 COMPUTE CONSTRAINT EQNS. (ADDIT.EQC)

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 (O-C) ANGLE REDUCTION F.

 G9005803496
 INITIALIZE INTERPOLATION

 G9005901356
 COMPUTE AND STORE CORR. ELEM.

 G9006002476
 LOAD AND STORE PRINT REQUEST CARDS

 G9006100101
 PRINT CORR. ELEM. AND 9.D. OF FIT

Q9006204101 SUM UNKNOWNS

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 O-C PRINT FUNCTION

 Q9006402751
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 RANGE RATE FUNCTION

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G9006803401 DELTA TAPE READ AND INTERP.

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 INPUT CONVERTER

 Q9007102521
 OUTFUT SCALE

 Q9007203901
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Q9007401941 SATELLITE IDENTIFICATION

Q9007503301 DAY COUNT

Q9007603351 OBSERVED DATE TO J.D.
Q9007704491 ORBIT GENERATOR INITIALIZE

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 ELEMENT LOAD

 Q9008102301
 MATRIX CLEAR

 Q9008202801
 OBSERVATION LOAD

99008302418 SPO

Q9008404501 ORBIT GENERATOR

Q9008501201 LOCAL STATION PREDICTION
Q9008601301 POSITION IN ELLIPSE

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e9008801601 OBSERVATION PARTIALS

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 AUGMENT MATRIX

 99009002361
 SOLVE EQUATIONS

 99009101701
 CONVERT CORRECTIONS

 99009303501
 LOAD REJECT CARDS

99009403521 SEARCH AND REJECT FUNCTION

99009602196 SQUARE ROOT

Q9009702101 SIN Q9009802156 ARC SIN

99009904331 LOAD INTERVAL CORE DUMP CARDS

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 NORMAL SETTING, FOR NO JUMP AT END OF

 V00461+160000000+02
 NO. WORDS IN PERT. TAPE TITLE RECORD

 V00615+100000000+01
 K, MULTIPLIER FOR DRAG DELTA M

 V00459-100000000+01
 NO. STORED IN LOC.00000 IF TAPE CHECK

 V00460+100000000+02
 NO. OF TIMES TO TRY TO READ TAPE REC

 V00465+100000000+02
 MAXIMUM NO. OF INTERVAL CORE DUMPS

 V00466+100000000+01
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 V00475+000000000+00
 NORMAL SETTING, FOR (O-C) "S PRINTED

V00498+10000000+03

 V00495+280000000+02
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 V02098+10000000+01
 SET FOR RANGE RATE (O-C), S NOT DIFF.

V90002+30000000+01 -

V00003+500000000+01

V00007+50000000+02

V00009+100000000+02

V00010+000000000+00 ZERO

V00011+10000000+01

V00012+19000000+02 NO. OF ELEMENTS

V00013+180000000+02 MAX. NO. OF VARIABLES
V00015+500000000+01 NO. OF WORDS/ELEMENT NAME

V00016+10000000+02

V00017+10000000-02

V00018+10000000+06

V00058+200000000+01

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W00086 \*

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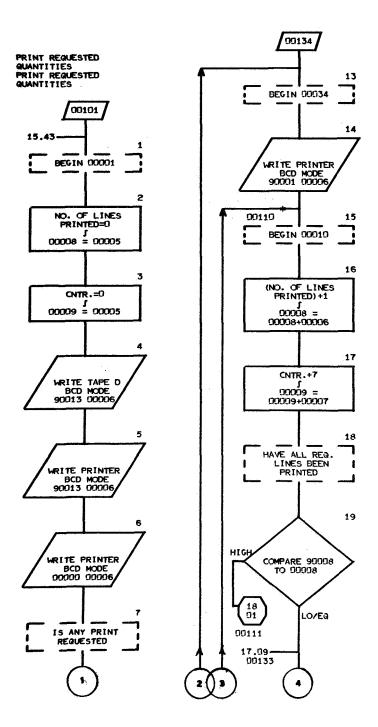
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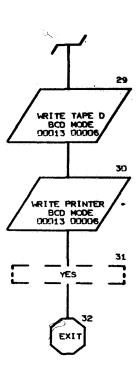
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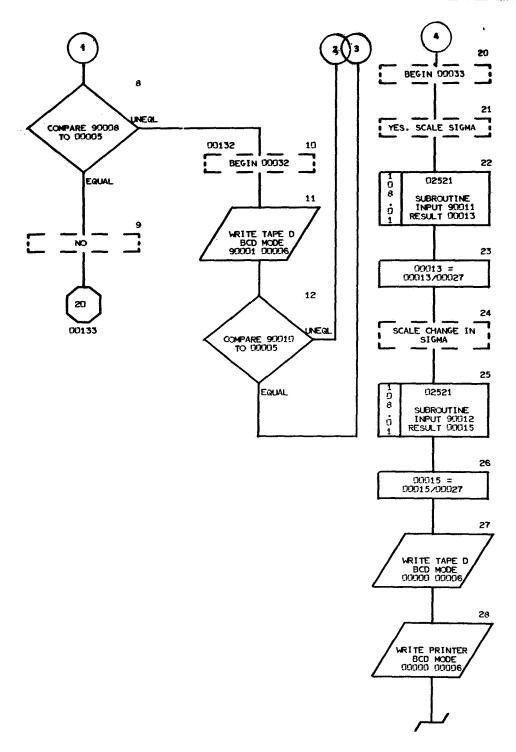
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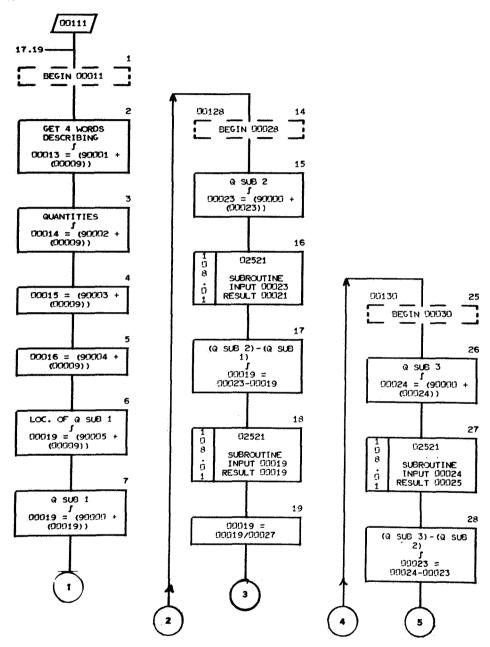
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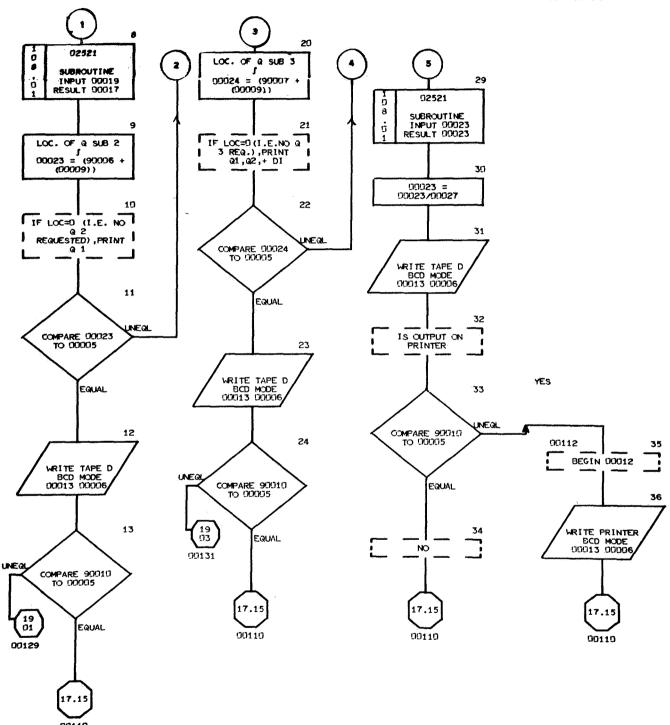


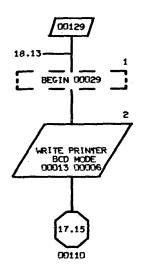


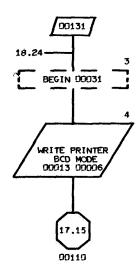
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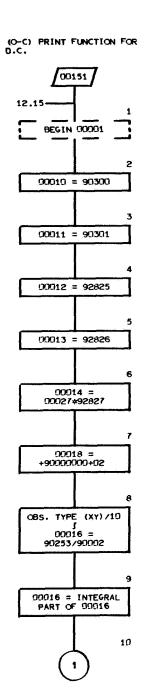
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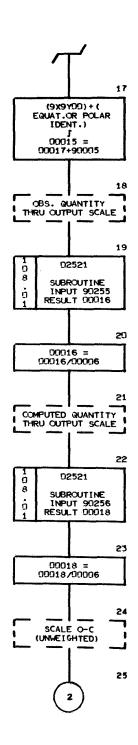
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Q9000503104	5 TH STG. LOC.
<b>Q9000603105</b>	6 TH STG. LOC.
Q9000703106	7 TH STG. LOC.
99000800060	NO. OF LINES TO PRINT
99000902521	OUTPUT SCALE
99001000042	PRINTER OUTPUT OPTION
99001100178	SIGMA
99001200065	CHANGE IN SIGMA
99001300179	ITERATION NO.
V90005+000000000+00	
V90096+10009000+91	
V00007+700000000+91	

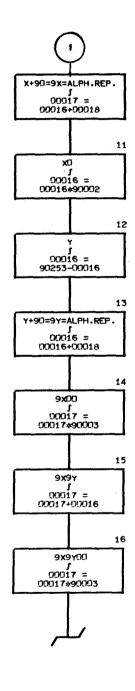
V00027+10000000+06

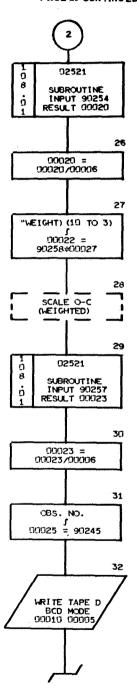
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17.10	00132	17.08						
17.13	80134	17.12		a "			46.00	19.04
17.15	00110	17.12	18.13	18.24	18.34	18.36	19,02	
17.20	00133	17.09						
18.01	00111	17.19						
18.14	00128	18.11						
18.25	00130	18.22						
18.35	00112	18.33						
19.01	00129	18.13						
19.03	90131	18.24						

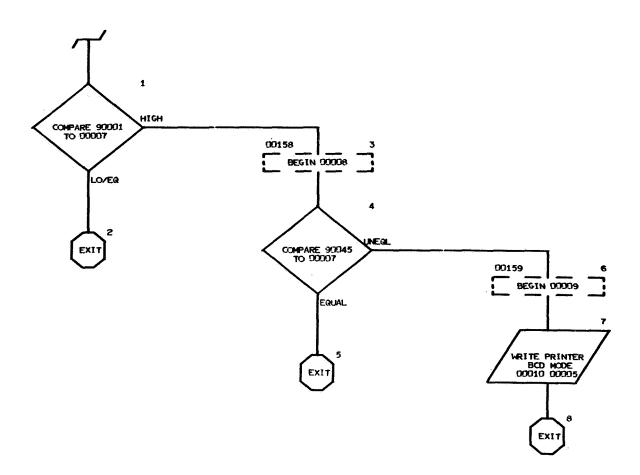




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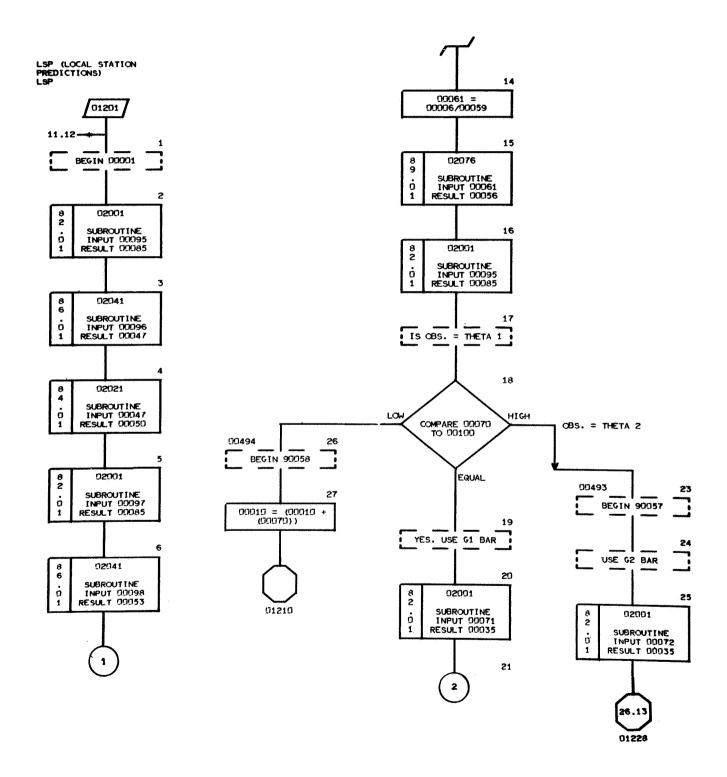




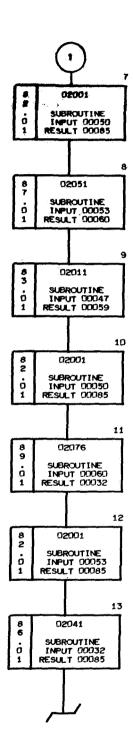


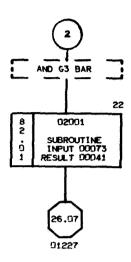
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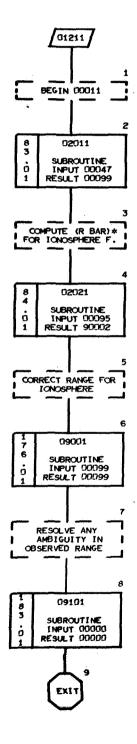
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21.06	00159	21,04
	LUE = 001	50
Q9000100042		
9900200009		TEN
Q9000300498		100
Q9000500 <b>49</b> 7		EQUATORIAL OR POLAR IDENT.
.09004500045		·
99030000300		STATION LABEL
29030100301		
Q9282502825		YR MO DY
99282602826		HR MI
99282702827		SE. SEC
99025300253		OBS TYPE
99025500255		OBSERVED QUANTITY
<b>Q9025600256</b>		COMPUTED QUANTITY
99025400254		O-C (UNMEIGHTED)
99025700257		O-C (WEIGHTED)
99025800258		OBSERVATION WEIGHT
99024500245		OBS. NO.
Q9252102521		OUTPUT SCALE
V00005+10000	0000+01	
V00006+10000	0000+06	
V00007+00000	0000+00	
V00027+10000	0000+04	10 TO 3
,		

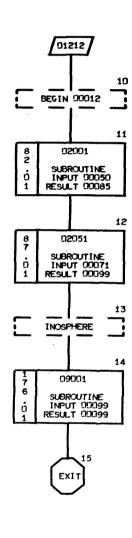


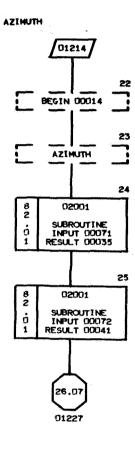
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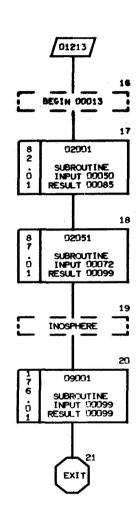


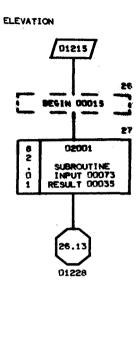


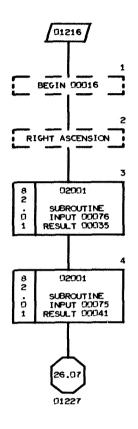


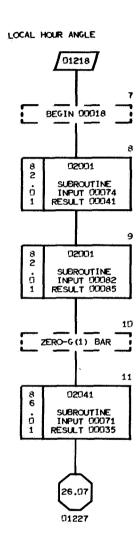


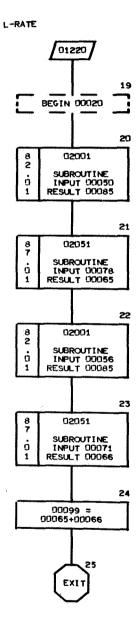




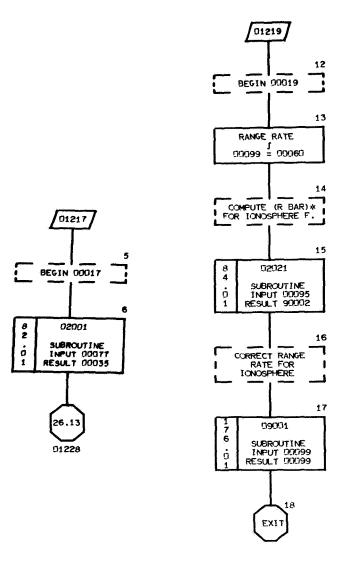




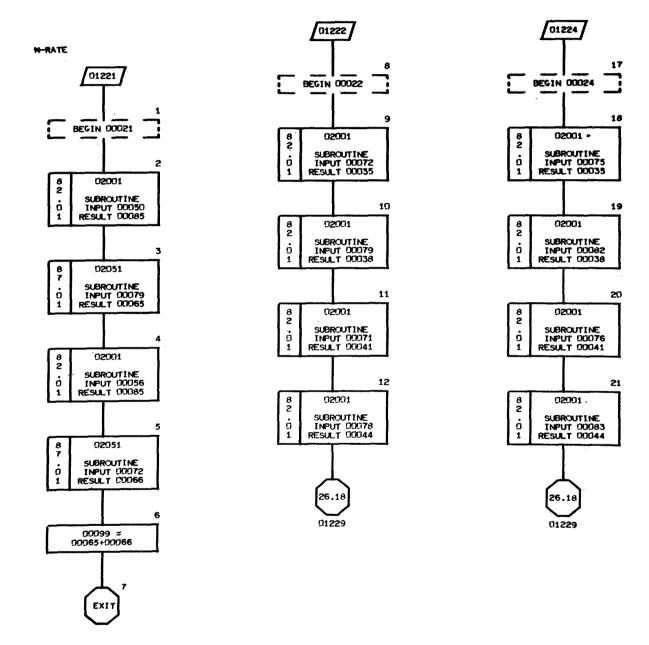




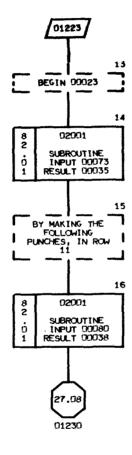
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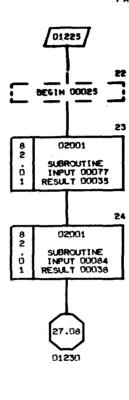


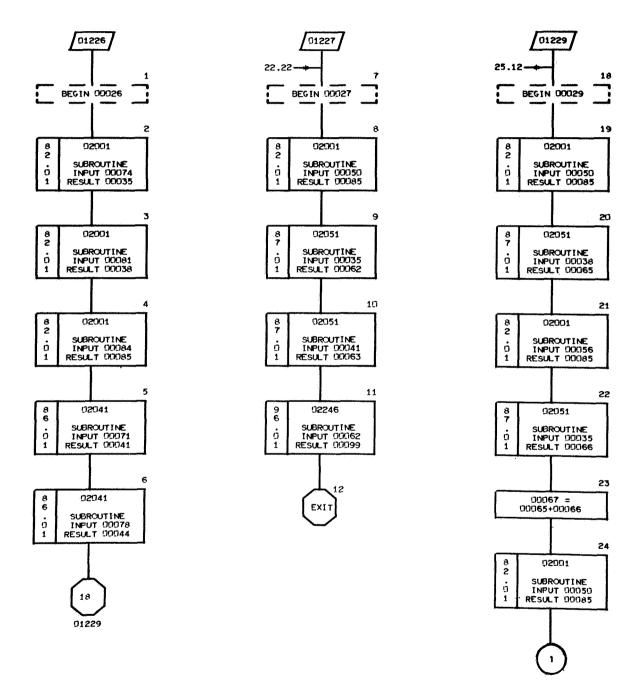
# NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI K VALUE = 01200

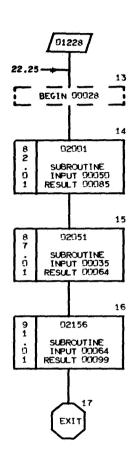


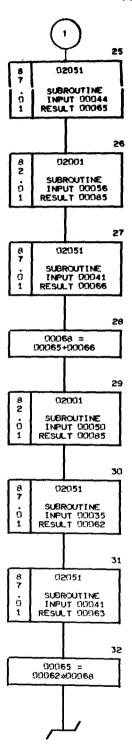
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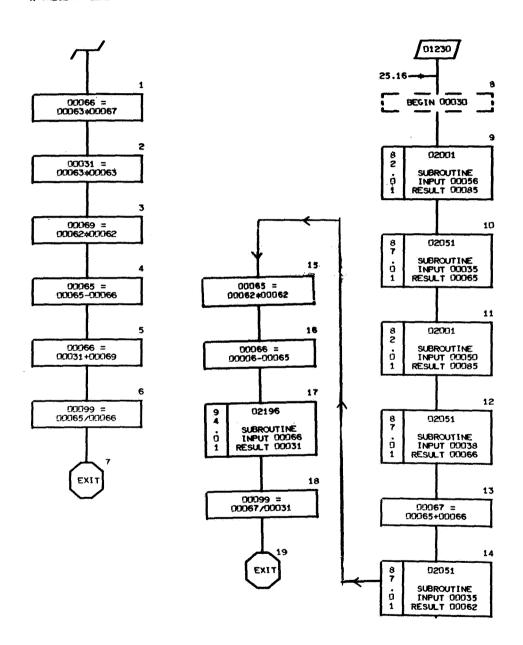












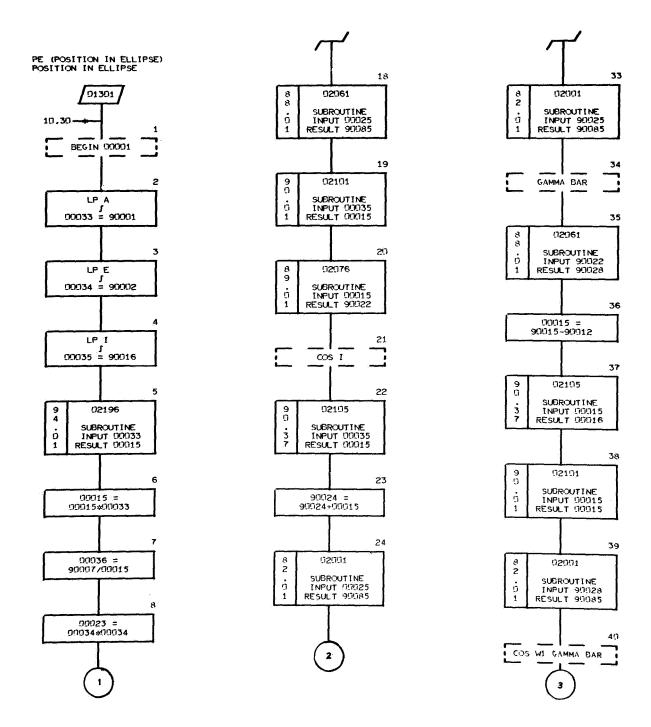
# NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI

99005700493 WORKING STG. (BEGIN COMMAND) Q9005800494 WORKING STG. (BEGIN COMMAND) 90004700358 RHO BAR 00005000361 (RHO BAR) \* @0005300365 RHO DOT BAR 00005600368 (RHO DOT BAR) \* 00005900364 RHO 000060000371 RHO DOT 00007000253 OBSERVATION TYPE @0007100316 G 1 BAR 00007200319 G 2 BAR 00007300322 .G 3 BAR QQQQ74QQ325 G 4 BAR 00007500328 G 5 BAR 00007600331 G 6 BAR Q0007700334 G 7 BAR G 1 DOT BAR Q0007800337 G 2 DOT BAR 00007900340 90008000343 G 3 DOT BAR G 4 DOT BAR 90008100346 00008200349 G 5 DOT BAR G 6 DOT BAR @0008300352 G 7 DOT BAR 00008400355 VQ 00008502085 00008602001 VECTOR MOVE DOT PRODUCT Q0008702051 90008802011 VECTOR MAGNITUDE VECTOR SUBTRACT 00008902041 VECTOR DIRECTION 00009002021 SCALAR-VECTOR MULTIPLY 00009102076 00009202246 ARC TAN (Y/X) @0009302156 ARC SIN SQUARE ROOT 00009402196 R BAR, SATELLITE POSITION VECTOR 00009500204 CAP R BAR, STATION POSITION VECTOR 00009600310 00009700207 R DOT BAR, SATELLITE VELOCITY VECTOR CAP R DOT BAR, STATION VELOCITY VECTO 00009800313 00009900256 COMPUTED OBSERVATION IONOSPHERE CORR. FUNCTION 09000109001 09000200375 (R BAR) \* RANGE AMBIGUITY RESOLUTION F. 09000309101 V00100+17000000+02 V00005+000000000+00

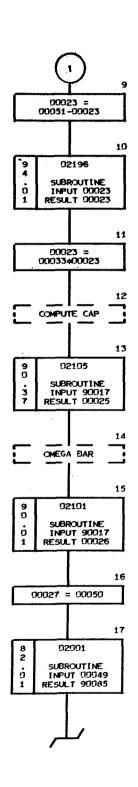
V00006+100000000+01

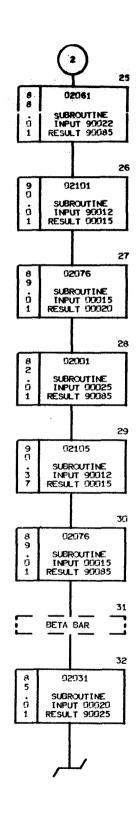
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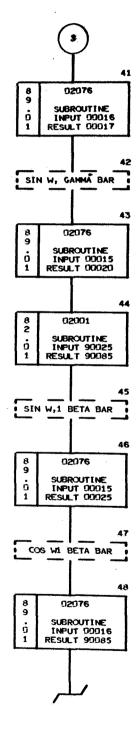
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22.26	00494	22.18				
23.01	01211					
23.10	01212					
23.16	01213					
23.22	01214					
23.26	01215					
24.01	01216					
24.05	01217					
24.07	01218					
24.12	01219					
24.19	01220					
25.01	01221					
25.08	01222					
25.13	01223					
25.1/	01224					
25.22	01225					
26.01	01226					
26.07	01227	22.22	23,25	24.04	24,11	
26.13	01228	22.25	23,2/	24.06		
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27.08	91230	25.16	25.24			

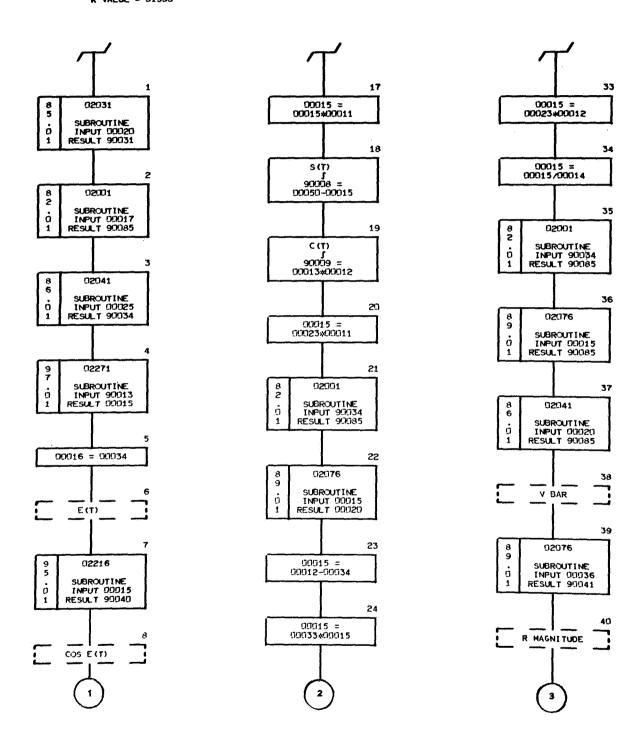


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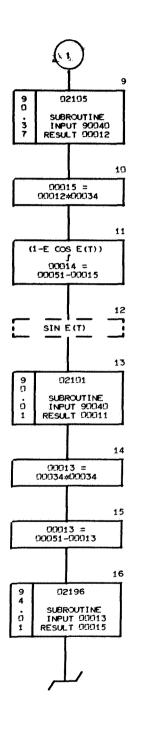


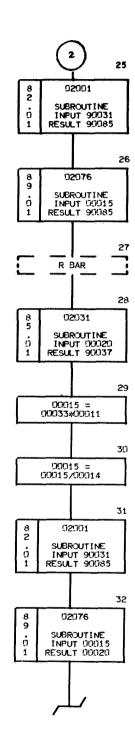


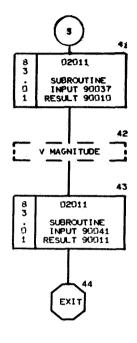




#### PAGE 29 CONTINUED







#### K VALUE = 01300

Q9000001100 T(0), EPOCH TIME IN C.U.T.

09000100218 SEMI MAJOR AXIS AT T, TIME OF OBS.

 Q9000200219
 ECCENTRICITY AT T

 Q9000301113
 MEAN ANOMALY AT T(0)

 Q9000401115
 ARG. OF PERIGEE AT T(0)

 Q9000501117
 LONG. OF ASC. NODE AT T(0)

 Q9000603864
 J

 Q9000703852
 MU

 Q9000800201
 S(T)

 Q9000900202
 C(T)

Q9001000210 R (MAGNITUDE OF R BAR)
Q9001100211 V (MAGNITUDE OF V BAR)

 Q9001201112
 DELTA AT T (0)

 Q9001300213
 MEAN ANOMALY AT T

 Q9001500215
 ARG. OF PERIGEE AT T

 Q9001600220
 INCLINATION AT T

99001700217 LONG. OF ASC. NODE AT T

Q9001901119 MEAN MOTION AT T(0)

 Q9002200222
 ALPHA
 BAR

 Q9002400224
 ALPHA
 SUB
 K

 Q9002500225
 BETA
 BAR

 Q9002800228
 GAMMA
 BAR

 Q9003100231
 P
 BAR

 Q9003400234
 Q
 BAR

 Q9003700237
 R BAR (POSITION VECTOR)

 Q9004000240
 E(T) (ECCENTRIC ANOMALY)

 Q9004100241
 V BAR (VELOCITY VECTOR)

 Q9005000200
 T, TIME OF OBS. IN C.U.T.

Q9008502085 VQ

 Q9008602001
 VECTOR MOVE

 Q9008702061
 CROSS PRODUCT

 Q9008802011
 VECTOR MAGNITUDE

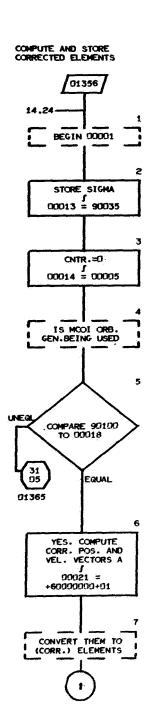
 Q9008902196
 SQUARE ROOT

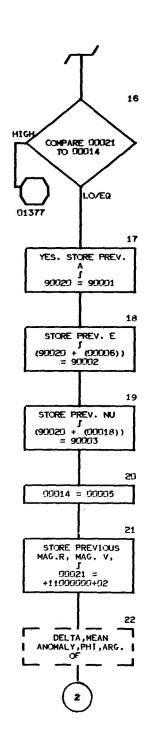
 Q9009002271
 PRINCIPAL VALUE

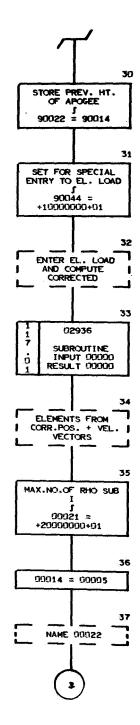
09009102101 SIN Q9009202105 ထန Q9009302216 KEPLER Q9009402031 VECTOR ADD 09009502041 VECTOR SUBTRACT SCALAR MULTIPLY 09009602076 V00007+500000000+01 V00008+400000000+01 V00049+0000000000+00 ZERO V00050+000000000+00 ZERO V00051+100000000+01 ONE

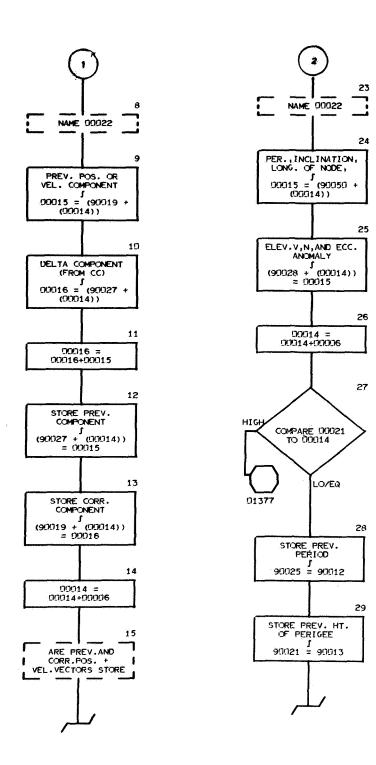
#### CROSS-REFERENCE LISTING.

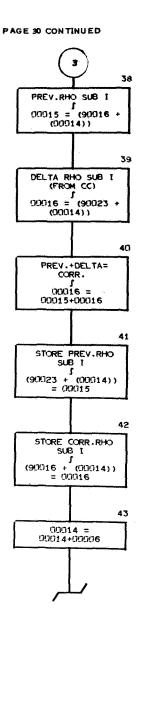
PAGE 90X LABEL REFERENCE 28.01 01301 10.304 86.274

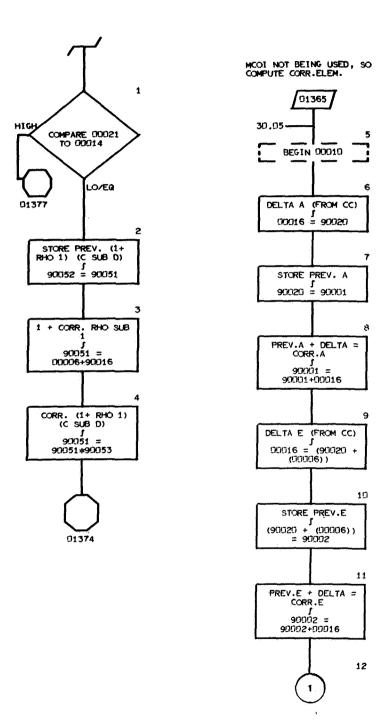


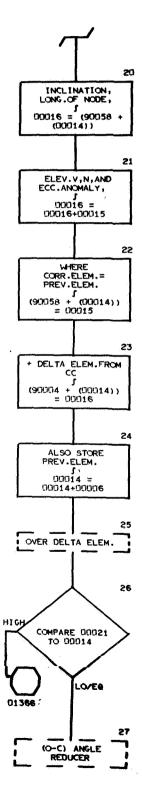




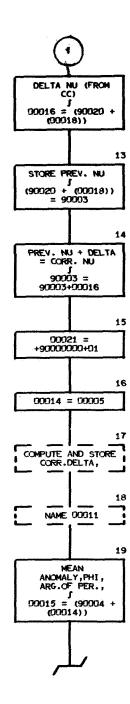








#### PAGE 31 CONTINUED



#### NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI MISCELLANEOUS STATEMENTS

K VALUE = 01355

 09000001100
 1ST OF 100 LOCS.FROM WHICH BIN. RECO

 09000101101
 A (PREVIOUS ELEMENTS, WHICH

 09000201102
 E REPLACED BY CORRECTED ELE

99000301103 NU THE FUNCTION)

Q9000401112 ANGLE DELTA

Q9000501113 M

Q9000601114 ANGLE PHI

Q9000701115 ARG. OF PERIGEE

Q9000801116

Q9000901117 LONG. OF NODE

Q9001001119

Q9001101120 ECC. ANOMALY Q9001201123 PERIOD

 Q9001301124
 HT. OF PERIGEE

 Q9001401125
 HT. OF APOGEE

 Q9001601150
 N(2,Q) OR RHO SUB I

Q9001701170 N3,Q Q9001801196 SIGMA

Q9001901104 FIRST LOC. OF POS. + VEL. VECTORS Q9002000500 DELTA A (REPL.BY PREV.A IN F.)

 Q9002100523
 PREV. HT. OF PERIGEE

 Q9002200524
 PREV. HT. OF APOGEE

Q9002300525 DELTA N(2,Q) OR DELTA RHO SUB I
Q9002400545 DELTA N3,Q (REPLACED BY PREV.N3,Q IN

Q9002500522 PREVIOUS PERIOD

Q9002700503 DELTA R BAR, V BAR (REPL.BY PREV. VECT Q9002800509 FIRST OF 11 LOC. OF PREV. ELEM.

Q9003003851 C Q9003103852 MU

09003200599 M (NO. OF UNKNOWNS)

Q9003300048

Q9003400049 SUM OF (O-C) SQ.

Q9003500178 SIGMA

Q9003600065 CHANGE IN SIGMA

Q9003700082 ABS. VALUE OF DELTA SIGMA

09003800019 (DELTA SIGMA)/SIGMA

Q9004002196 SQ.RT.

#### PAGE 4A CONTINUED

#### K VALUE = 01355

**Q9004102271** ANGLE RED. **Q9004202751** ABSOLUTE VALUE

 Q9004302936
 ELEMENT LOAD (CONVERSION OF ELEM.)

 Q9004400090
 IND. OF NORMAL OR SPECIAL ENTRY TO E

 Q9005001110
 FIRST OF 11 LOC. OF PREV. ELEM.

 Q9005100567
 CORR. VALUE OF (1+ RHO 1) (C SUB 0)

 Q9005200566
 PREV. VALUE OF (1+ RHO 1) (C SUB 0)

Q9005300267 C SUB D

Q9005402011 VECTOR MAGNITUDE F.

 09005501107
 PREV.VEL.VECTOR (REPL.BY CORR. IN F.)

 09005601111
 PREV.MAG.VEL. (REPL.BY CORR. IN F.)

 09005700510
 DELTA MAG.VEL. (REPL.BY PREV. IN F.)

 09005800511
 1ST LOC.OF 9 DELTAS (REPL.BY PREV.EL

Q9006904491 ORBIT GENERATOR INITIALIZE

Q9007004501 ORBIT GENERATOR
Q9007100200 OBS. TIME IN C.U.T.
Q9010000099 ORBIT GENERATOR IDENT.

Q0000803842 2 PI

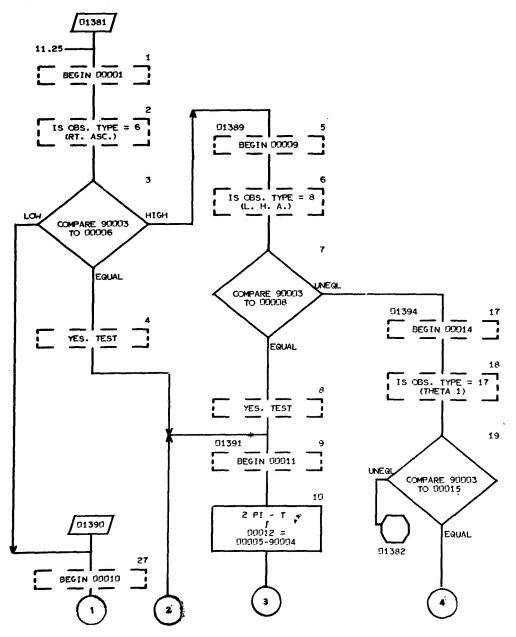
V00005+00000000+00
V00006+10000000+01
V00009-10000000+01
V00012+10000000+03
V00018+20000000+01

#### CROSS-REFERENCE LISTING

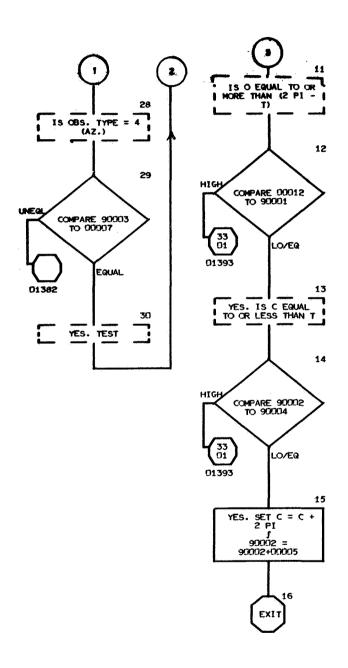
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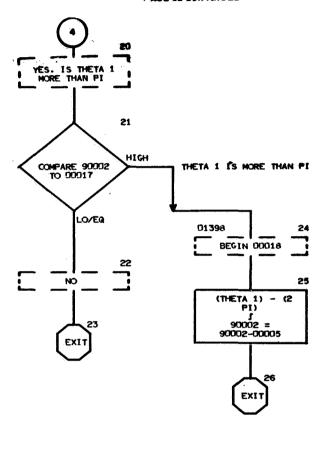
K VALUE = 01380 PAGE 32

## ANGLE REDUCER FOR ABS. VALUE OF (O-C)



#### PAGE 32 CONTINUED





01393 32.12-BEGIN 00013 2 IS O EQUAL TO OR LESS THAN T 3 HIGH COMPARE 90001 TO 90004 LO/EQ YES. IS C EQUAL TO OR MORE THAN (2 PI - T) 5 HIGH COMPARE 00012 TO 90002 LO/EQ

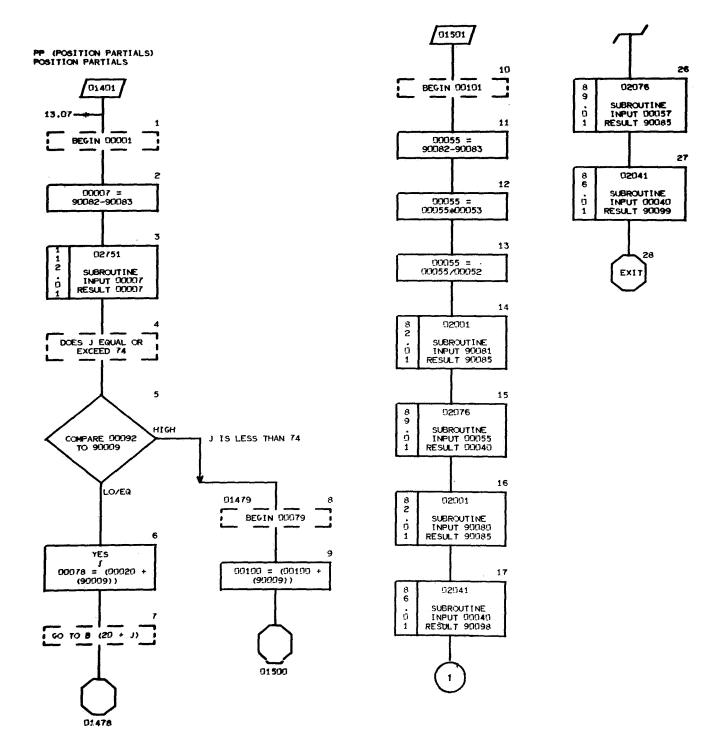
01382

YES. SET C = C -2 PI 90002 = 90002-00005

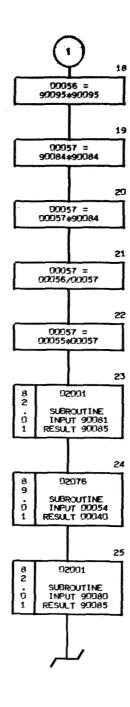
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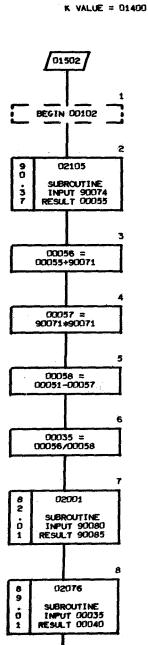
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32.09	01391	32.04	32.30
32.17	01394	32.07	
32.24	01398	32.21	
32.27	01390	32.03	
33.01	01393	32.12	32.14

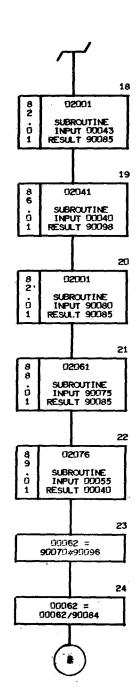
K VALUE = 01380	
Q9000100255	O (OBS. QUANTITY IN RAD.)
Q9000200256	C (COMPUTED QUANTITY IN RAD.)
Q9000300253	OBSERVATION TYPE
Q9000400496	T (TOLERANCE IN RAD.)
Q0000593842	2 PI
Q0001703839	PĪ
V00006+60000000+01	(PURPOSE IS TO REDUCE COMPUTED ANGL
V00007+490000009+01	WHEN (O-C) IS LARGE DUE TO DIFF.
V00008+80000000+01	AROUND 2 PI. ALL INPUT AND OUTPUT I
V00015+17000000+02	USES 18 LOCS.)

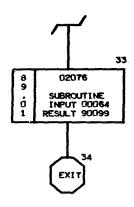


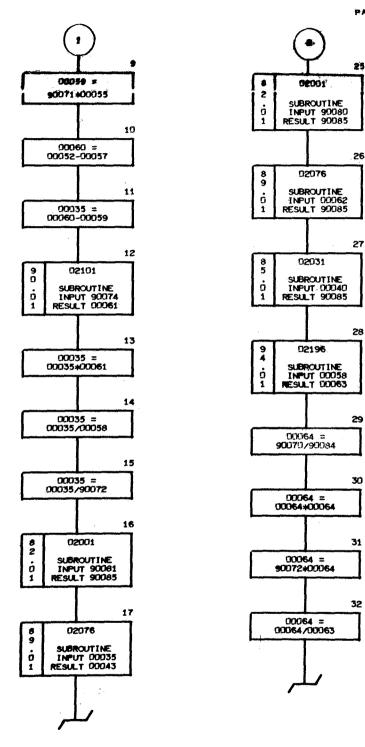
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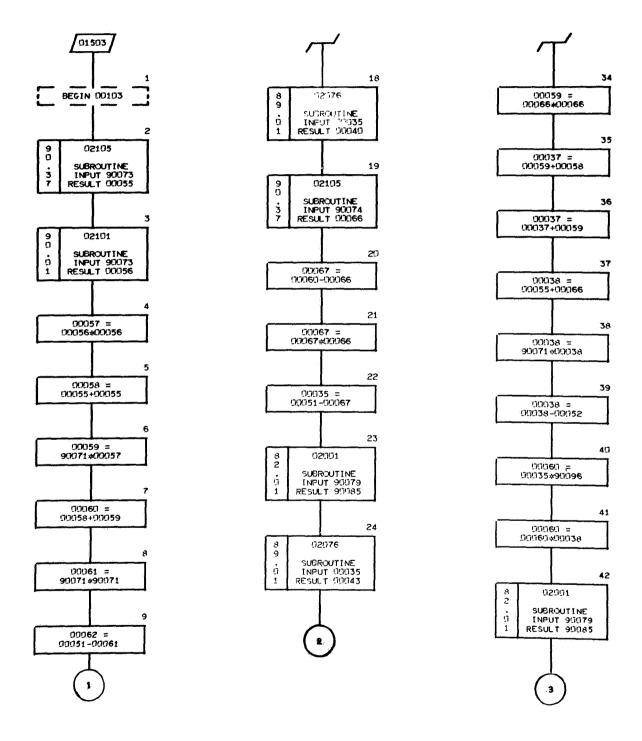




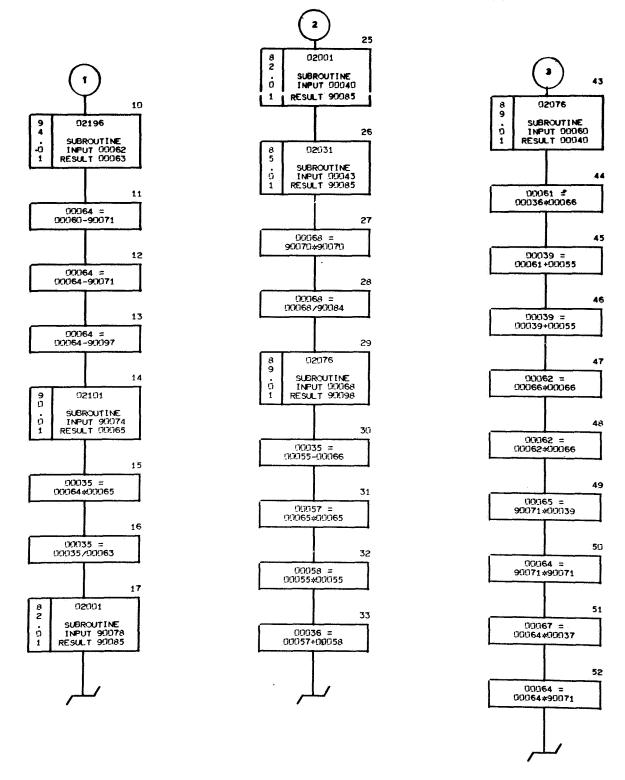


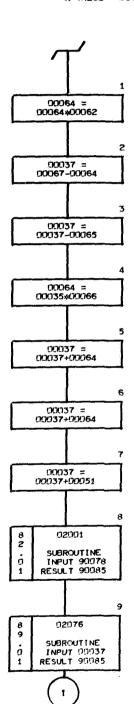


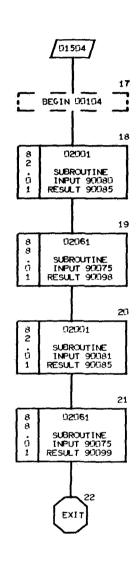


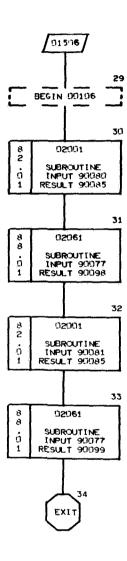


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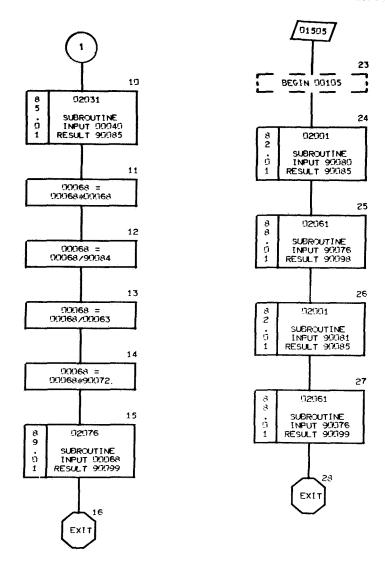


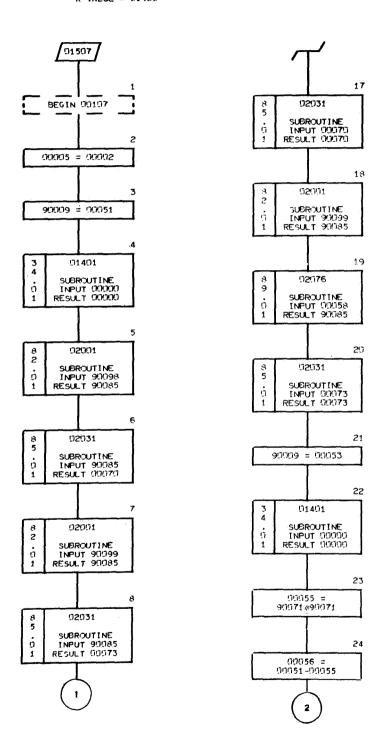


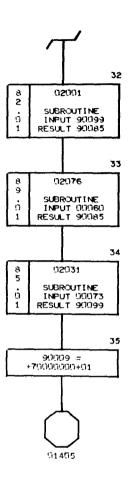




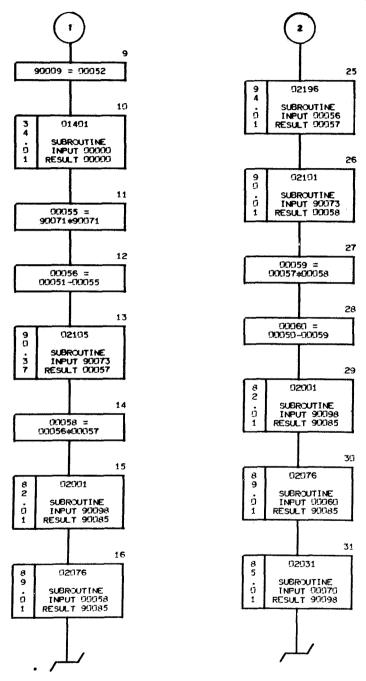
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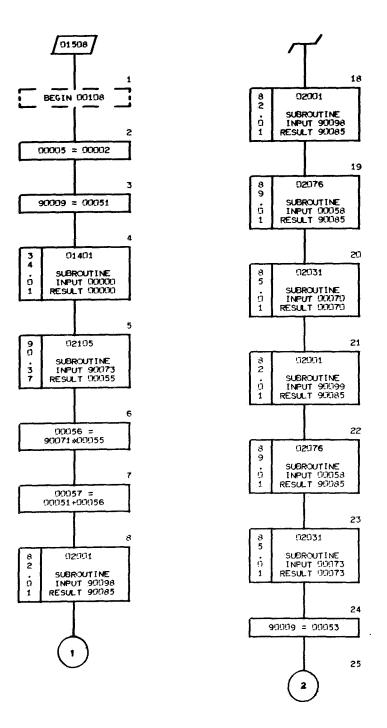


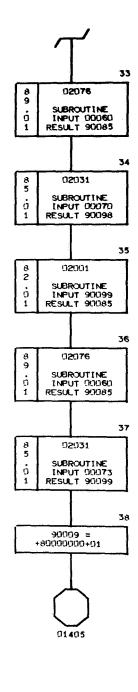


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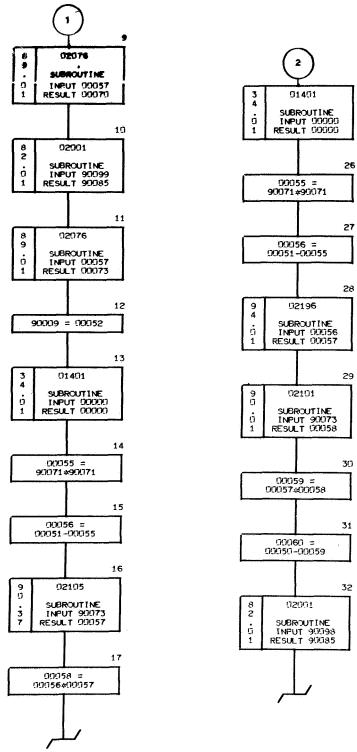


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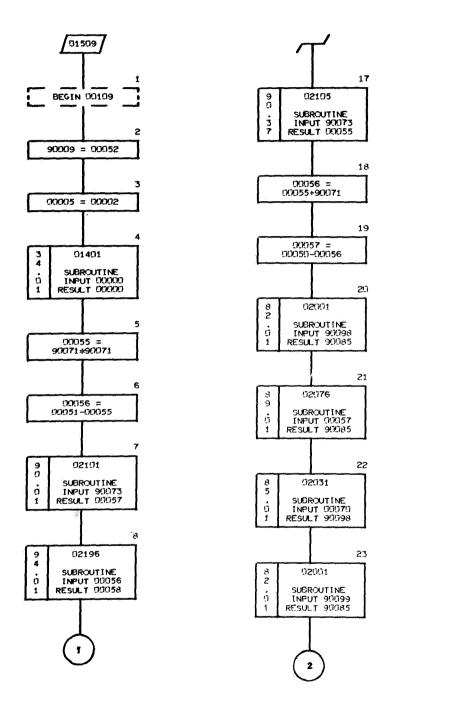


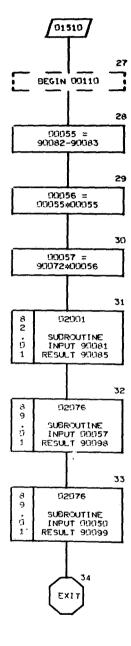


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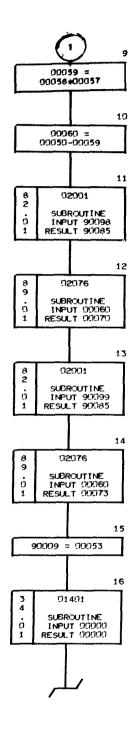


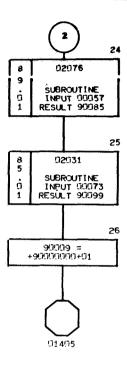
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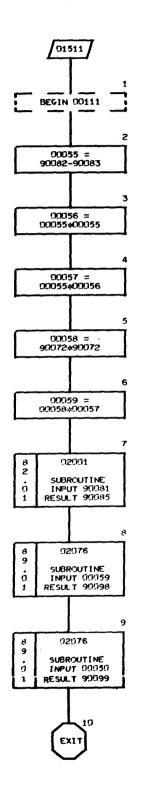


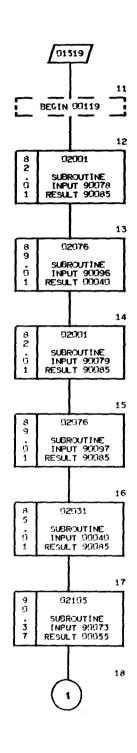


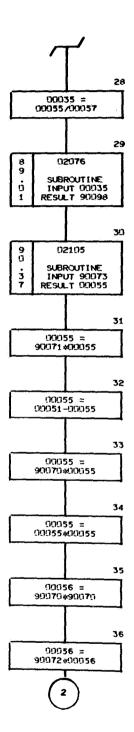
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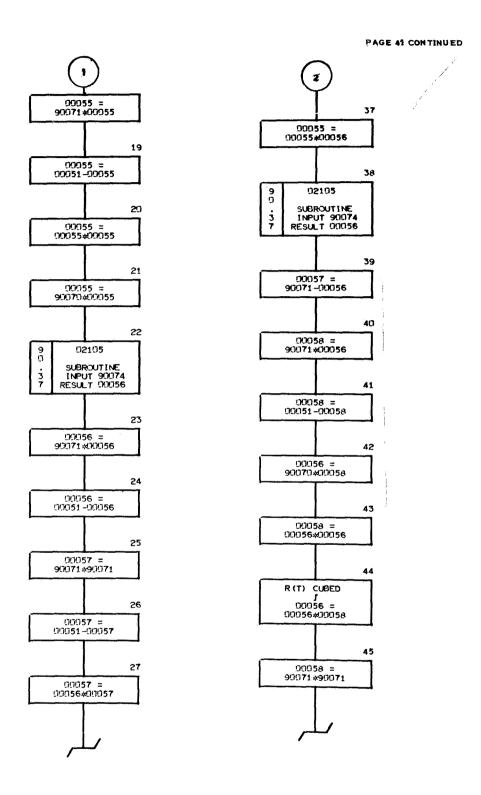


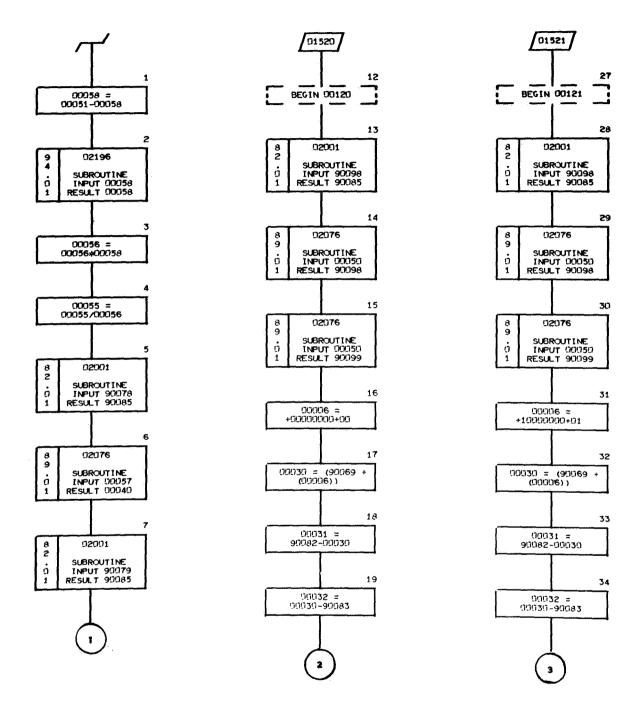




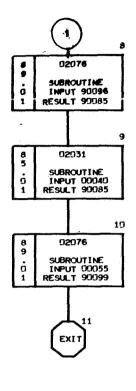


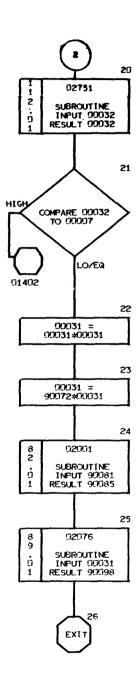


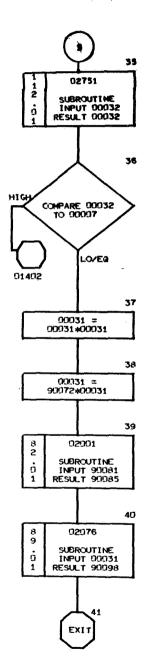


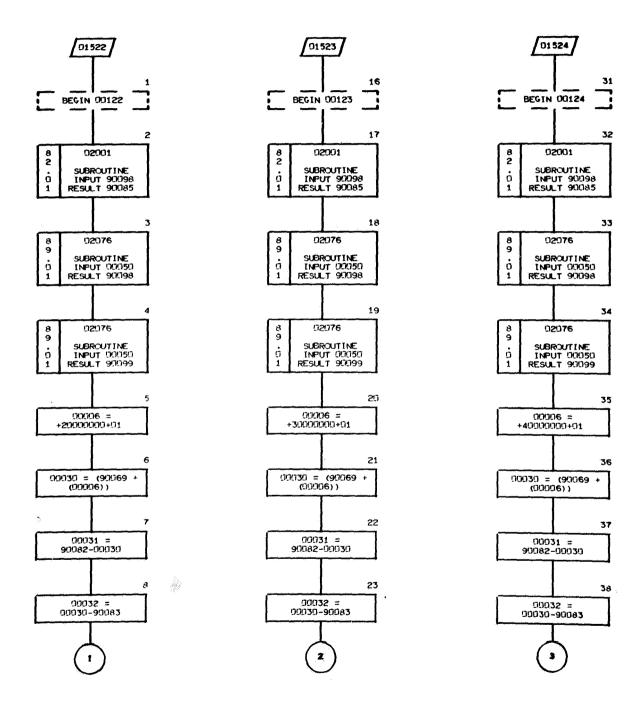


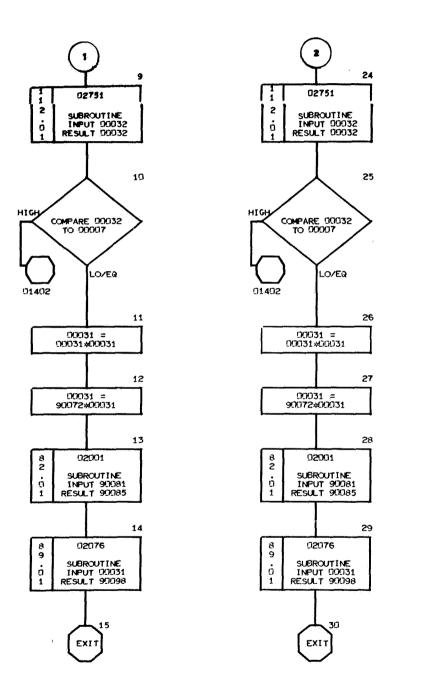
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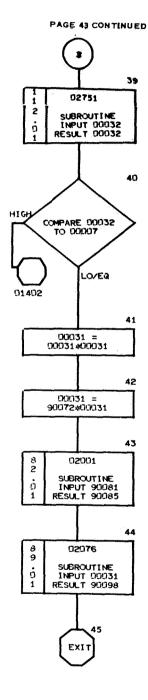


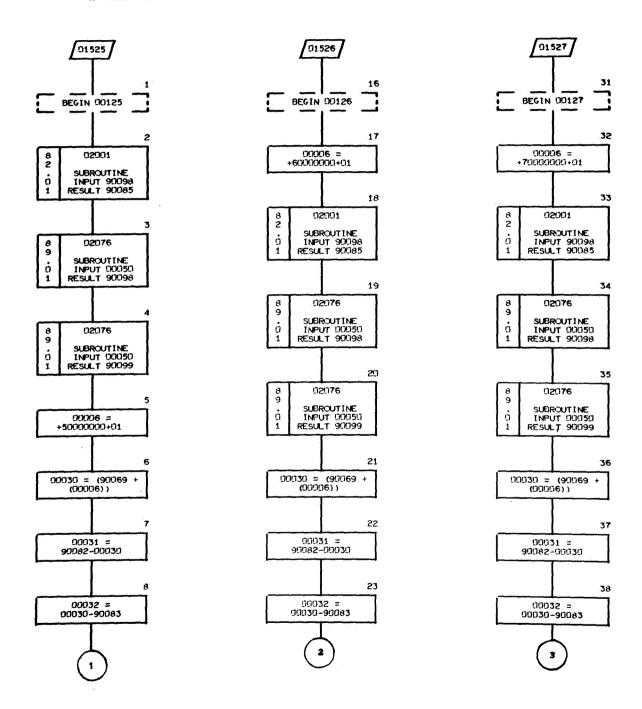


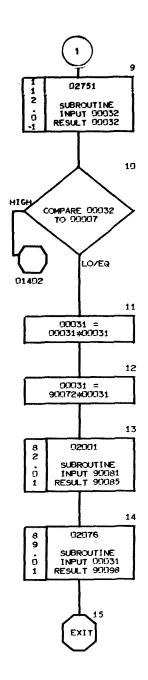


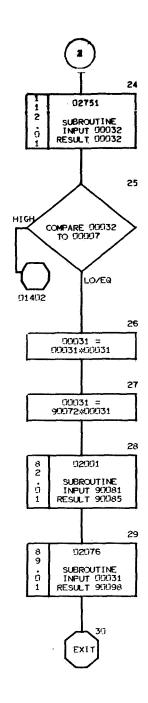


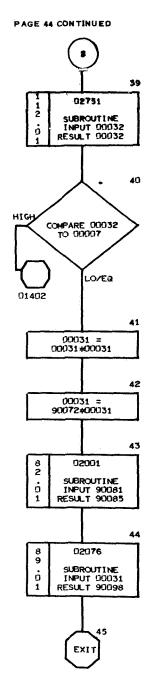


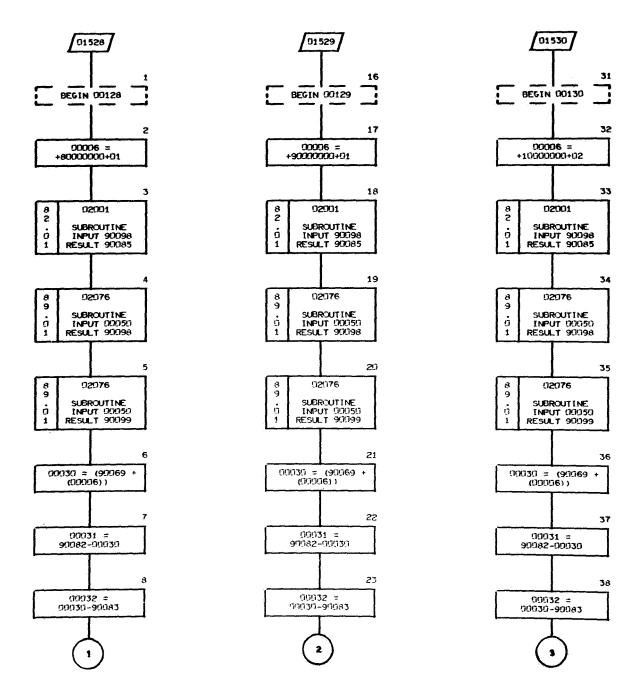


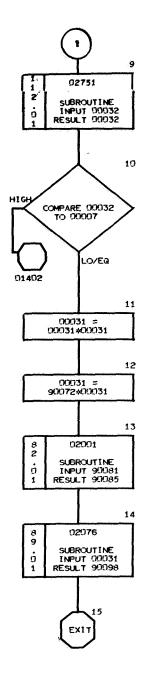


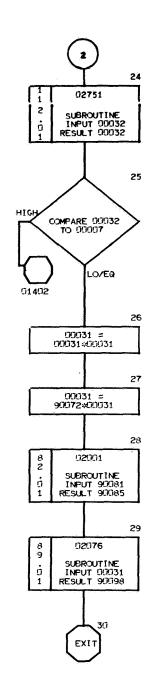


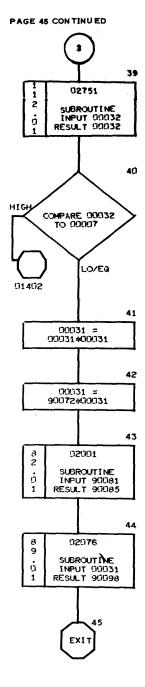


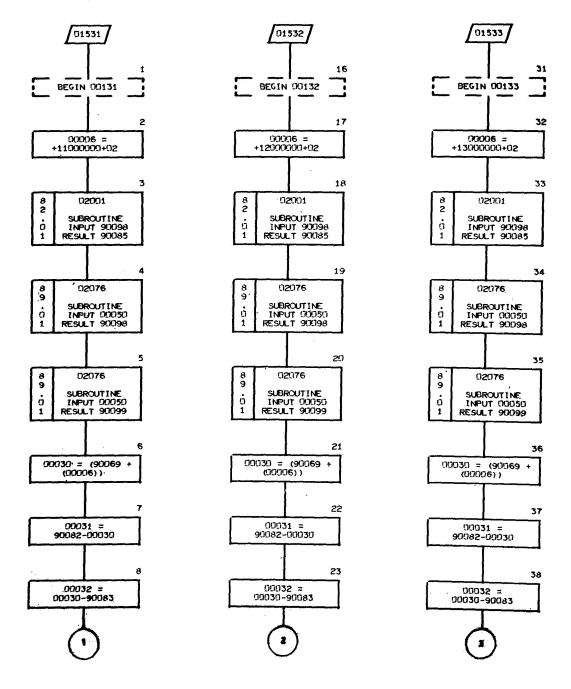


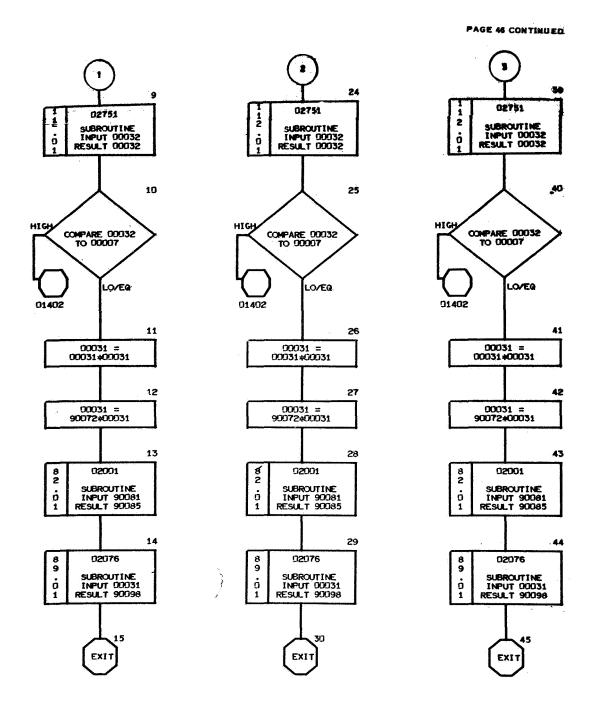




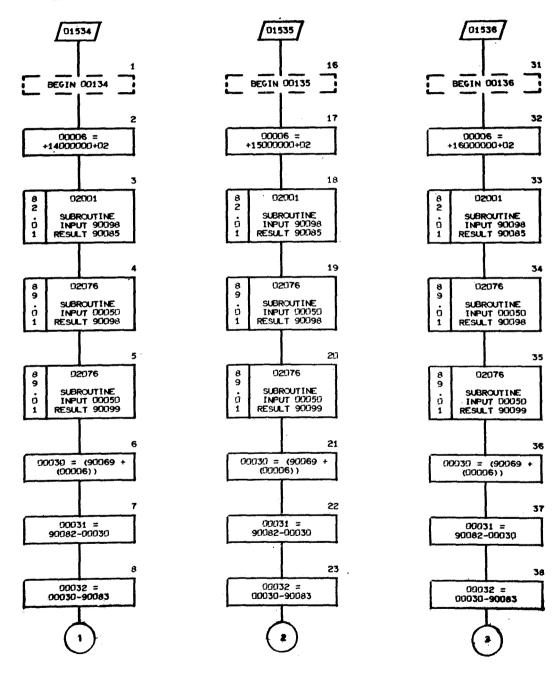


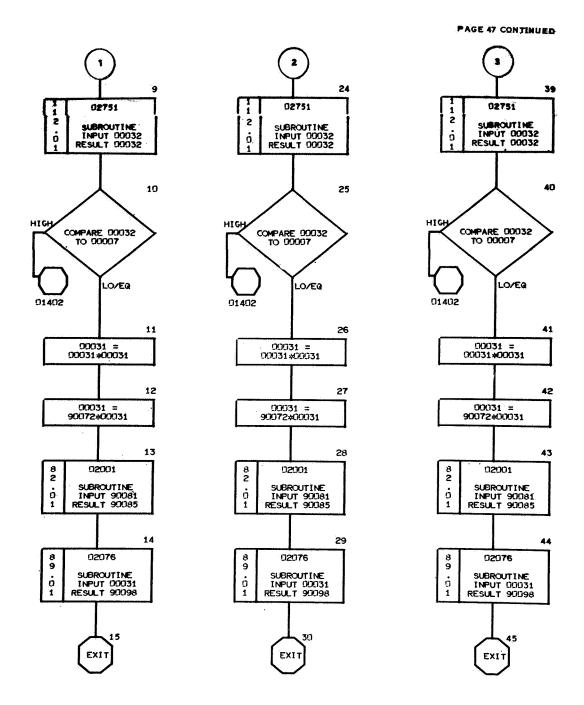


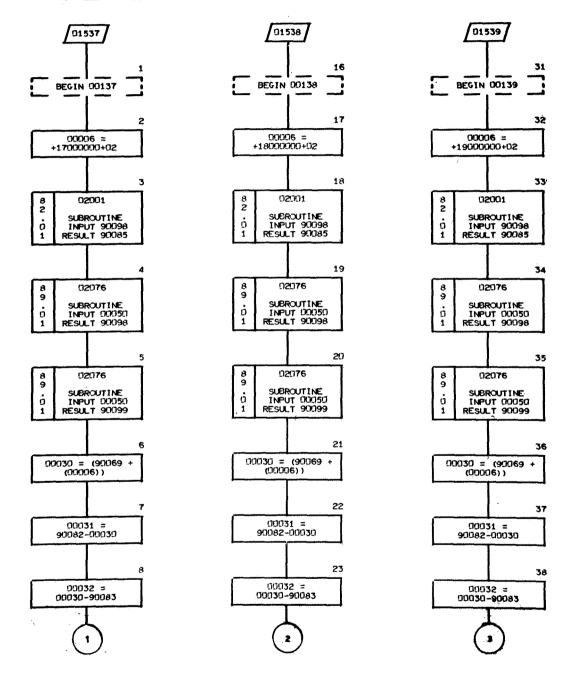


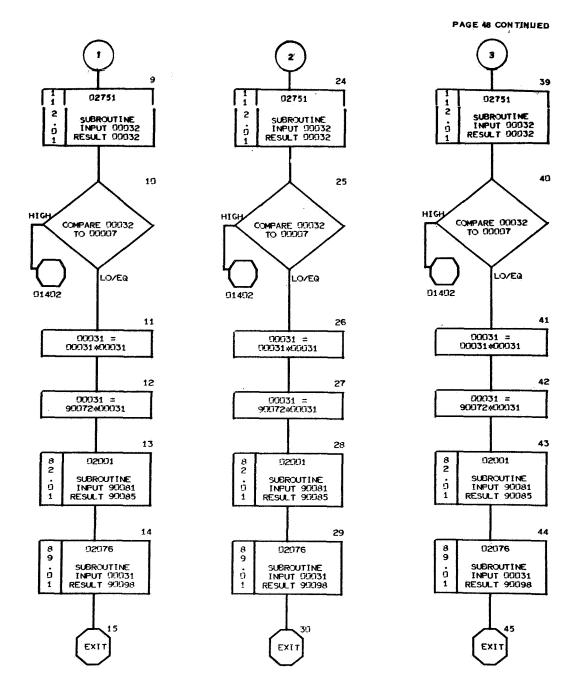


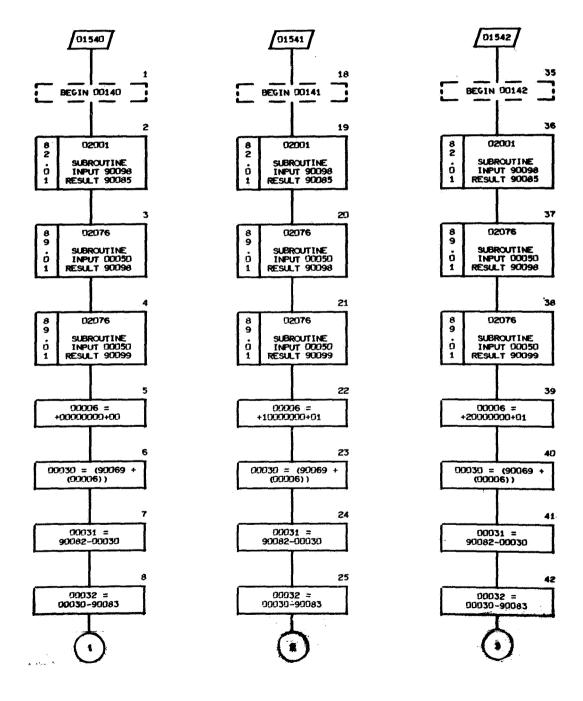
NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI

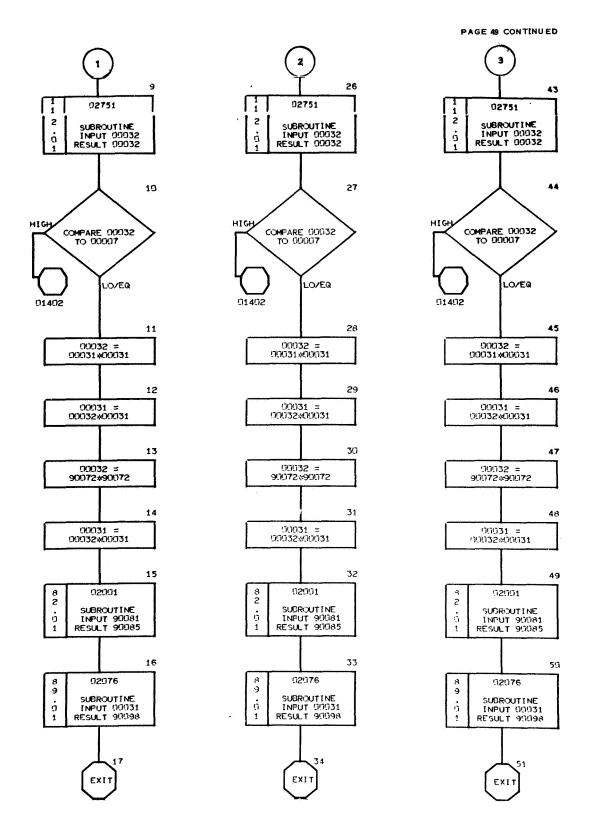




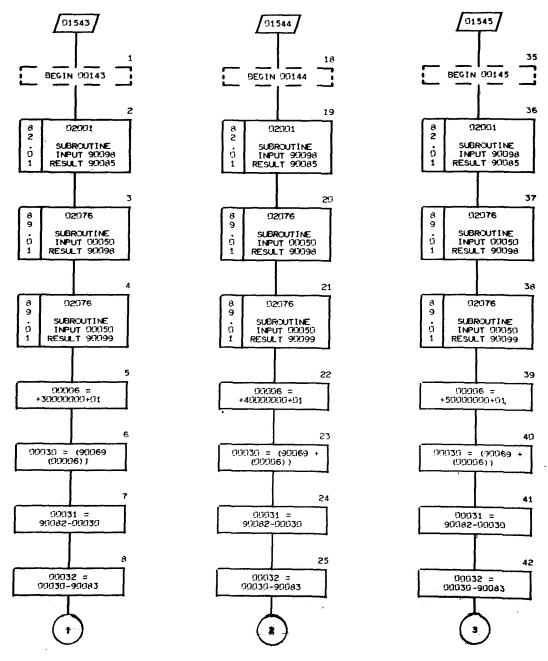


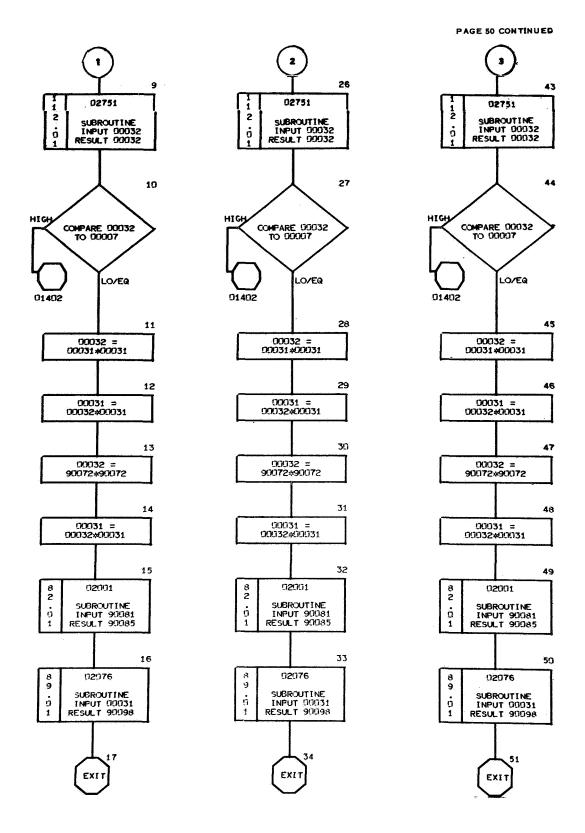


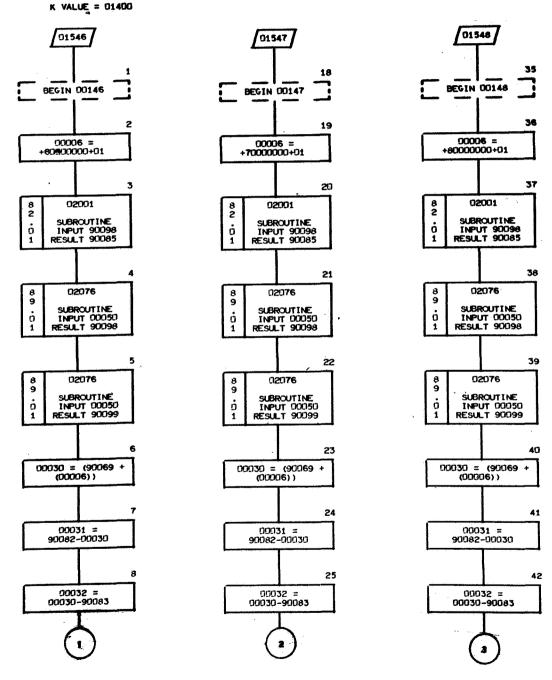


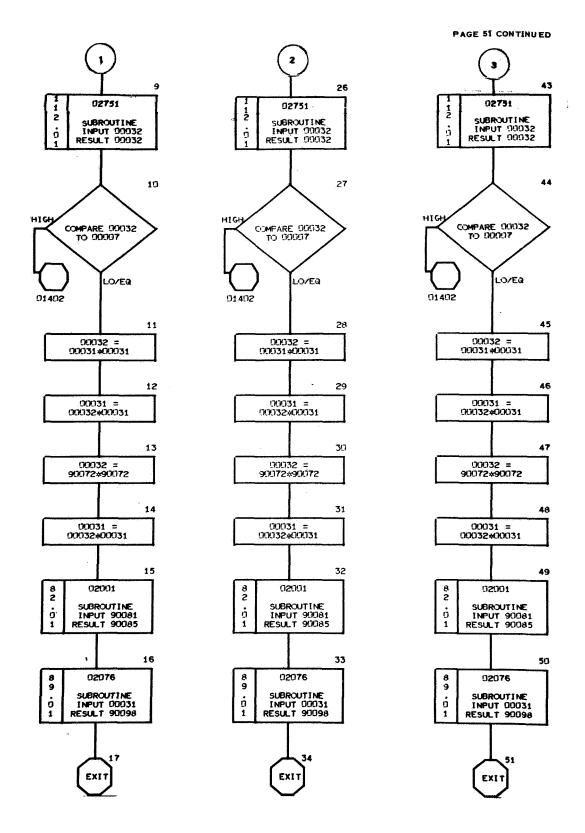


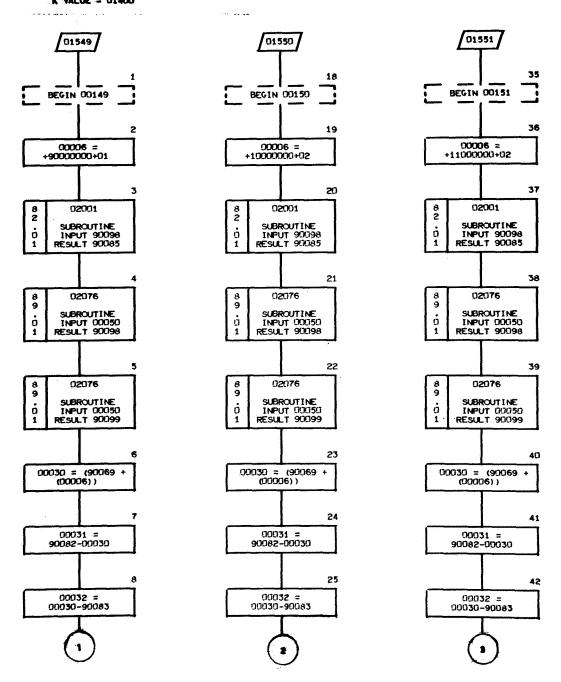


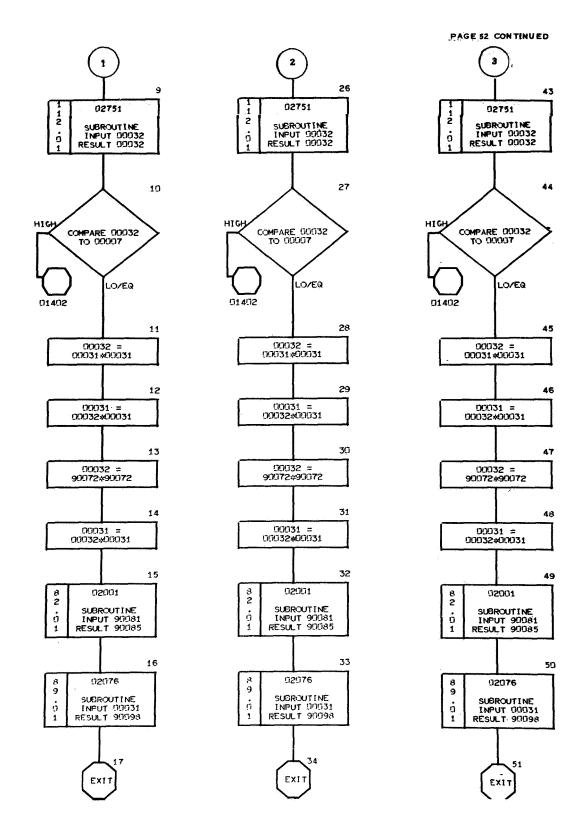


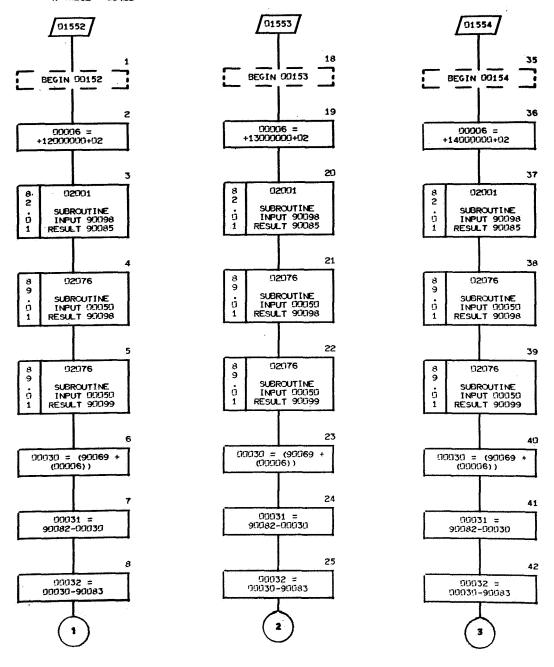


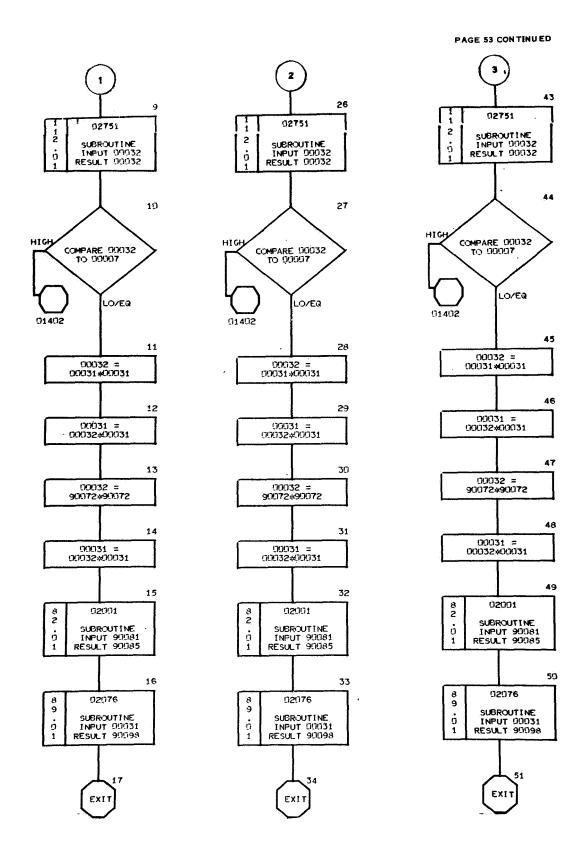


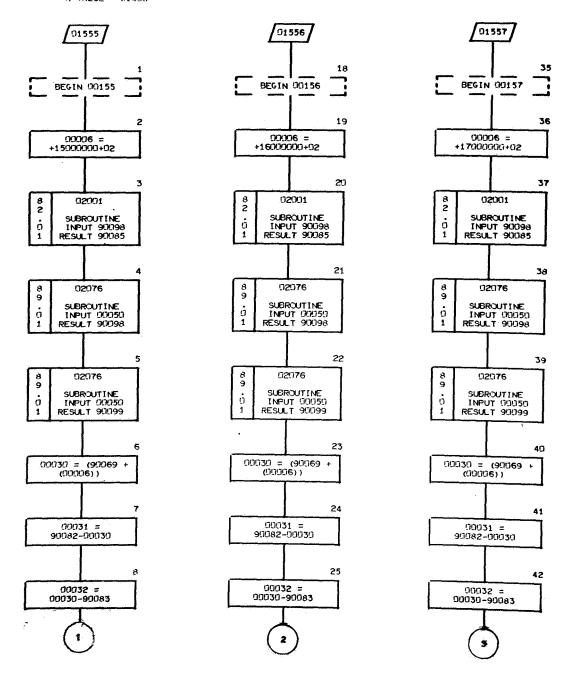


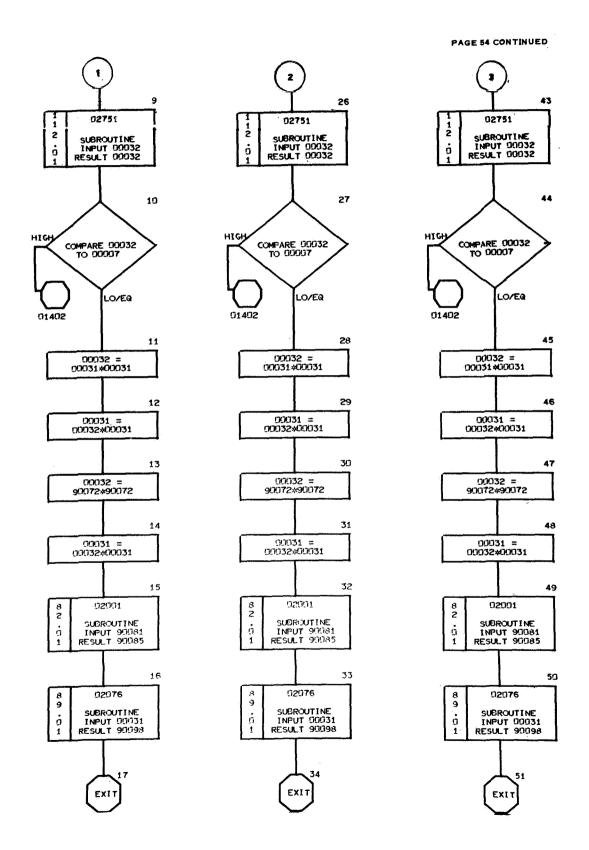


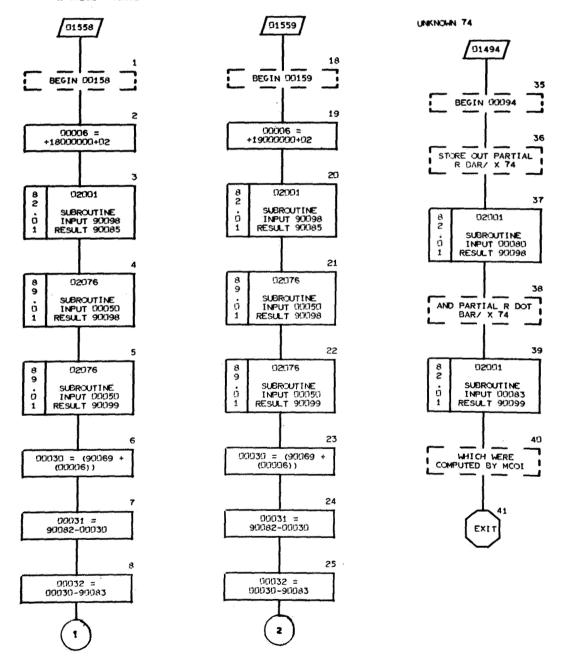




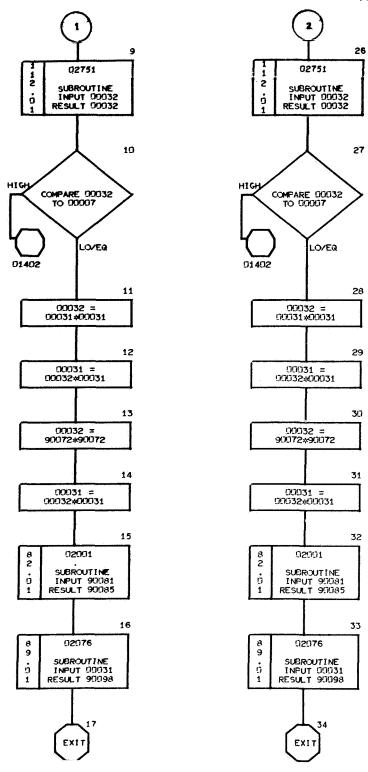








## PAGE 33 CONTINUED



## NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI K VALUE = 91400

## UNKNOWN 75 01495 BEGIN 00095 2 STORE OUT PARTIAL R BAR/ X 75 3 8 2 0 1 02001 SUBROUTINE INPUT 00086 RESULT 90098 AND PARTIAL R DOT BAR/ X 75 5 8 02001 SUBROUTINE INPUT DDD89 RESULT 90099 . 1 6 WHICH WERE COMPUTED BY MCOI

## NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI

#### MISCELLANEOUS STATEMENTS

## K VALUE = 01400

99000900252 J, NO. OF UNKNOWN 09006802751 ABSOLUTE VALUE FUNCTION FIRST LOC. OF T(P,Q)"S, TIMES OF DRA @9006901130 99007001101 A, SEMI-MAJOR AXIS AT T(0) 99007101102 ECCENTRICITY AT T(0) N, MEAN MOTION AT T(0) 09007201119 ECCENTRIC ANOMALY AT T(D) 99007301120 ECCENTRIC ANOMALY AT T Q9007400240 09007500222 ALPHA BAR BETA BAR 99007600225 09007700228 GAMMA BAR 09007800231 P BAR 09007900234 Q BAR Q9008000237 R BAR (COMPUTED BY POS. IN ELLIPSE) R DOT BAR (COMPUTED BY POS.IN ELLIPS 09008100241 T, OBSERVATION TIME IN C.U.T. 09008200200 T(0), EPOCH TIME IN C.U.T. Q9008301100 99008400210 MAG.OF R BAR (COMPUTED BY POS.IN E.) 09008502085 VQ 09008602001 VECTOR MOVE 09008702076 SCALAR MULTIPLY 09008802041 VECTOR SUBTRACT 09008902031 VECTOR ADD Q9009002061 CROSS PRODUCT 09009102271 PRINCIPAL VALUE 09009202196 SQUARE ROOT Q9009302101 SINE 09009402105 COSINE Q9009503852 MU 09009600201 S (T) 09009700202 C (T) 09009800380 PARTIAL R BAR / x J 09009900383 PARTIAL R DOT BAR / X J V00092+740000000+02 V00050+000000000+00 V00051+100000000+01 V00052+200000000+01 V00053+30000000+01

V00054+500000000+00

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		40.04*	40.16*	67.05*					
34.08	D1479	34.05							
34.10	01501								
35.01	01502								
36,91	01503								
37.17	01504								
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37.29	01506								
38.01	01507								
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40.27	01510								
41.01	91511								
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49.01	01540								
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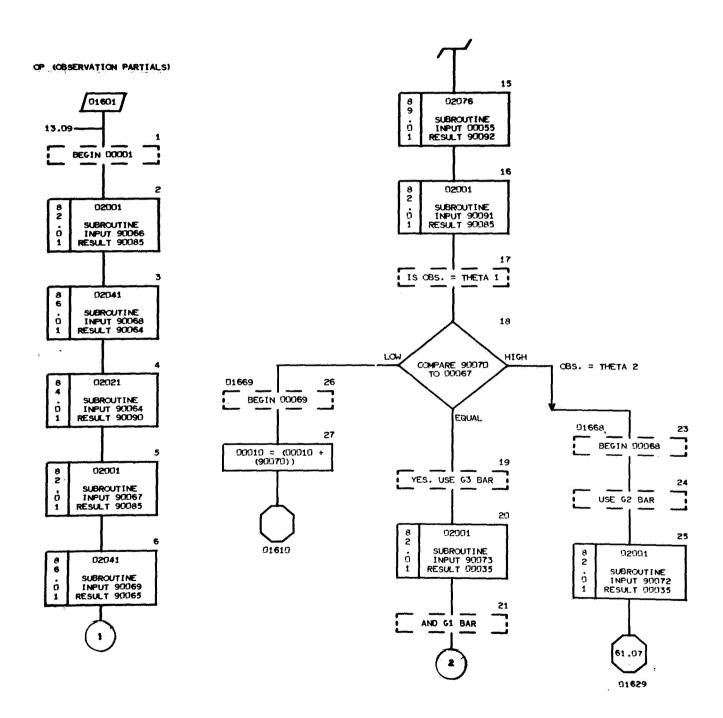
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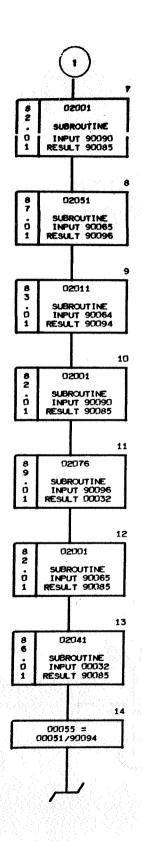
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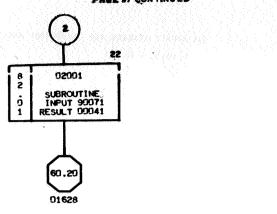
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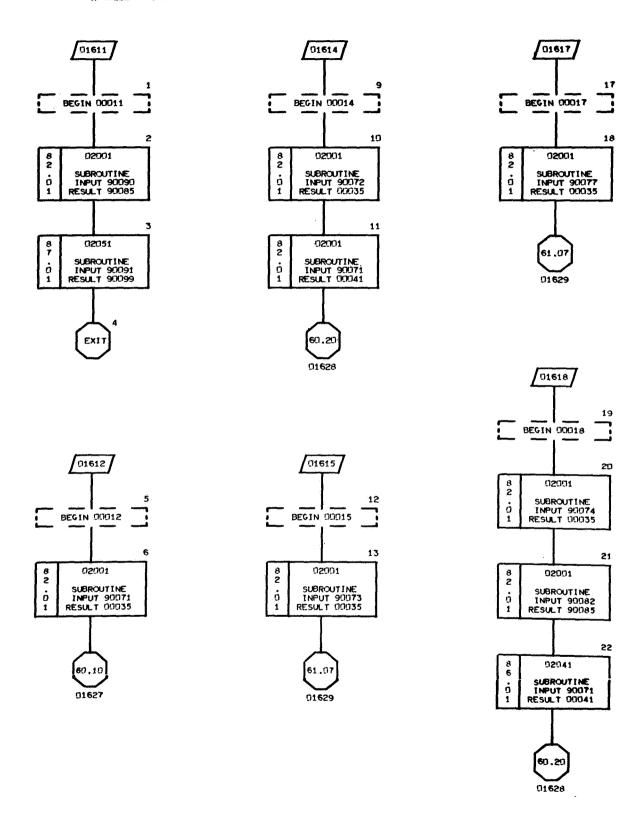
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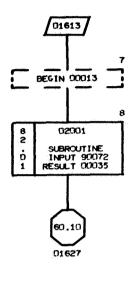
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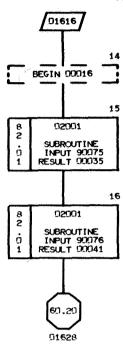




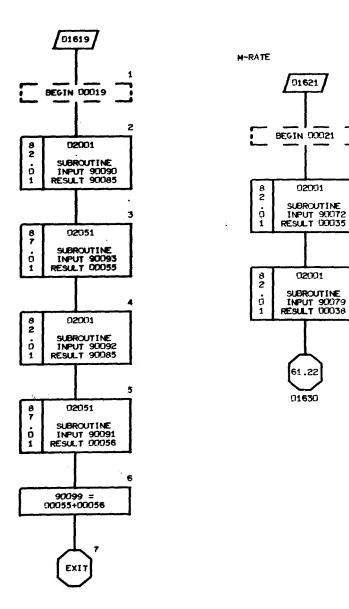


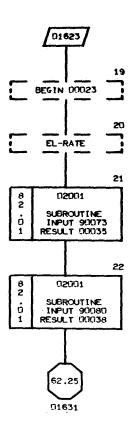
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NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI K VALUE = 01600



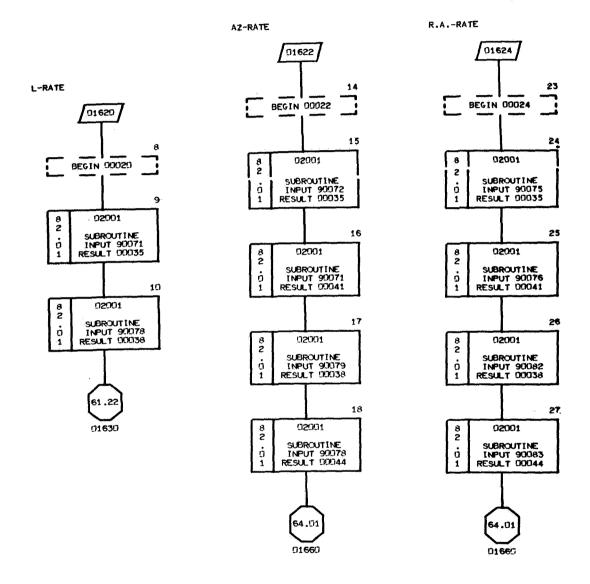


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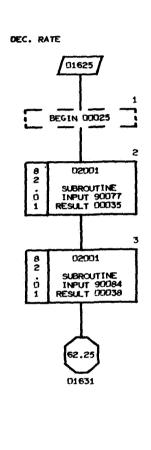
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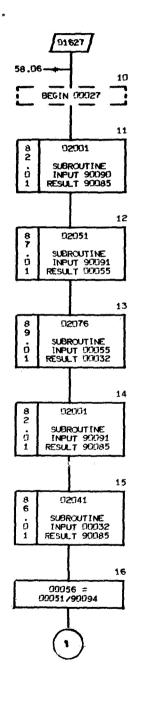
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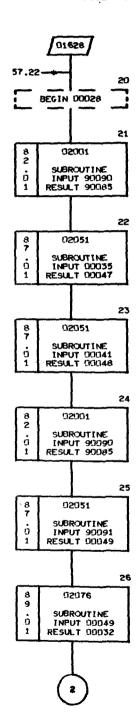
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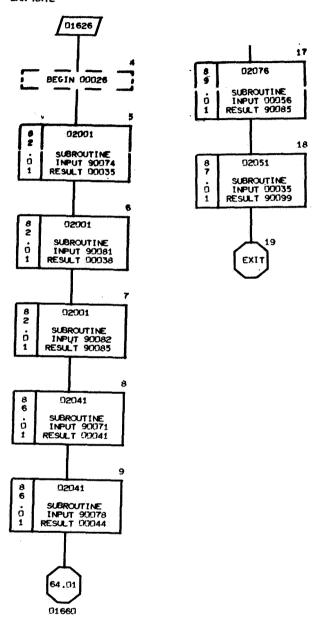
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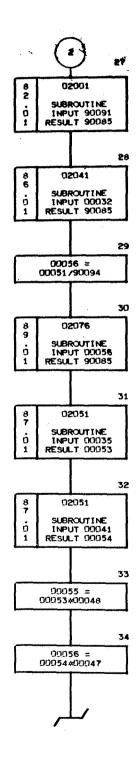


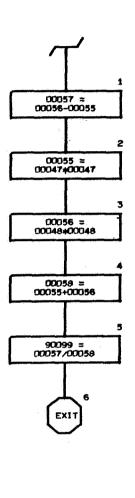


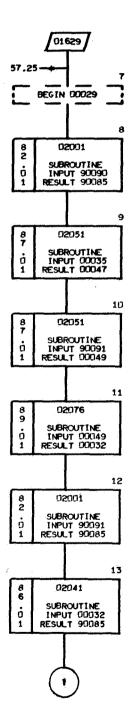


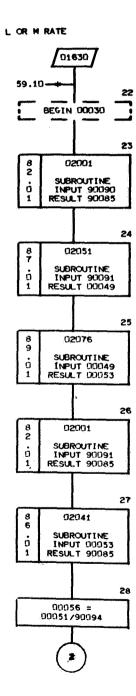


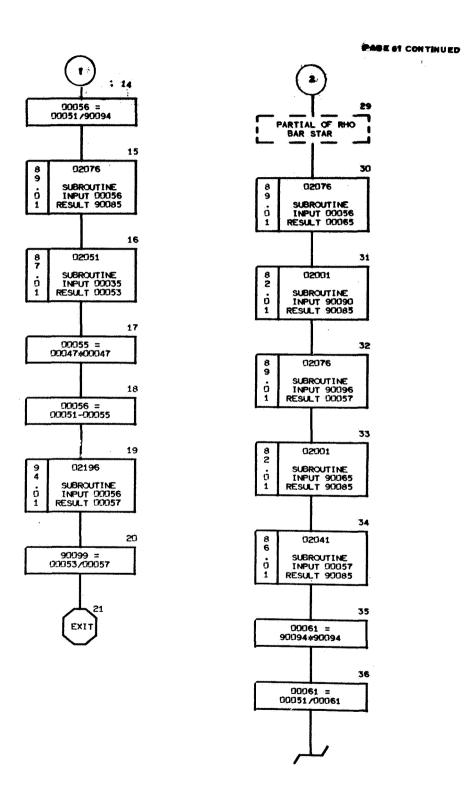






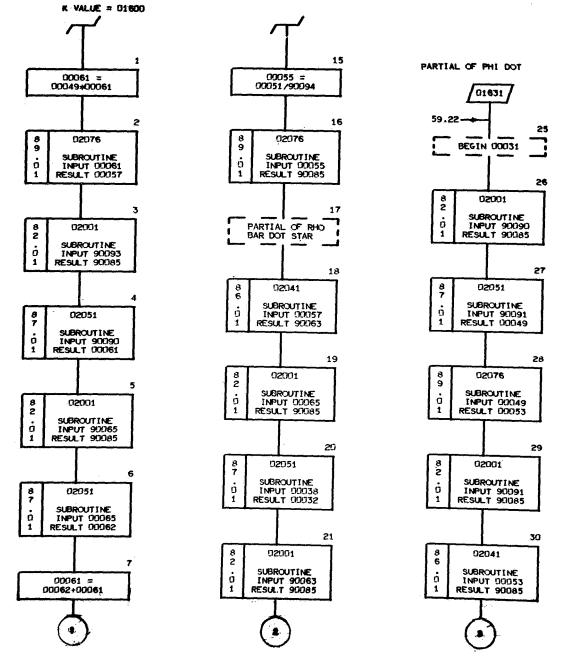


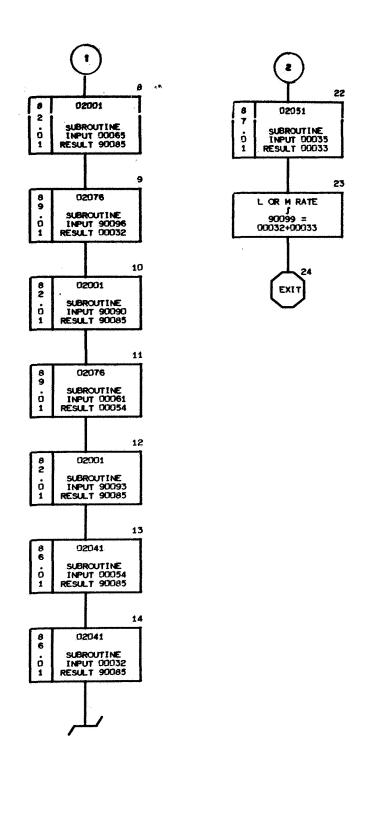


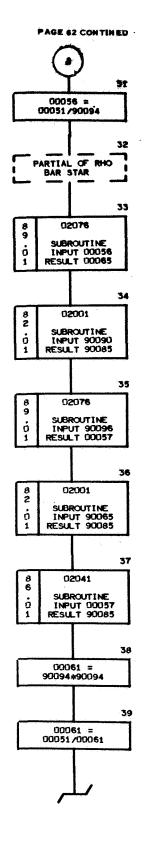


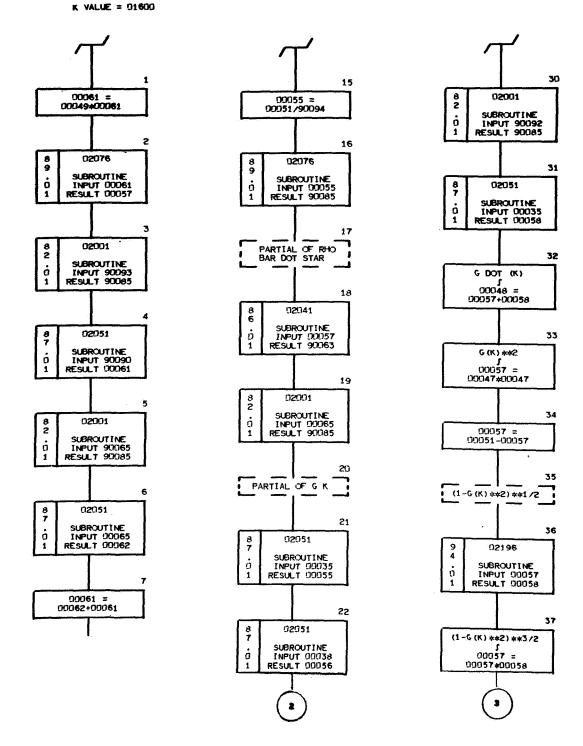


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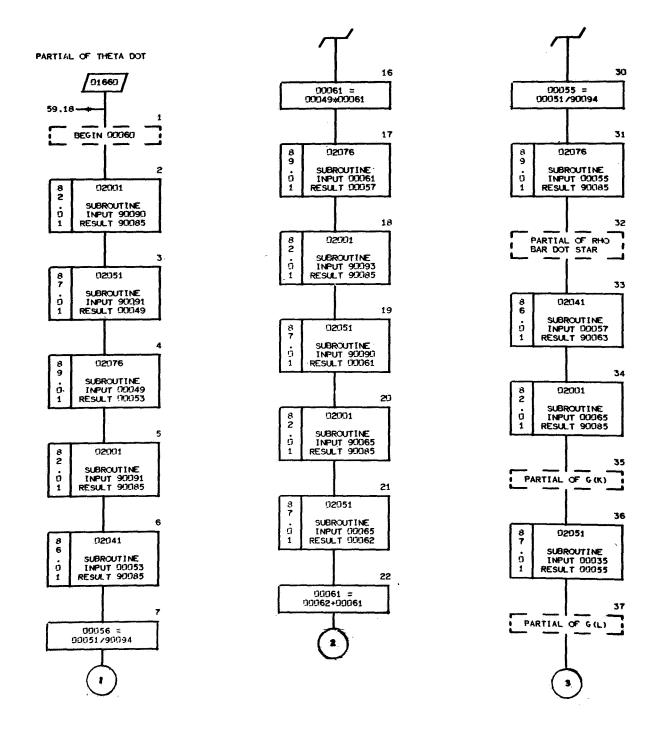




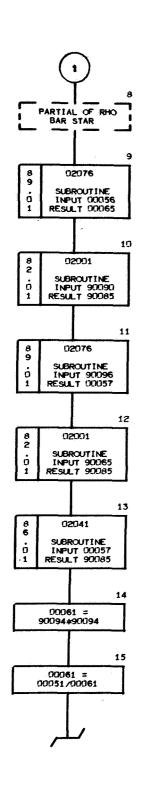


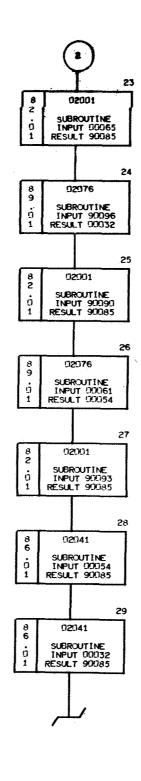
#### PAGE 63 CONTINUED 02001 SUBROUTINE INPUT 00065 RESULT 90085 38 23 00059 = 00047+00048 8 02001 SUBROUTINE INPUT 90063 RESULT 90085 ó 1 30 8 02076 00059 = 00059+00055 SUBROUTINE INPUT 90096 RESULT 00032 24 0 8 02051 SUBROUTINE INPUT 00035 RESULT 00057 40 . 1 10 00059 = 00059/00057 02001 8 2 SUBROUTINE INPUT 90090 RESULT 90085 25 . 1 41 PARTIAL OF G DOT 00058 = 00056/00058 90056 = 90057+99056 11 8 02076 42 SUBROUTINE INPUT 00061 RESULT 00054 PARTIAL OF PHI 26 ó 8 02001 90099 = 00058+00059 SUBROUTINE INPUT 90090 RESULT 90085 . 1 12 8 2 02001 SUBROUTINE INPUT 90093 RESULT 90085 . 1 EXIT G (K) 13 28 02041 8 8 02051 SUBROUTINE INPUT 00054 RESULT 90085 SUBROUTINE INPUT 90935 RESULT 90947 ó . 1 14 29 8 02041 8 7 92951 SUBROUTINE INPUT 00032 RESULT 90085 SUBROUTINE INPUT 00038 RESULT 00057 . 1 ö

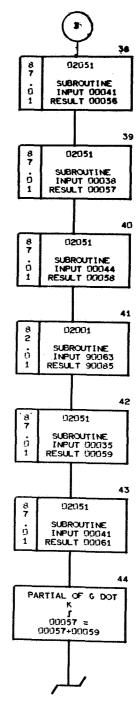
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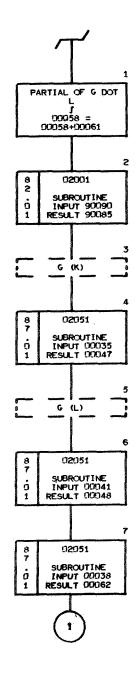
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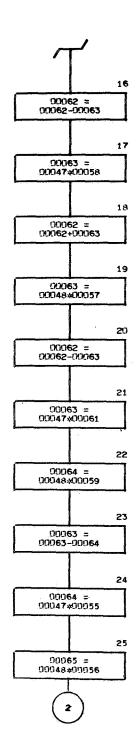




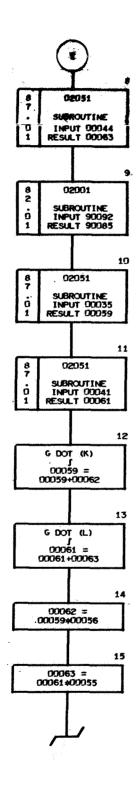


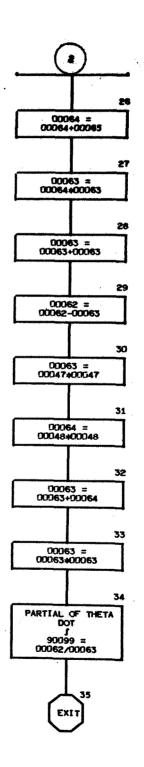
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#### K VALUE = 01600

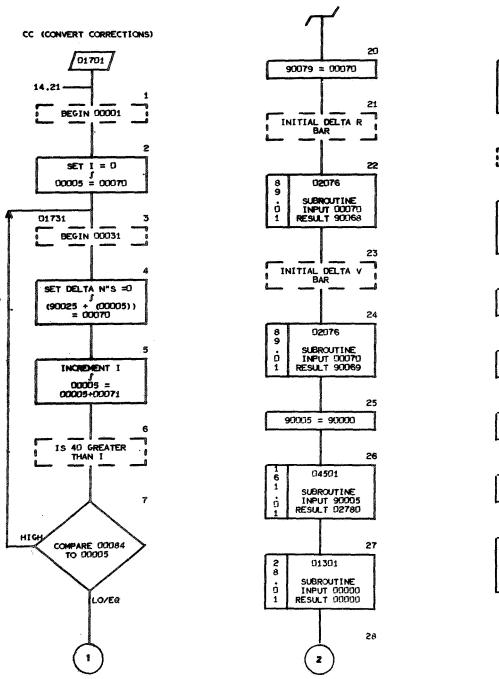
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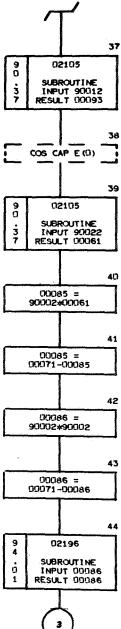
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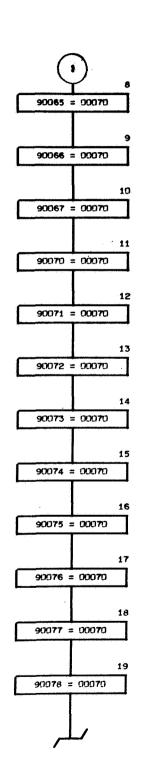
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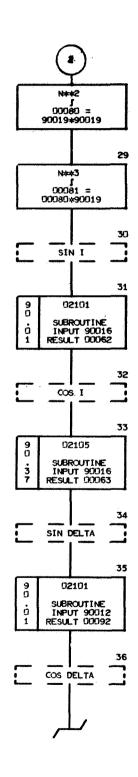
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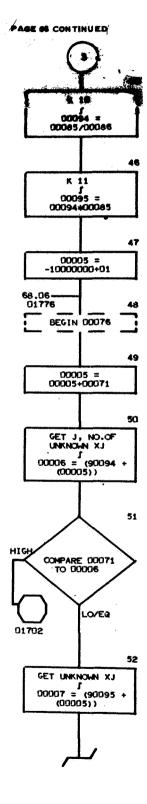
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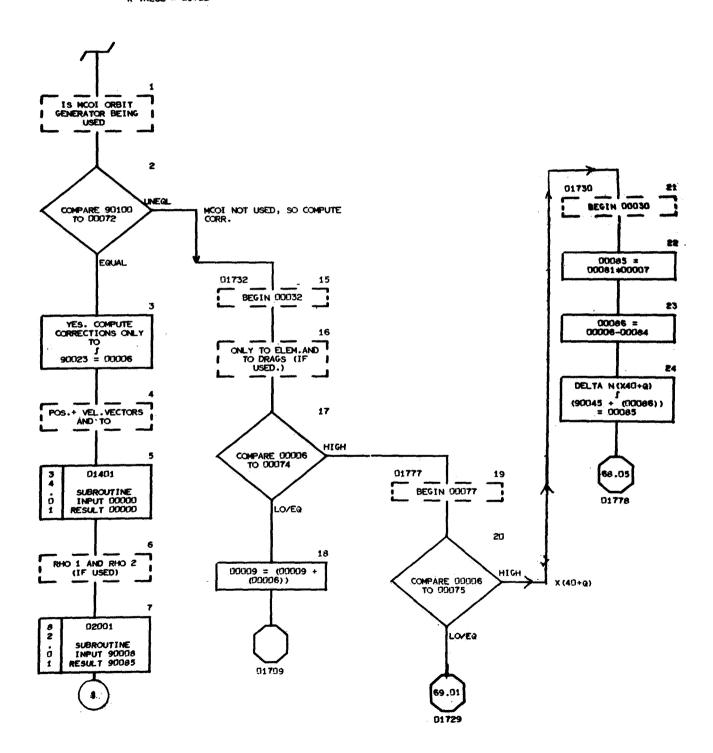




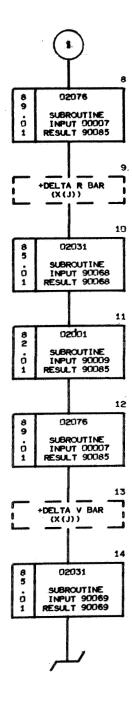


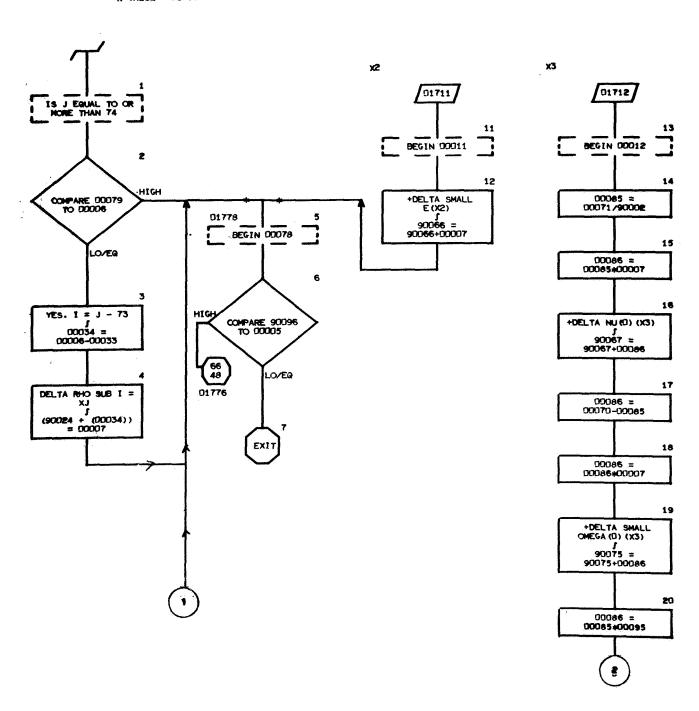


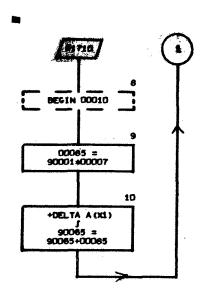


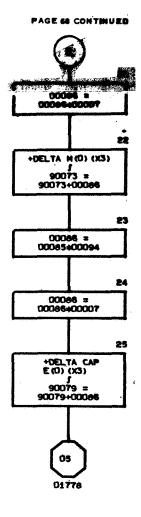


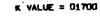
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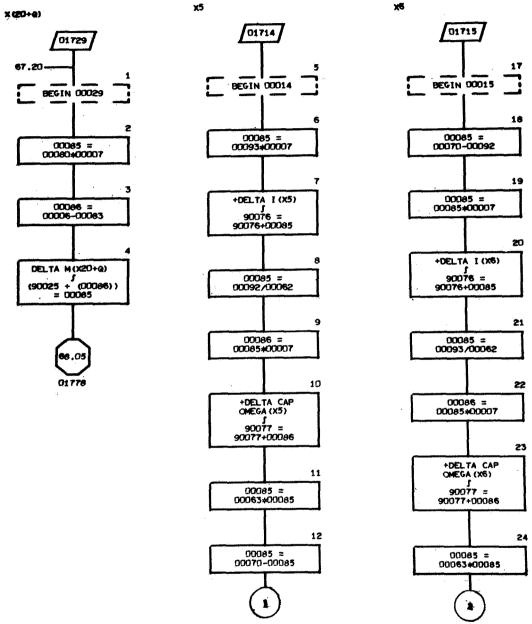




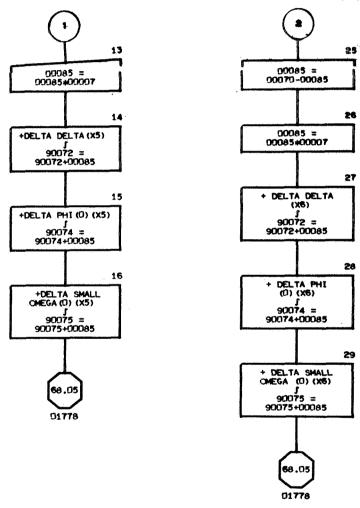




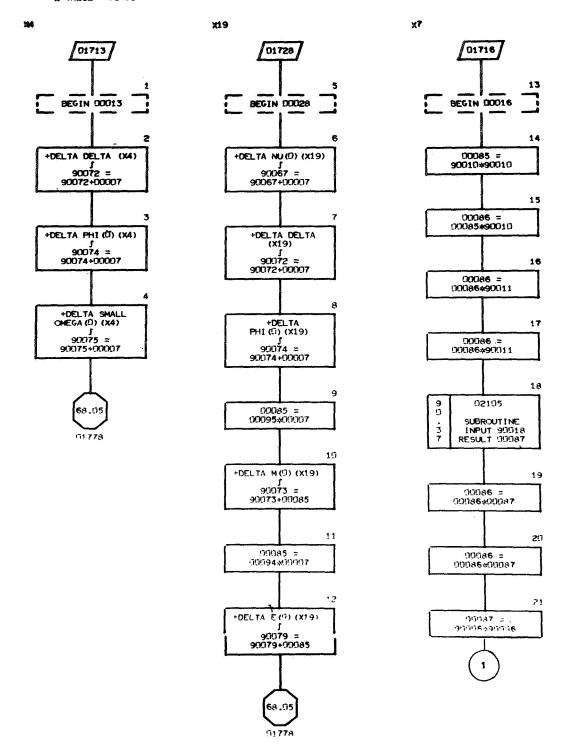




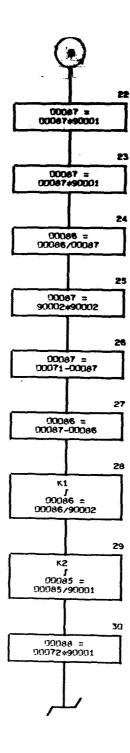
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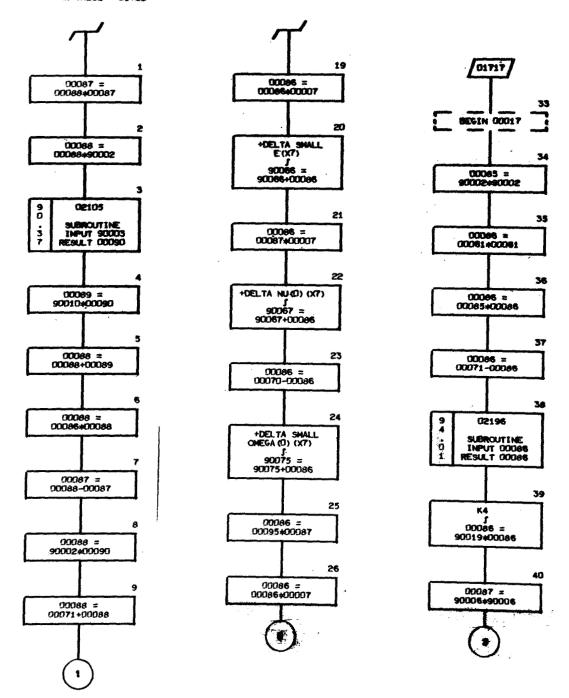
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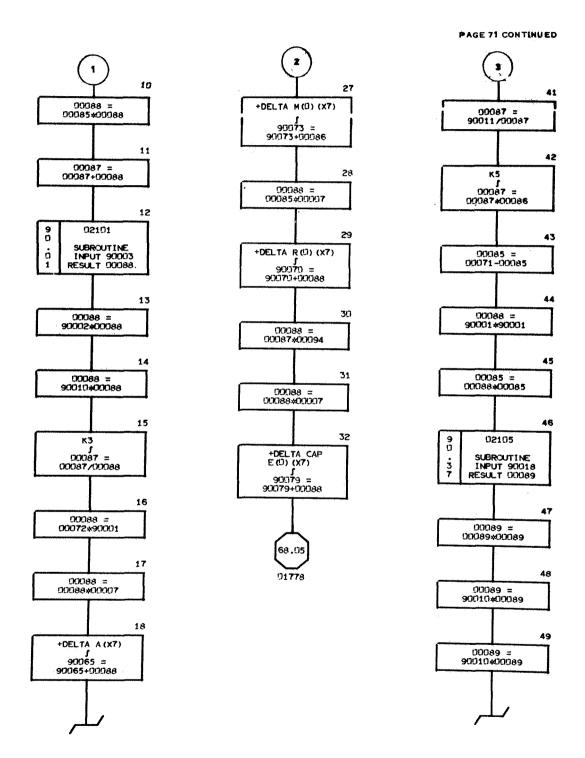


# PAGE 76 CONTINUED



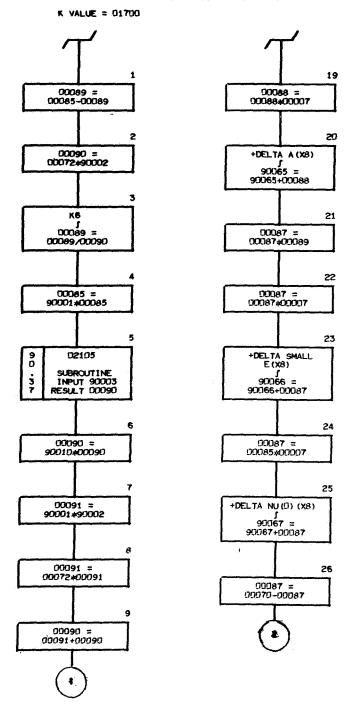
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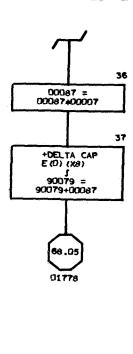




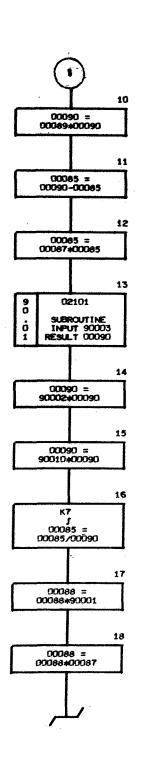
# MASA-BSFC MISSION AND TRAJECTORY ANALYSIS DIVI

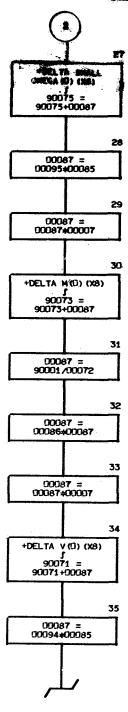
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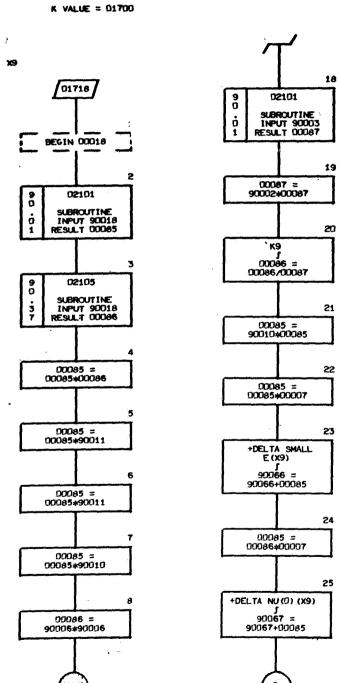


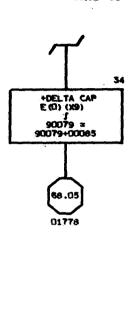
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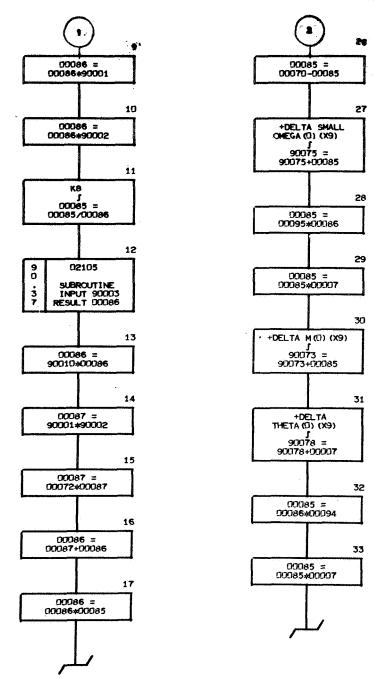


MASA-GSFC HISSION AND TRAJECTORY ANALYSIS DIVI





# PAGE 73 CONTINUED



# K VALUE = 01700

 Q9000001100
 T(0)

 Q9000101101
 A

 Q9000201102
 SNALL E

 Q9000301103
 NU(0)

 Q9000401104
 R BAR

 Q9000500200
 T(N)

 Q9000603852
 MU

 Q9000701107
 V BAR

Q9000000380 PARTIAL R BAR (X(J))
Q9000900383 PARTIAL V BAR (X(J))

 Q9001001110
 R (0)

 Q9001101111
 V (0)

 Q9001201112
 DELTA

 Q9001301113
 M (0)

 Q9001401114
 PHI (0)

Q9001501115 SMALL OMEGA (0)

Q9001601116 I

Q9001701117 CAP OMEGA Q9001801118 THETA (D) Q9001901119 N

**Q9002201120** CAP E (0) **Q9002300252** J,UNKNOWN NO.

99002400524 LOC.PRECEDING DELTA N(2,Q) OR DELTA
99002500525 DELTA N(2,Q) OR DELTA RHO SUB I

 Q9004500545
 DELTA N(3,Q)

 Q9006500500
 DELTA A

 Q9006600501
 DELTA SMALL E

 Q9006700502
 DELTA NU(0)

#9006900503 DELTA R BAR #9006900506 DELTA V BAR #9007000509 DELTA R(D) #9007100510 DELTA V(D) #9007200511 DELTA DELTA #9007300512 DELTA H(D) #9007400513 DELTA PHI (O)

99007500514 DELTA SMALL CHEGA (D)

99007600515 DELTA I

 Q9007700516
 DELTA CAP OMEGA

 Q9007800517
 DELTA THETA (D)

 Q9007900519
 DELTA CAP E (G)

 99008502085
 VQ

 99008602001
 VMV

 99008702076
 SM

99008802031 VECTOR ADD

 09000901301
 PE

 09009001401
 PP

 09009102101
 SIN

 09009202105
 COS

 09009302196
 SQ RT

99009400400 J (UNKNOWNS)

99009500441 X(J)

 Q9009600013
 LIMIT OF UNKNOWNS

 Q9009700002
 M(LIMIT OF Q"S)

 Q9009804501
 ORBIT GENERATOR

 Q9010000099
 ORBIT GENERATOR IDENT.

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V00075+390000000+02 V00079+740000000+02

V00074+190000000+02

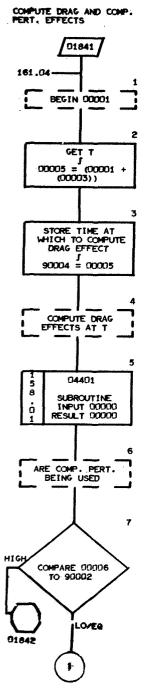
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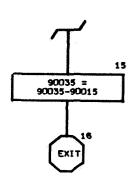
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# V00063+20000000+02 V00064+40000000+02

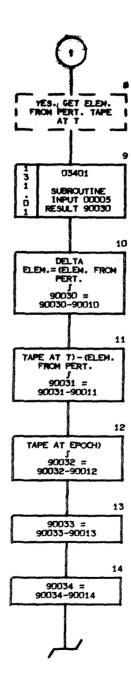
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60.05	01778	67.24	50.88	66.04	66.10	92.00	66.25	9,34
	ide and	69.16	69.29	70.04	70.12	71.32	72.37	76.34
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71.33	01717							
73,01	01718							





# PAGE 74 CONTINUED



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#8000104401 DRAG (DELTA N) F.
#80000200085 ONP. PERT. OPTION

99000303401 COMP. PERT. TAPE READ F.

49000400200 TINE OF DRAG EFFECTS

99001000270 A ELEM. FROM PERT. TAPE

99001100271 E AT EPOCH

99001200272 I

Q9001300273 M

G9001400274 SMALL OMEGA
G9001500275 CAP OMEGA
G9003000276 DELTA A
G9003100277 DELTA E
G9003200278 DELTA 1

Q9003300279 DELTA N

99003400280 DELTA SMALL OMEGA
99003500281 DELTA CAP OMEGA

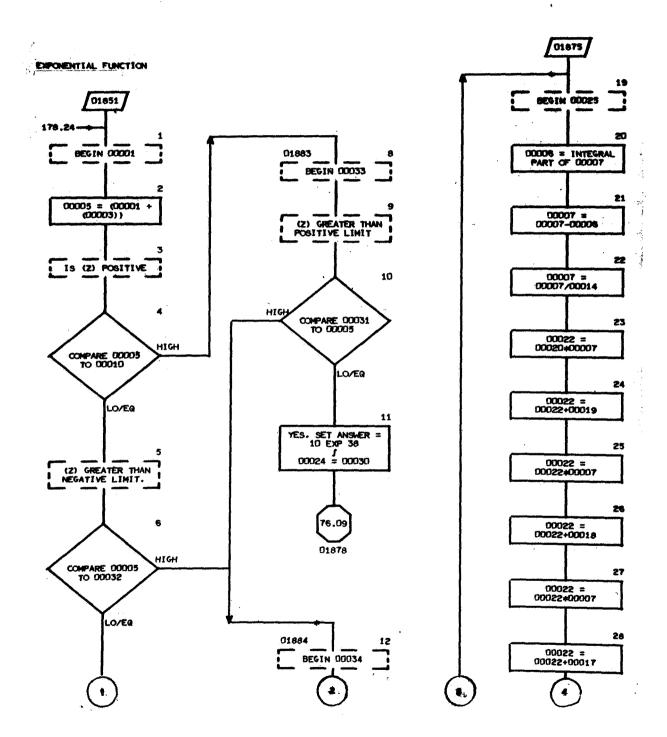
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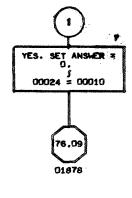
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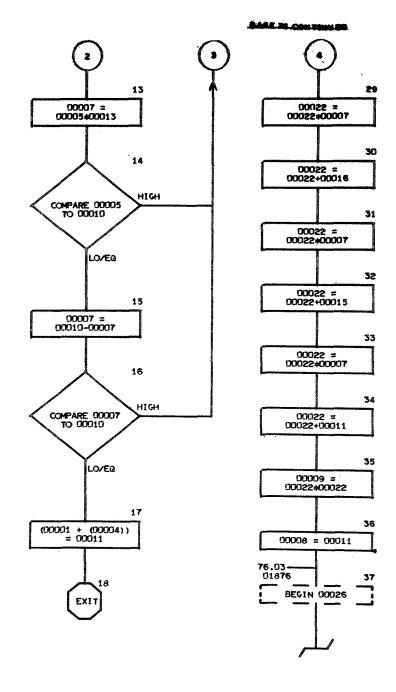
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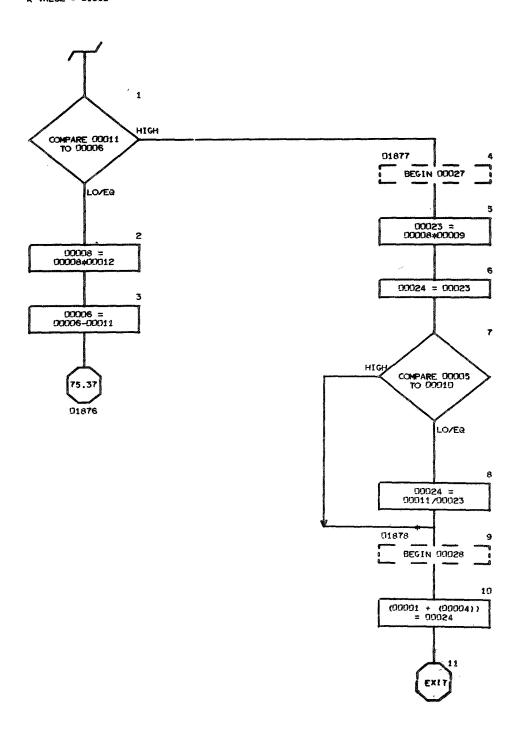
(4.01 01841 161.0cc







K VALUE = 01850



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V00011+100000000+01

V00012+10000000+02

V00013+43429448+00

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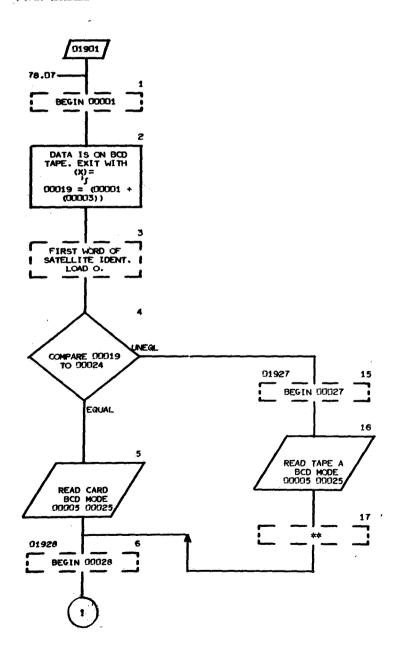
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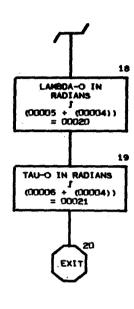
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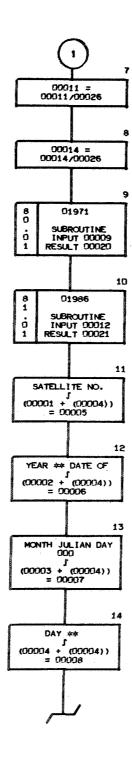
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# SATELLITE IDENTIFICATION LOAD FUNCTION





PAGE 77 CONTINUED



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 77.01
 01901
 78.07\*

 77.06
 01928
 77.17

 77.15
 01927
 77.04

K VALUE = 01900

 Q0002201971
 H-M-S TO RAD) FUNCTION

 Q0002301986
 D-M-S TO RAD, FUNCTION

V00024+000000000+00 (USES LOCATIONS 1 TO 34)

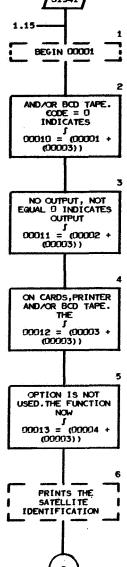
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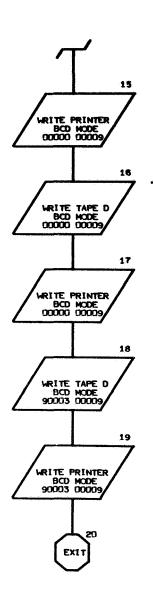
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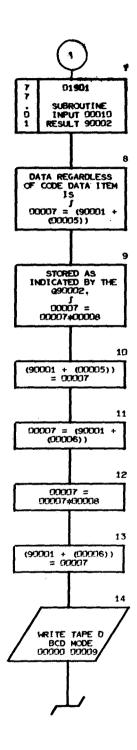
SATELLITE IDENTIFICATION
LOAD AND PRINT
SATELLITE IDENT L. AND P.
FUNCTION (USES

01941

1.15







K VALUE = 01940

Q9000101901 LOCAT OF SAT. IDENT. PACKAGE

Q9000200294 LOCAT OF SAT DATA ITEM
Q9000301905 LOCAT OF SAT. IDENT.LD +4

V00005+10000000+02 LOCATIONS 1 TO 24)

 V00006+13000000+02
 ENTER WITH (2)=0 IF INPUT DATA IS

 V00008+10000000+04
 CARDS OR NOT EQUAL 0 IF INPUT IS

 V00009+100000000+01
 TAPE, (Z+1),(Z+2),AND (Z+3) = TO

 V00014+00000000+00
 OUTPUT OPTION CODES FOR CARDS,PRI

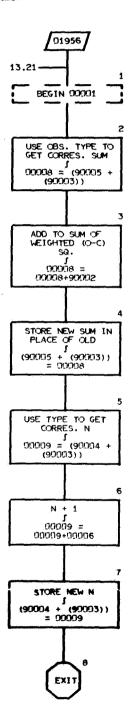
CROSS REFERENCE LISTING

PAGE BOX LABEL REFERENCES

K VALUE = 01940

78.01 01941 1.15\*

ADD TO CORRES. SUM OF (O-C) SQ. AND TO N FOR ONE OBS.



### PAGE # CONTINUED

R VALUE ... D1965

99000100258 WEIGHT FOR OBS.

Q9000200047 WEIGHTED (O-C) SQUARED

99000300253 OBS. TYPE

Q9000401899 LOC.PRECEDING 20 LOCS. OF N°S

Q9000501919 LOC.PRECEDING 20 LOGICE SUNS OF 60-C
V00005+000000000+00 CBS.TYPE.ALSO ADDITION N, THE NO.OF
V00006+10000000+01 INCLUDED IN SUM (\$\frac{1}{2}\$ TYPE. USES

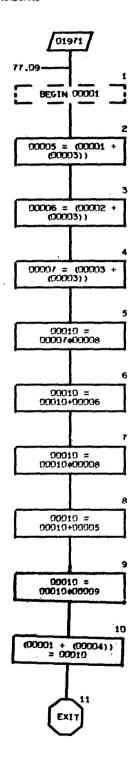
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PAGE BOX LABEL REFERENCES

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79,01 01956 13.284

# HOURS-MINUTES-SECONDS TO RADIANS



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@0000803875

1/60 RADIANS/HOUR 00000903874

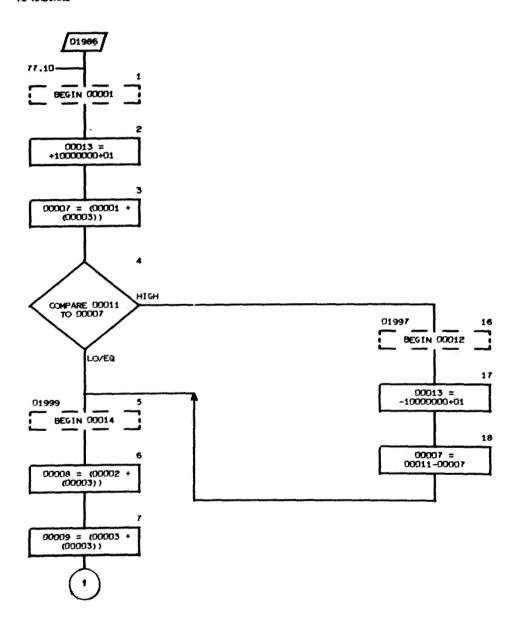
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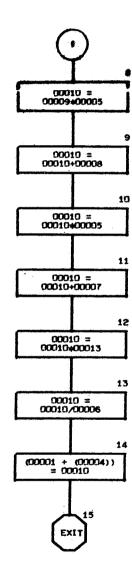
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88.01 D1971 77.09+

#### DEGREES-MINUTES-SECONDS TO RADIANS



# PAGE 81 CONTINUED



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90000603847 DEGREES/RADIAN

V00011+00000000+00

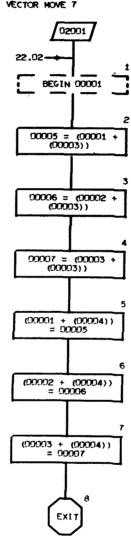
CROSS REFERENCE LISTING

PAGE BOX LABEL REFERENCES

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81,01 01986 /7.10q 81,05 01999 81.18 81,16 01997 81.04

## VECTOR PACKAGE VECTOR MOVE 7



#### CROSS REFERENCE LISTING

PAGE BOX LABEL

REFERENCES

K VALUE = 02000 22.12+ 22.19+ 22.89e 22.07\* 22.104 82.91 02001 #50.55 22.05+ 23.17\* 23.24# 23.25e 23.274 22.22+ 22.25\* 23.11\* 24.03+ 24.04\* 24.064 24.06\* 24.09+ 24.20+ 24.22+ 25.11\* 25.12+ 25.14\* **25.02**\* 25.04+ 25.09+ 25.10\* 25.164 25.18\* 25.19# 25.20\* 25.21+ 25.234 25.24# 26.04# 26.02\* 26.03\* 26.084 26,14+ 26.194 26.214 26.24\* 26.264 26.29# 27.09\* 27.11\* 29.17# 28.24\* 28.284 28.33\* 28.39# 28.44\* 29,02+ 29.21+ 29.25+ 29.31\* 29.35\* 34.14\* 34.16+ 34.23\* 34,25+ 35.07\* 35.16# 35.18\* 35.20\* 35.25\* 36.17\* 36.23+ 36.25\* 36.42\* 37.06\* 37.18\* 37.20+ 37.24# 37.264 37.30+ 37.32\* 38.05\* 38.07\* 38.15\* 38.18\* 38.294 36.32\* 39.08\* 39.10\* 39.18\* 39.21\* 39.32\* 39.35\* 40.11\* 40.20\* 40.23\* 41.07\* 40.13\* 40.31\* 41.12+ 41.14\* 42.07\* 42.05\* 42.13\* 42.24\* 42.28\* 42.39+ 43.02\* 43.43\* 43.17\* 43.28\* 44.02\* 43.13\* 43.32\* 44.13\* 45.03# 44.18\* 44.28\* 44.33\* 44.43\* 45.13\* 45.18\* 45.28\* 45.33\* 45.43\* 46.03\* 46.13\* 46.18\* 46.28\* 46.33\* 46,43\* 47.03\* 47.13\* 47.18\* 47.28\* 47.33\* 48.03\* 47.43\* 48.13\* 48.18\* 48.28\* 48.33\* 48.43\* 49.02\* 49.15\* 49.19\* 49.32\* 49.36\* 49.49\* 50.02\* 50.15\* 50.194 50.32\* 50.36\* 50.49\* 51.03\* 51.15\* 51.20\* 51.32\* 51.37\* 51.49\* 52.03\* 52.15\* 52.20\* 52.32\* 52.37\* 52.49\* 53.03\* 53,15\* 53.20\* 53.32\* 53.37\* 53.49\* 54.03\* 54.15\* 54.20\* 54.32\* 54.37\* 55.03\* 54.49\* 55.15\* 55.20\* 55.32\* 55.37\* 55.39\* 56.03\* 56.05\* 57.02\* 57.05\* 57.07\* 57.10\* 57.12\* 57.20\* 57.16\* 57.22\* 57.25\* 58.02\* 58.06\* 58.06\* 58.10\* 58.11\* 58.13\* 58.15\* 58.16\* 58.18\* 58.20\* \$15.80 59.02\* 59.04\* 59.09\* 59.10\* 59.12\* 59.13\* 59.16\* 59.15\* 59.17\* 59.18\* 59.21\* 59.22\* 59.24\* 59.25\* 59.26\* 59.27\* 60.02\* 60.03\* 60.05\* 60.06\* 60.07\* 60.11\* 60.14\* 60.21\* 60.24\* 60.27\* 61.06\*

61.26#

61.31\*

61.33\*

62.03\*

62.05+

61.12\*

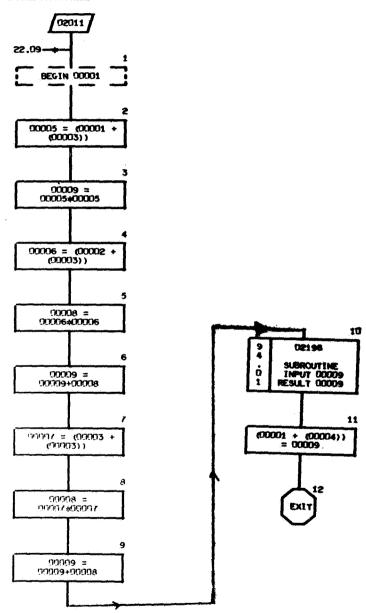
61.23\*

# 

124.09+ 124.20+ 124.23+ 124.31+ 124.33+ 180.15+ 181.38+

K VALUE = 02010

# VECTOR MAGNITUDE 9



# CROSS REFERENCE LISTING

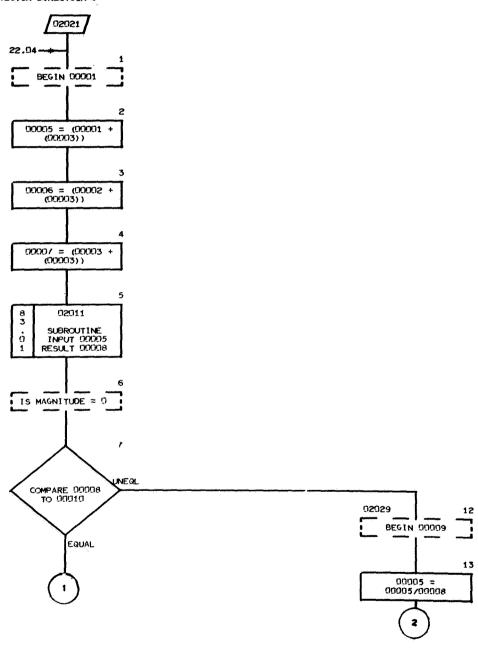
PAGE BOX LABEL REFERENCES

85.01 02011 22.09\* 25.02\* 29.41\* 29.43\* 57.09\* 84.05\* 118.10\*

K VALUE = 02010

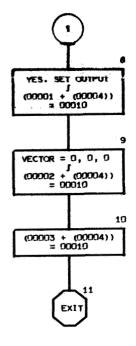
90002502196 SQUARE ROOT

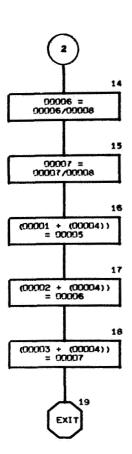
# VECTOR DIRECTION 8



#### 3 1 10

PAGE 84 CONTINUED





### CROSS REFERENCE LISTING

PAGE BOX LABEL REFERENCES

K VALUE = 02020

84.01 02021 22.04\* 23.04\* 24.15\* 57.04\* 119.21\* 119.23\* 119.30\*

119.34\* 124.38\* 124.39\*

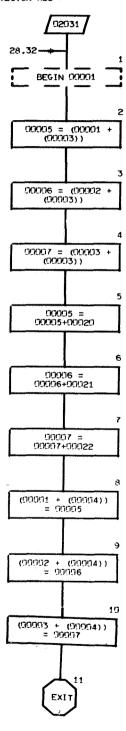
84.12 02029 84.07

K VALUE = 02020

QD001102011

V00010+000000000+00





# NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI

PAGE 8 A

K VALUE = 02030

00002002085

**40002102088** 

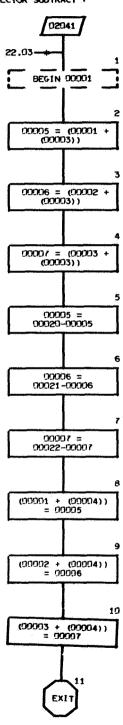
**40**0002202087

CROSS REFERENCE LISTING

39.34\* 39.37\* 40.22\* 40.25\* 41.16\* 42.09\* 67.10\*

67.14\* 123.23\* 124.06\* 124.26\*

#### VECTOR SUBTRACT 7



## PAGE 86 CONTINUED

### CROSS REFERENCE LISTING

PAGE BOX LABEL REFERENCES

K VALUE = 02040

86,01 02041 22.03\* 22.06\* 22.13\* 24.11\* 26.05\* 26.06\* 29.03\* 29.37\* 34.17\* 34.27\* 35.19\* 57.03\* 57.06\* 57.13\*

58.22\* 60.08\* 60.09\* 60.15\* 60.26\* 61.13\* 61.27\* 61.34\* 62.13\* 62.14\* 62.18\* 62.30\* 62.37\* 63.13\* 63.14\* 63.18\* 64.06\* 64.13\* 64.28\* 64.29\* 64.33\*

124.14 124.35\*

K VALUE = 02040

20002002085

90002102086

00002202087

# DOT PRODUCT 9 02051 22.08-BEGIN 00001 2 00005 = (00001 + (00003)) 00009 = 00005<del>4</del>00020 00006 = (00002 + (00003)) 90008 = 90006#09921 6 00009 = 00009+00008 00007 = (00003 + (00003)) 90008 = 920004700022 00009+00008 10 (99991 + (99994)) = 99999

### PAGE 87 CONTINUED

PAGE BOX	LABEL	REF	ERENCES					
· J.K. VALUE	= 08090	*					:	
87.01	02051	22.08*	23.12*	23.18*	24.21*	24.23*	25.034	25.054
		26.09#	26.10*	26.15*	26.20*	26.22*	26.25*	26.27#
		26.30*	26.31*	27.10*	27.12*	27.14*	57.08*	58.03+
		59.03*	59.05*	60.12*	60.18*	*55.00	60.23*	90.25*
		60.31*	60.32*	61.09*	61.10*	61.16*	61 ,24+	62.04+
		62.06*	62.20 <b>*</b>	62.22*	62.27#	63.04*	63.06+	63.21*
		63.22*	63,24*	63.28*	63.29*	63.31*	64,03*	64,194
		64.21*	64.36#	64.38*	64.39*	64.40*	64.42+	64,484
		65.04*	65.06+	65.07*	65.08*	65.10*	65.11*	119.24+
		119.36+	119.40*	119.43#	120.01*	120.04*	120.18+	124.494
		180.16+	181.33+					

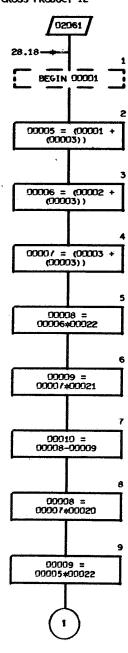
K VALUE = 02050

@0002002085

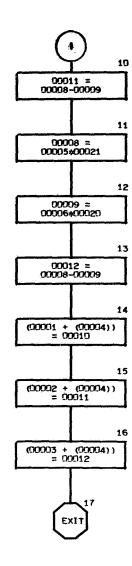
00002102086

00002202087

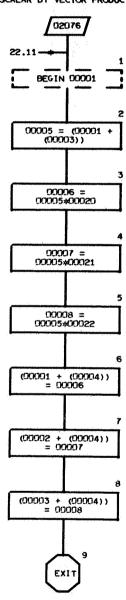
# CROSS PRODUCT 12



### PAGE 88 CONTINUED



### SCALAR BY VECTOR PRODUCT



# PAGE 88 CONTINUED

PAGE BOX LABEL REFERENCES

K VALUE = 02060

88.01 02061 28.18\* 28.25\* 28.35\* 35.21\* 37.19\* 37.21\* 37.25\*

37,27\* 37,31\* 37,33\* 119,28\* 119,32\* 119,42\* 120,03\*

123.12+ 123.18+ 123.26+

K VALUE = 02060

00002002085

90002102086

90002202087

NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI CROSS-REFERENCE LISTING

PAGE BOX LABEL REFERENCES

*	VALUE = 02075							
69.01	02076	22.11*	22.15*	28.20*	28,27#	28.30*	28.41*	28.43#
		28.46*	28.48*	29.22*	29.26*	29.32*	29.36*	29.39+
		34.15*	34.24*	34.26#	*80.25	35.17*	35.22+	35.26+
		35.33*	36.18*	36.24*	36.29*	36.43*	37,094	37.15*
		38.16*	38.19*	38.30*	38,33*	39.09*	39.11*	39.19#
		39.22*	39.33*	39.36*	40.12*	40.14*	40.21*	40.24*
		40.32*	40.33*	41.08*	41,09*	41,13*	41.15*	41.29*
		42.06*	42.08*	42.10*	42.14*	42.15*	42,25*	42.29*
		42.30*	42.40*	43.03*	43,04*	43.14*	43,18*	43.19#
		43.29*	43.33*	43.34*	43.44*	44.03*	44.04*	44.14*
		44.19*	44.20*	44.29*	44.34*	44.35*	44.44*	45.04*
		45.05*	45.14*	45.19*	45.20*	45.29*	45.34*	45.35*
		45.44*	46.04*	46.05*	46.14*	46.19*	46.20*	46.29*
		46.34*	46.35*	46.44*	47,04*	47.05*	47.14*	47.19*
		47.20*	47.29#	47.34*	47,35*	47.44*	48.04*	48.05*
		48.14*	48.19*	48.20*	48.29*	48.34*	48.35*	48.44*
		49.03*	49.04*	49.16*	49.20*	49.21*	49.33*	49.37*
		49.38*	49.50*	50.03*	50.04*	50.16*	50.20*	50.21*
		50.33*	50.37*	50.38*	50,50*	51.04*	51.05*	51.16*
		51,21*	51.22*	51.33*	51.38*	51.39*	51.50*	52.04*
		52.05*	52.16*	52.21*	<b>52.22</b> *	\$2.33*	52.38*	52.39*
		52.50*	53.04*	53.05*	53.16*	53.21*	53.22*	53.33*
		53.38*	53.39*	53.50*	54.04*	54.05*	54.16*	54.21*
		54.22*	54.33*	54.38*	54.39*	54.50*	55.04*	55.05*
		55.16*	55.21*	55.22*	55.33*	57.11*	57.15*	60.13*
		60.17*	60.26#	60.30+	61,11*	61.15*	61.25*	61.30*
		61.32*	62.02*	<b>62.09</b> *	62.11*	62.16*	62.28*	62.33*
		62.35*	63.02*	63.09*	63,11*	63.16*	64.04*	64.09*
		64.11*	64.17*	64.24*	64.26*	64.31*	66.22*	66.24+
		67.08*	67.12*	123.13*	123.19*	123.21*	124.03*	124.05*
		124.08*	124.10*	124.21*	124.24*	124.32*	124.34*	124.37#

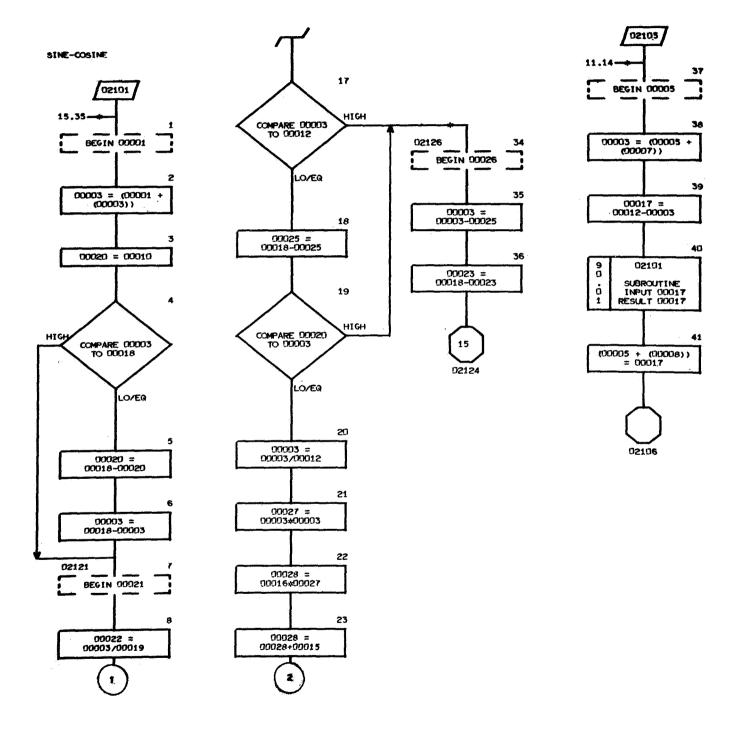
K VALUE = 02075

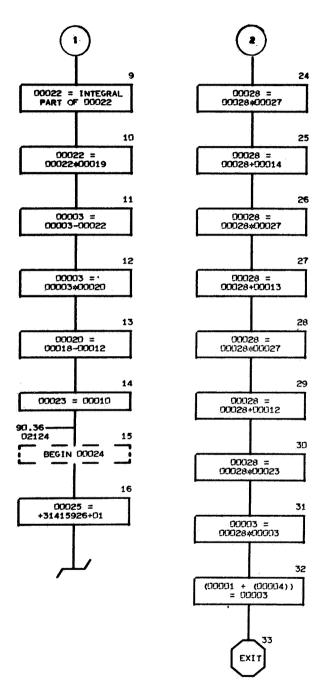
90002002085 00002102086 90002202087

# NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI

PAGE 90

#### K VALUE = 02100





PAGE BOX LABEL REFERENCES

K VAL	UE = 02100							
90.01	02101	15.35*	15.37*	28.15*	28,19*	28.26*	28.38*	29.13#
		35.12*	36.03*	36.14*	38,26*	39.29*	40,07*	66.31*
		66.35*	71.12*	72.13*	73.02*	73.18#	90.40*	95.15*
		105.08*	105.10*	121.18*	122.25*	123.01*	123.06*	123.10*
		123.37*	124.01*	124.15*	125.06*	145.13*	163.29*	169.03*
		169.32*	169.36*	169.46*	170.12*	170.14*	170,47*	171.36*
		172.03*	172.07*	172.11*	180.04*	181.22*		
90.07	02121	90.04						
90.15	02124	90,36						
90.34	02126	90,17	90.19					
90.37	02105	11.14*	28.13*	28.22*	28.29*	28.37*	<b>29.</b> 09∗	35.02*
		36.02*	36.19*	38.13*	39.05*	39.16*	40.17*	41.17*
		41.22*	41.30*	41.38*	66.33*	66.37#	66.39*	70.18*
		71.03*	71.46*	72.05*	73.03#	73.12*	95.14*	105.09*
		105.11*	105.17*	122.23*	122.49*	123.04*	123.08*	123.30*
		123.43*	124.13*	125.04*	145.11*	162.09*	166.47*	169.05*
		169.07*	169.30*	169.48*	170.16*	170.20*	170.22*	171.38*
		172.05*	172.09*	172.13*	180.10*	181.03*		

K VALUE = 02100

V00010+100000000+01

V00012 15707963 01

V00013-64596371 00

V00014 79689679-01

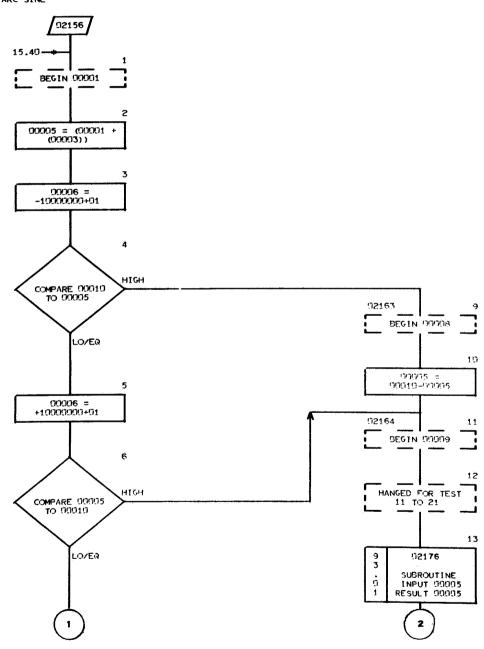
V00015-46737660-02

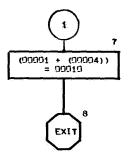
V00016 15148400-03

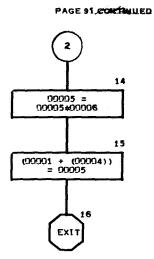
V00018+000000000+00

V00019+62831853+01

# ARC SINE

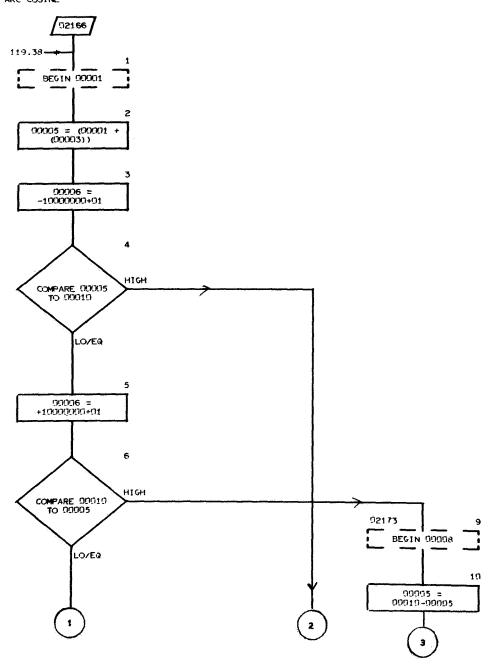


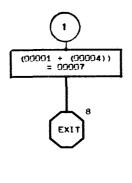


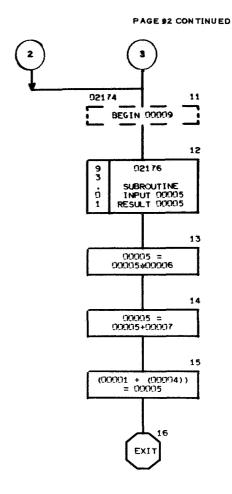


PAGE BOX	LABEL	REF	ERENCES			
K VAL	UE = 02155					
91.01	02156	15.40*	26.16#	119.25*	124.41*	174.10#
91.09	02163	91.04				
91.11	02164	91,06				

# ARC COSINE







PAGE BOX LABEL REFERENCES

K VALUE = 02165

92.01 02166 119.38\* 120.41\* 123.36\* 174.18\*

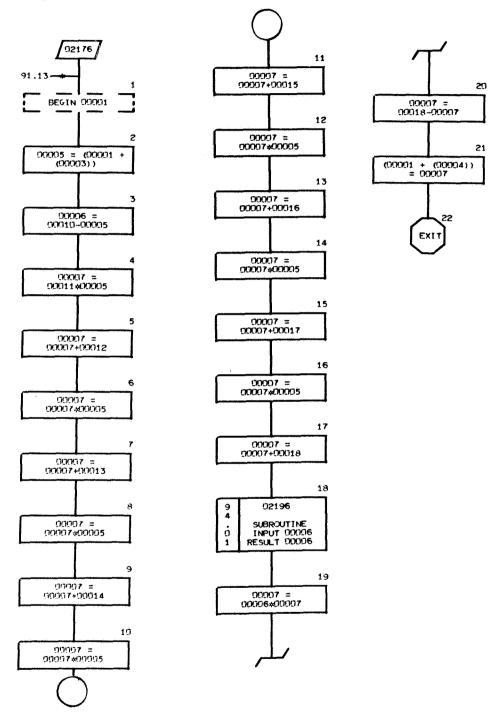
 92.09
 02173
 92.06

 92.11
 02174
 92.04

K VALUE = 02165

V00007+15707963+01

K VALUE = 02175



K VALUE = 02175

V00010+10000000+01

V00011-12624911-02

V00012+66700901-02

V00013-17088125-01

V00014+30891881-01

V00015-50174304-01

V00016+88978987-01

V00017-21459880+00

V00018+15707963+01

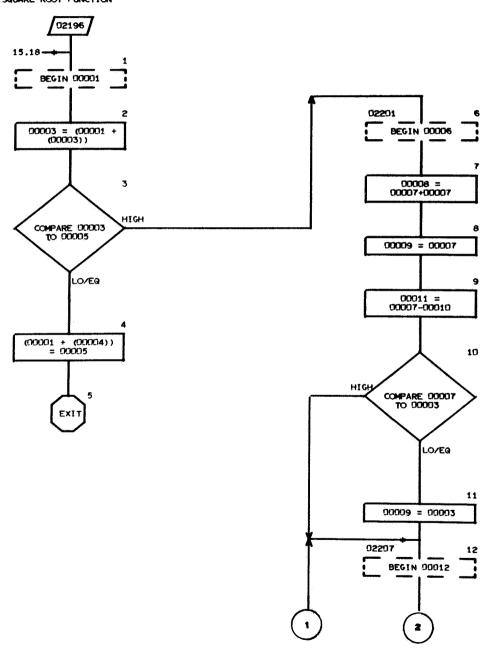
CROSS REFERENCE LISTING

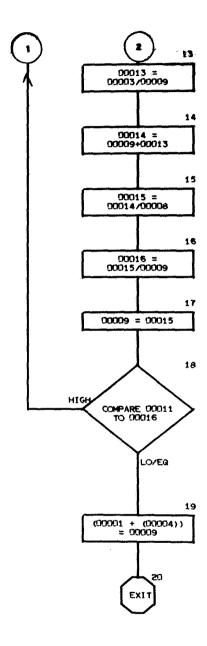
PAGE BOX LABEL REFERENCES

K VALUE = 02175

93.01 02176 91.13\* 92.12\*

#### SQUARE ROOT FUNCTION

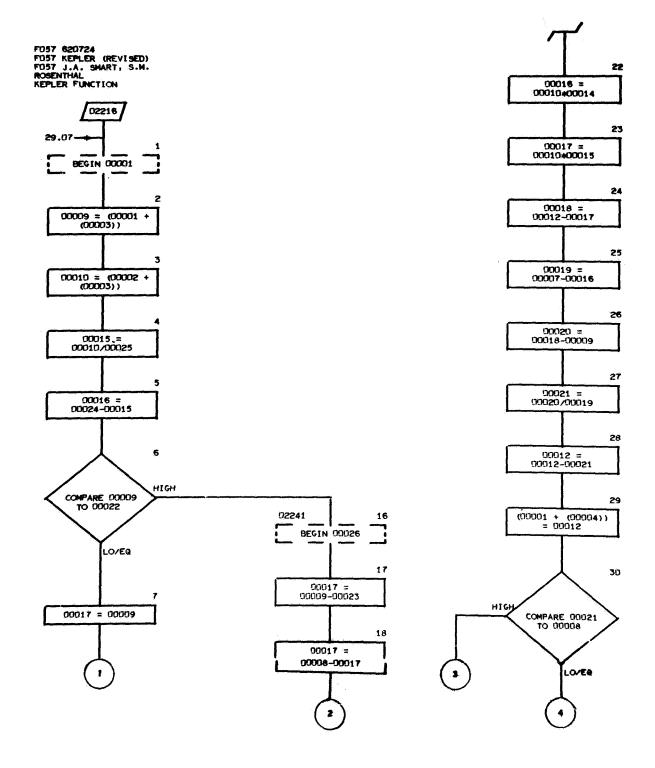


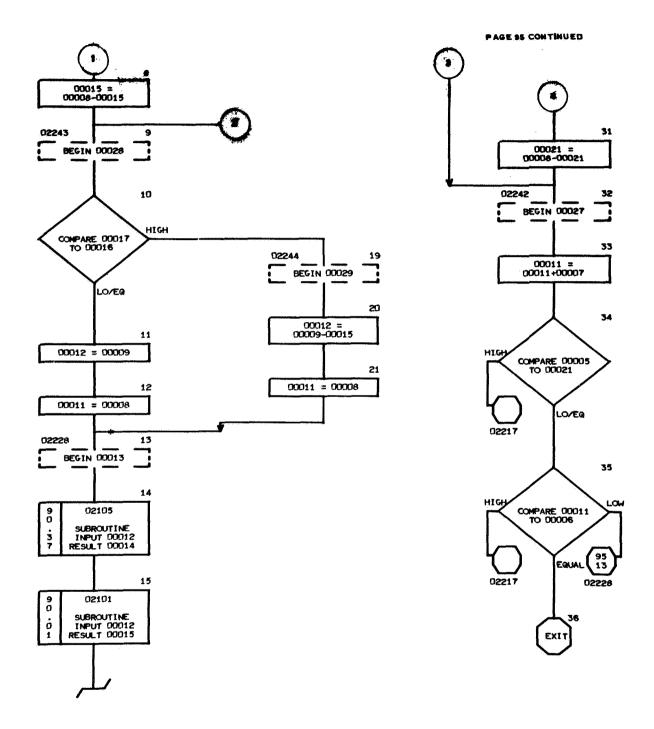


PAGE BOX	LABEL	REFERENCES						
,к. у	VALUE = 02195							
94.01	02196	15.18*	15.24*	27.17*	28.05*	28.10*	29.16+	35.26+
		36.10#	38.25*	*85.66	40.08*	480,98	61.19*	63.36±
		66.44*	71.38*	83.10*	93.18*	100.26+	105.46*	120.23+
		120.38*	120.49*	120.54*	121.16#	121.28+	122.32*	122.42#
		122.47*	145.15*	161.22*	161.46*	163.17#	163.43*	166.28*
		167.05*	171.46*	172.19*	181.12*	181.14*		
94.06	02201	94.03						
94.12	70220	94.10	94.18					

K VALUE = 02195

V00007 10000000 01 V00010 20000000-07





#### K VALUE = 02215

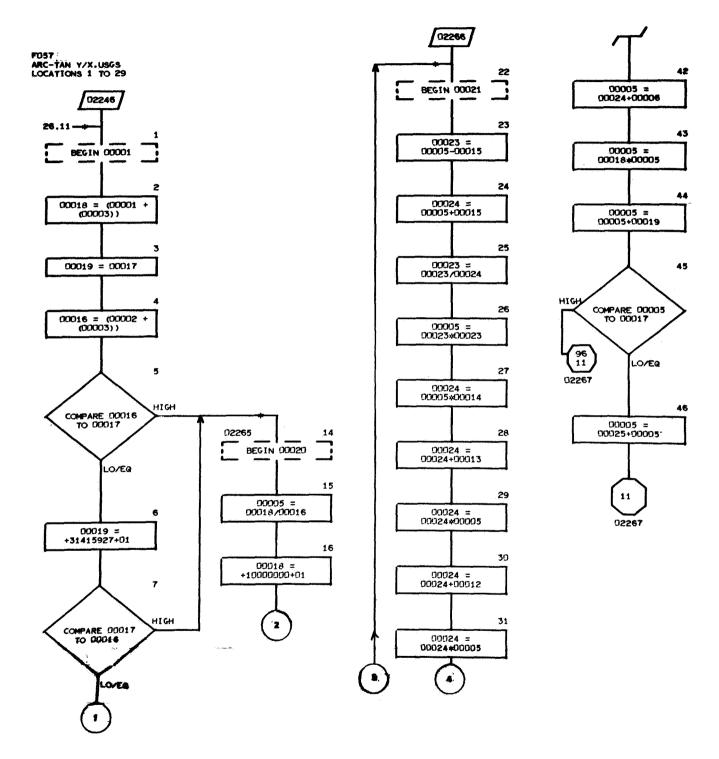
Q0002203839	PI
90002303842	2 PI
Q0002403857	PI / 6
00061202105	COSINE
Q0061402101	SINE

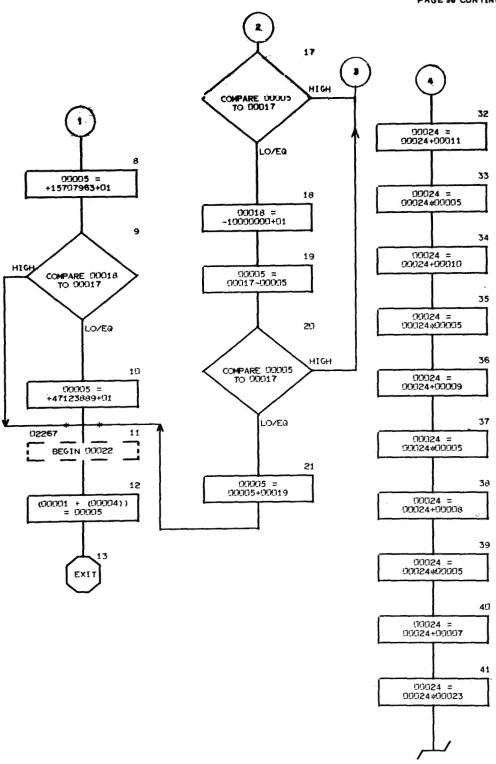
V00005+50000000-08 S.P. CONVERGENCE CRITERIUM

V00006 10000000 02 V00007 10000000 01 V00008 00000000 90 V00025+20000000+01

#### CROSS REFERENCE LISTING

PAGE BOX	LABEL	REF	FERENCES		
K VA	LUE = 02215				
95.01	02216	29.07*	122.21*	169.44*	171.334
95.09	02243	95,18			
95,13	02228	95,21	95.35		
95,16	02241	95.06			
95,19	02244	95.10			
95.32	02242	95.30			





PAGE BOX	LABEL	REI	FERENCES			
K VAL	UE = 02245					
96.01	02246	26.11*	119.45*	120.05*	121.22*	122.36*
96,11	02267	96,09	96.21	96.45	96.46	
96.14	02265	96.05	96.07			
96.22	02266	96.17	96.20			

K VALUE = 02245

V00006+78539816+00

V00007+99999933+00

v00008-33329856+00

V00009+19946536+00

V00010-13908533+00

V90011+96420044-01

V00012-55909886-01

V00013+21861229-01

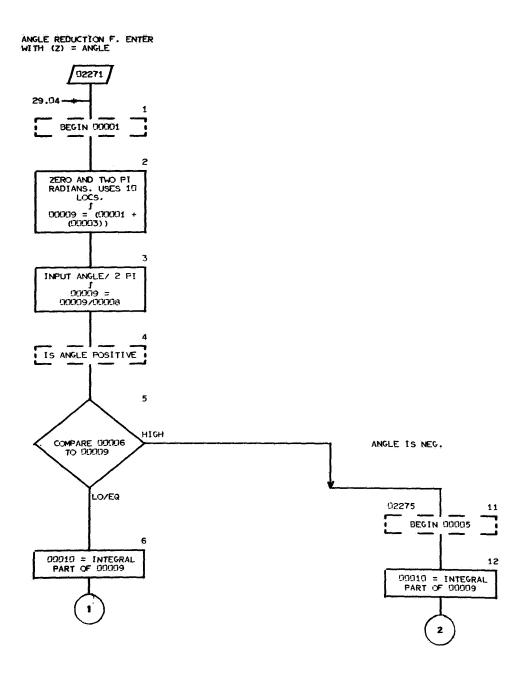
V00014-40540580-02

V00015+100000000+01

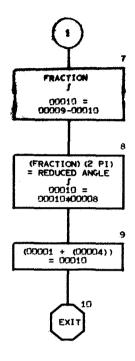
V00017+000000000+00

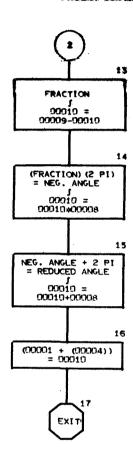
V00025+62831853+01

v00015+100000000+01



#### PAGE 97 CONTINUED





PAGE BOX LABEL REFERENCES

k VALUE = 02270

97.01 02271 29.04\* 105.03\* 105.05\* 105.07\* 120.06\* 121.26\* 122.30\*

158.06\* 158.20\* 158.25\* 168.10\* 168.12\* 168.19\* 168.25\*

168.26\* 168.27\* 168.31\* 168.32\* 168.33\* 169.26\* 169.27\*

169.28\* 171.27\* 171.28\* 171.29\* 171.34\*

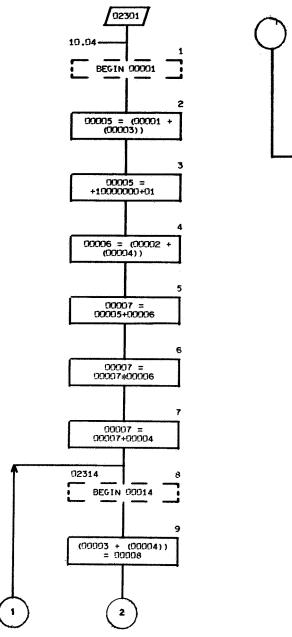
97.11 02275 97.05

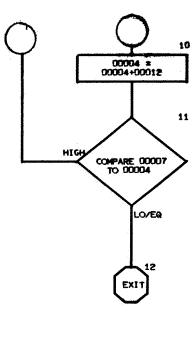
K VALUE = 02270

Q0000803842 2 PI

V00006+00000000+00 IN RADIANS. EXIT WITH (X) = ANGLE B

# MATRIX CLEAR FUNCTION





# NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI

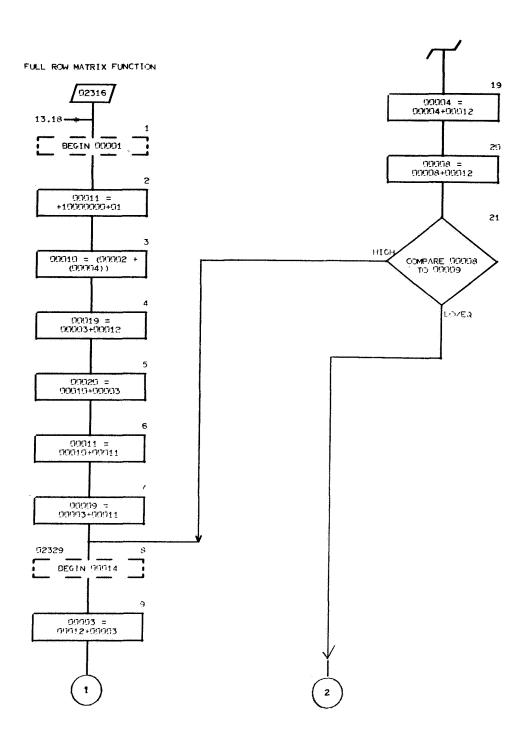
PAGE BOX LABEL REFERENCES

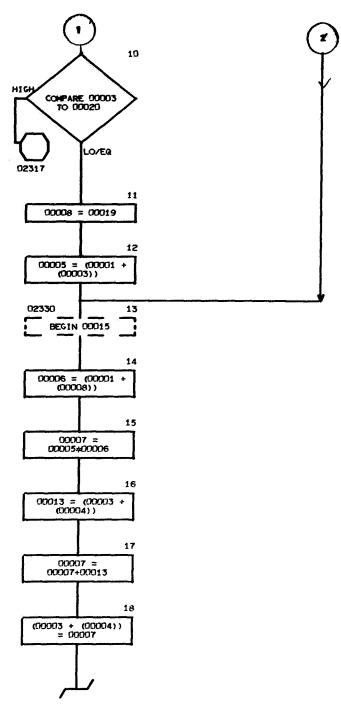
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98.01 02301 10.04\* 98.08 02314 98.11

K VALUE = 02300

V00008+00000000+00 V00012+10000000+01





PAGE BOX LABEL REFERENCES

K VALUE = 02315

99.01 02316 13.18+ 152.18+

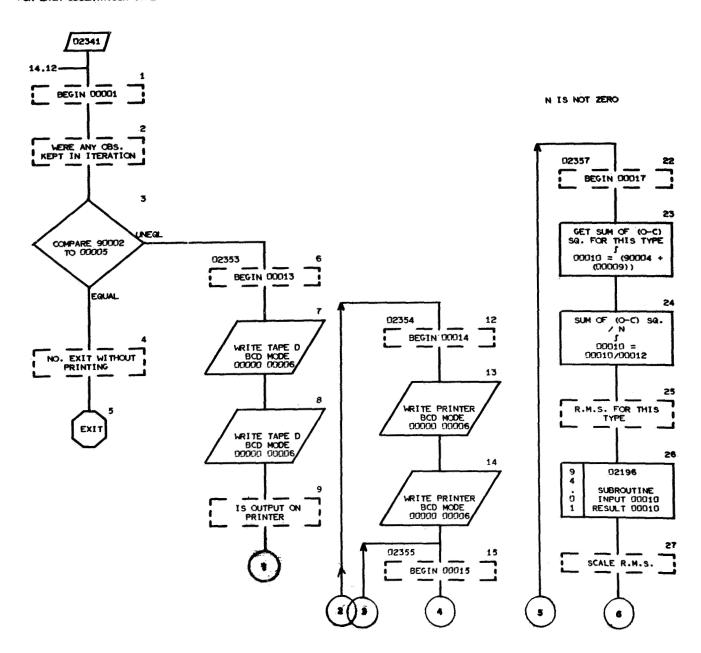
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 02329
 99.21

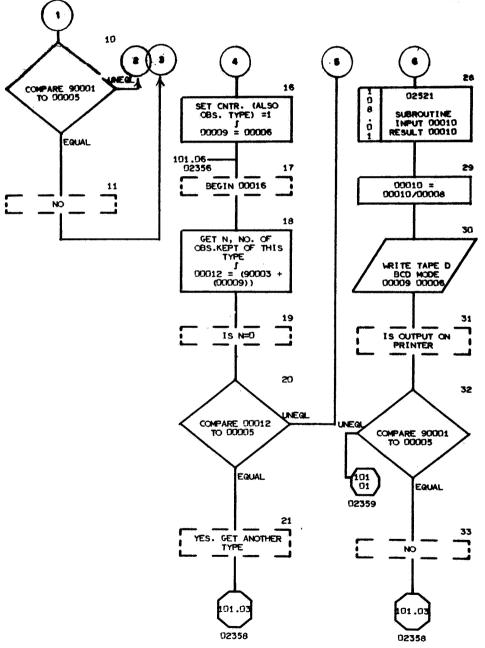
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 02330
 99.21

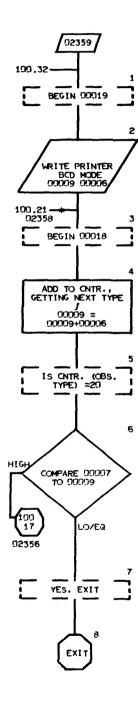
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V00012+10000000+01

# COMPUTE AND PRINT R.M.S. FOR EACH OBSERVATION TYPE







#### PAGE VII

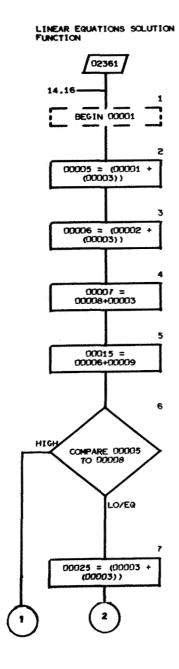
# NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI CROSS-REFERENCE LISTING

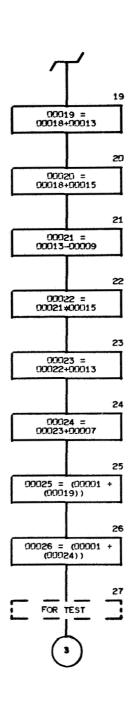
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100.17	02356	101.06
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101.01	02359	190.32
101.03	02358	100.21 100.33
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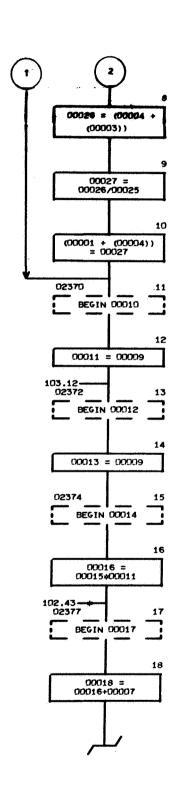
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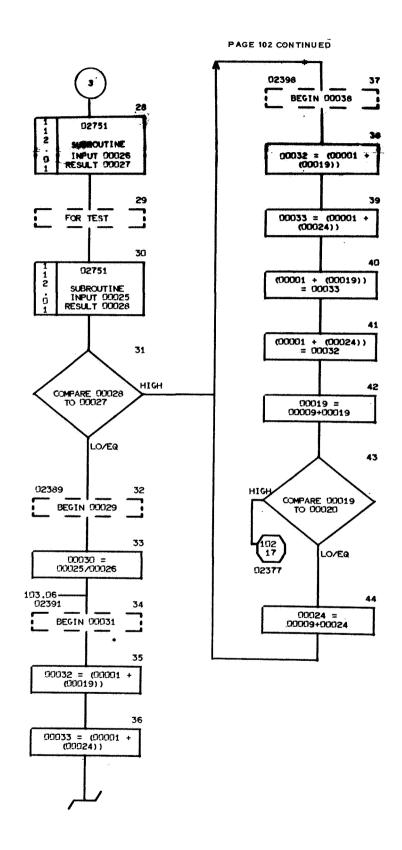
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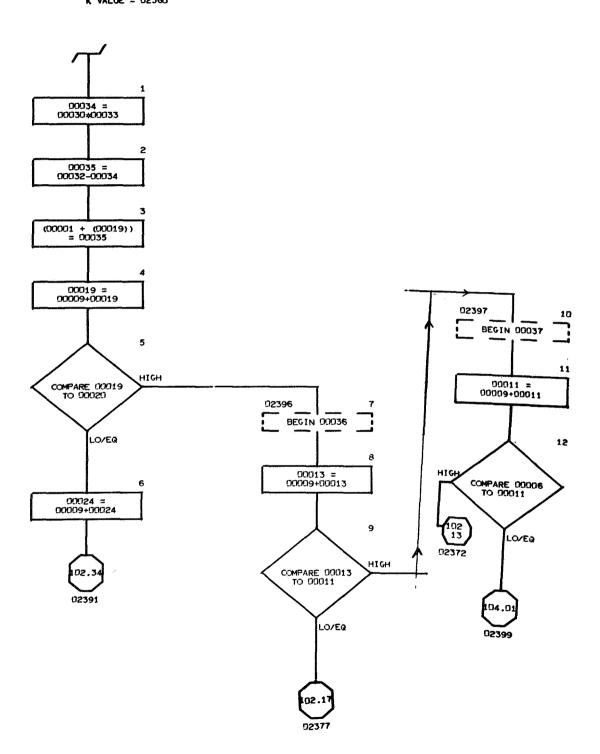
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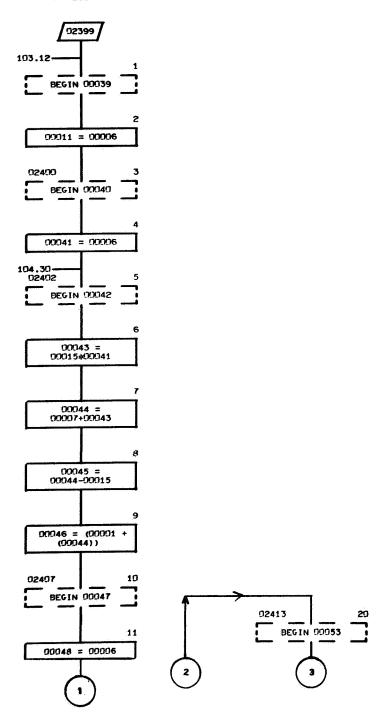


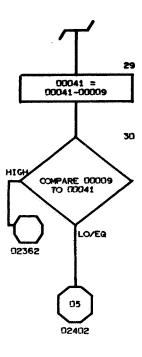


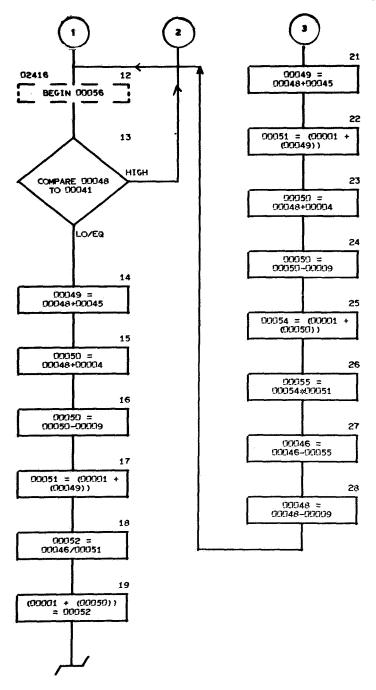




K VALUE = 02360







### CROSS REFERENCE LISTING

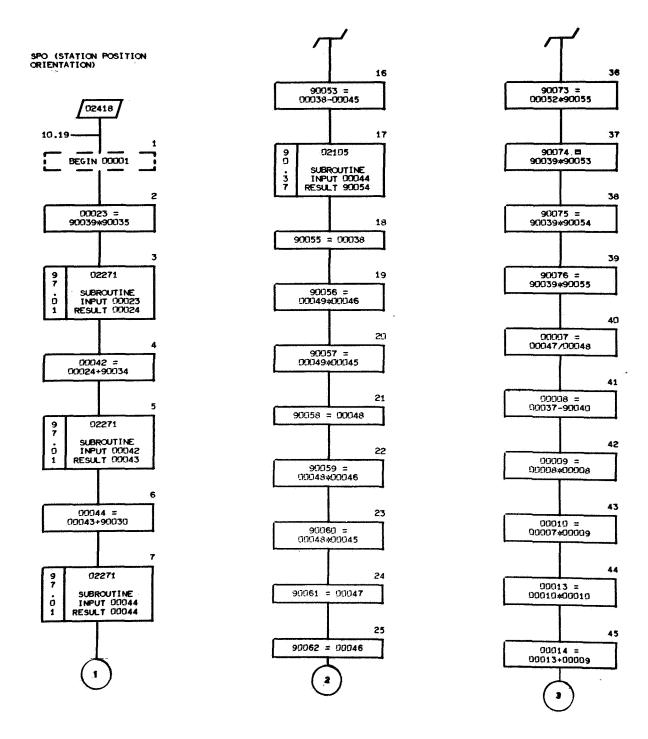
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102.13	02372	193.12	
102.15	02374		
102.17	02377	102.43	103.09
102.32	02389		
102.34	02391	193,96	
102.37	02398	102.31	102.44
103.07	02396	103.05	
103.10	02397	103.09	
104.01	02399	103.12	
104.03	02400		
104.05	02402	104.30	
104.10	02407		
104.12	02416	104.28	
194.20	02413	104.13	

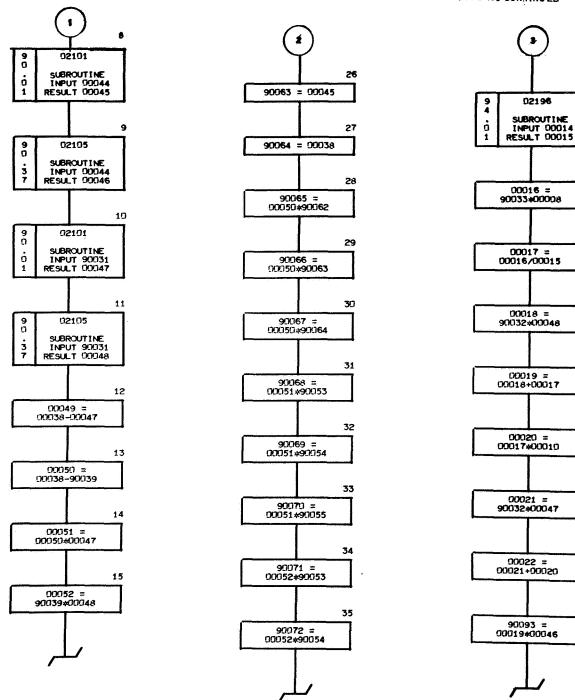
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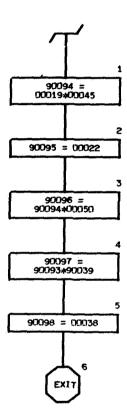
09000102751

ABSOLUTE VALUE FUNCTION

V00008+100000000+01 V00009+10000000+01







#### PAGE 106 CONTINUED

K VALUE = 02417

@9006200325

Q9006300326

99003000302 STATION LONGITUDE (RADIANS) 09003100303 STATION LATITUDE (RADIANS) 99003200304 STATION HEIGHT (C.U.L.) 09003300305 DIST.FROM EARTH CENTER TO STATION (C 99003400298 LAMBOA SUB ZERO (RADIANS) Q9003500200 T, OBSERVATION TIME IN C.U.T. Q9003903835 ROTATION OF EARTH (RAD/C.U.T.) 09004003834 FLATNESS OF EARTH 99005300316 G 1 BAR 09005400317 Q9005500318 09005600319 G 2 BAR 09005700320 99005800321 09005900322 G 3 BAR 09006000323 **Q9006100324** 

G 4 BAR

#### PAGE 10A

09006400327 09006500337 G 1 DOT BAR Q9006600338 99006700339 99006800340 G 2 DOT BAR 09006900341 99007000342 Q9007100343 G 3 DOT BAR 09007200344 Q9007300345 09007400346 G 4 DOT BAR 09007500347 99007600348 99007700328 G 5 BAR 09007800329 Q9007900**33**0 09008000331 G 6 BAR 99008100332 @9008200333 Q9008300**334** G 7 BAR Q9008400**335** 09008500336 G 5 DOT BAR Q9002000349 09002100350 Q9002200351 G 6 DOT BAR Q9002300352 Q9002400353 Q9002500354 G 7 DOT BAR Q9002600**35**5 09002700356 Q9002800357 Q9009300**31**0 CAP R BAR, STATION POSITION VECTOR 09009400311 09009500312 Q9009600313 CAP R DOT BAR, STATION VELOCITY VECT

#### NASA-GSEC MISSION AND TRAJECTORY ANALYSIS DIVI

**Q9**009700314

99009800315

Q9008602271 ANGLE REDUCER

Q9061202105 COSINE Q9061402101 SINE

99062202196 SQUARE ROOT

Q0000503842 2 PI

V90077+100000000+01

V90078+0000000000+00

V90079+000000000+00

V90080+000000000+00

v90081+10000000+01

V90082+000000000+00

V90083+000000000+00

V90084+000000000+00

V90085+100000000+01

V90020+000000000+00 G 5 DOT BAR

V90021+0000000000+00

V90022+000000000+00

V90023+000000000+00 G 6 DOT BAR

V90024+000000000+00

V90025+000000000+00

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v90028+000000000+00

V00037+100000000+01

V00038+000000000+00

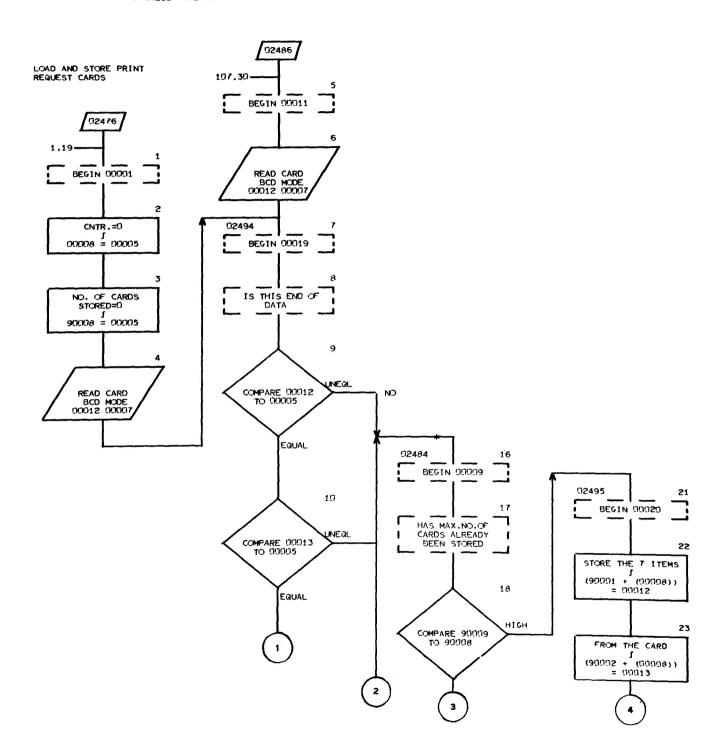
V00049+0000000000+00

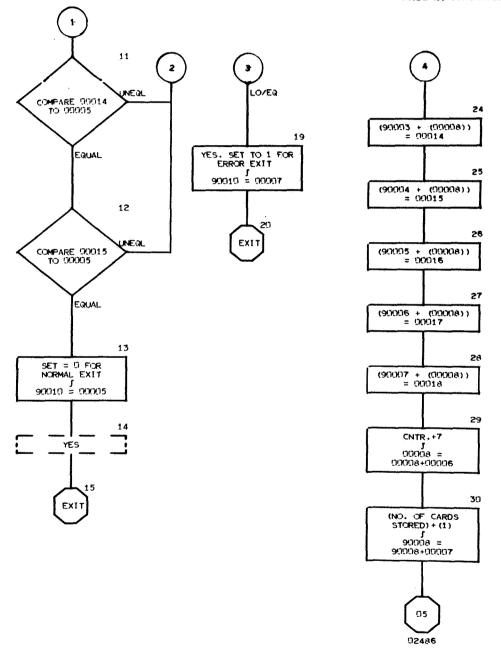
CROSS REFERENCE LISTING

PAGE BOX LABEL REFERENCES

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105.01 02418 10.19\*





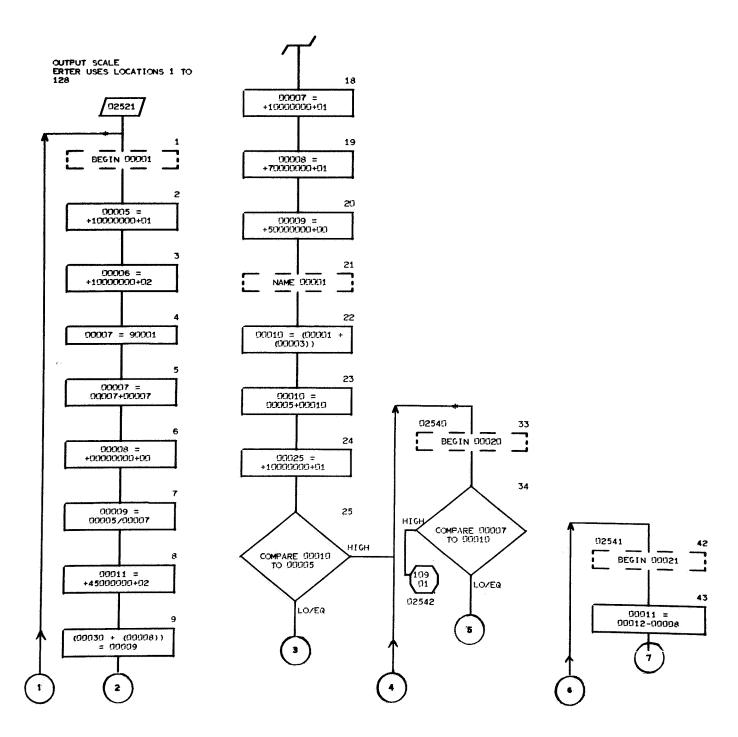
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99000203101	2 ND STG. LOC.
99000303102	3 RD STG. LOC.
Q9000403103	4 TH STG. LOC.
99000503104	5 TH STG. LOC.
Q9000603105	6 TH STG. LOC.
Q9000703106	7 TH STG. LOC.
99000800060	NO. OF CARDS STORED
Q9000900495	MAX.NO. OF CARDS ALLOWED
Q9001000191	ERROR EXIT
	CLOADS IN CARDS AND STORES

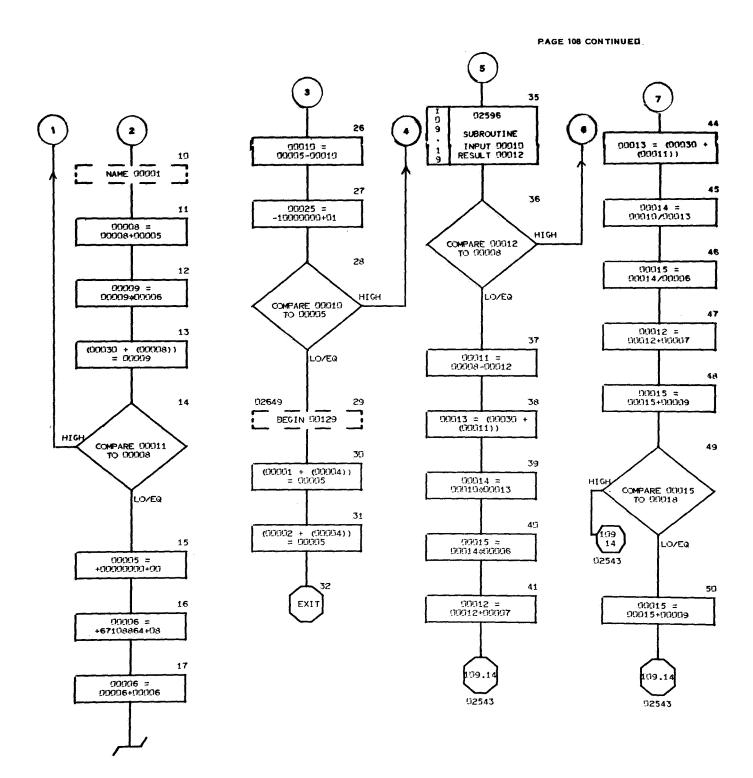
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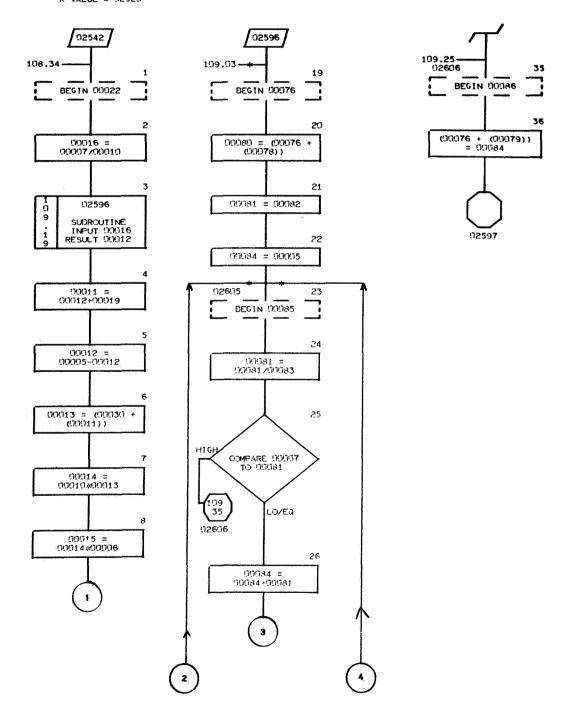
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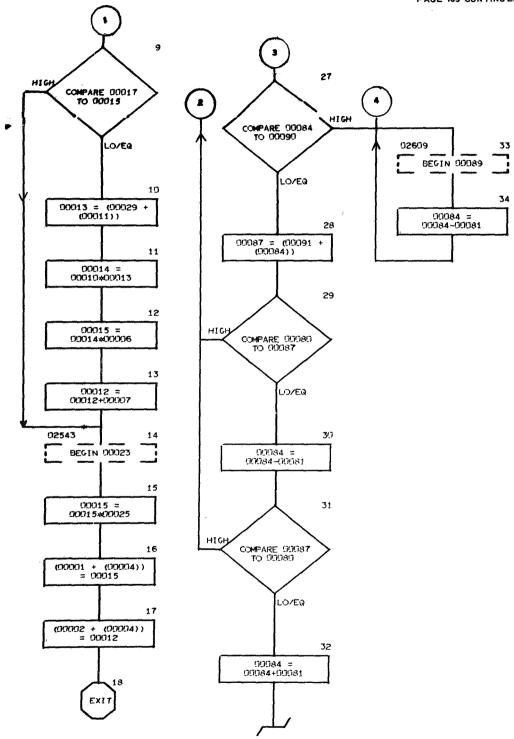
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107.05	02486	107.30			
107.07	D2 <b>494</b>	107.04			
107.16	02484	107.09	107.10	107.11	107.12
107.21	D2495	107.18			









## NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI

PAGE 11 A

#### MISCELLANEOUS STATEMENTS

V00113+10000000+23

V00114+100000000+24

V00115+100000000+25

V00116+10000000+26

V00117+10000000+27

V00118+10000000+28

V00119+100000000+29

V00120+10000000+30

V00121+10000000+31

V00122+100000000+32

V00123+100000000+33

V00124+100000000+34

V00125+10000000+35

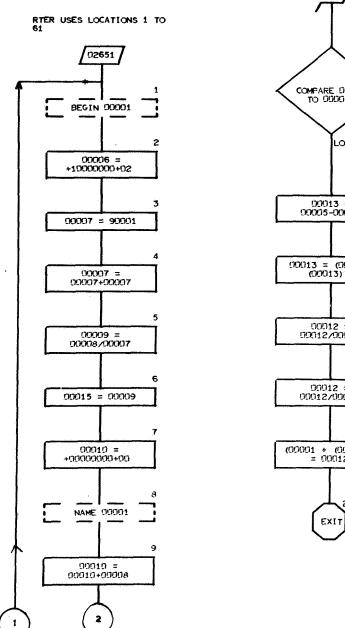
V00126+100000000+36 V00127+100000000+37

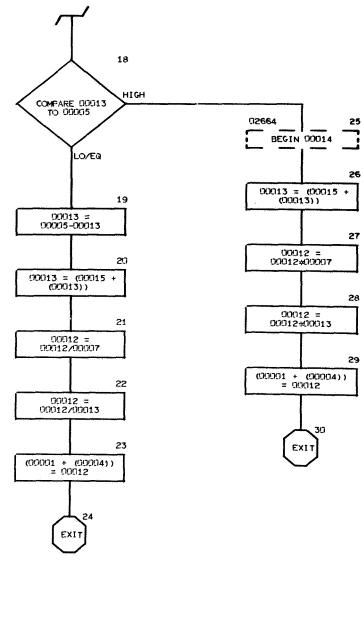
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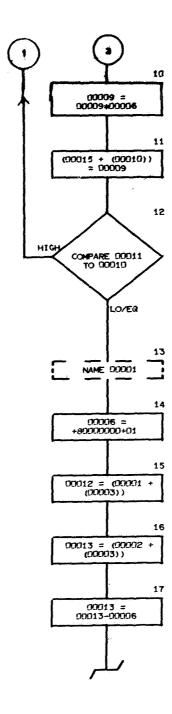
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		18.29*	20.19*	20.22*	20.25*	20.29*	100.28*	108.14
		137.15*	137.17*	137.19*	137.21*	137.23*	137.25#	138.10*
		138.12*	138.14*	138.16*	138.18*	138.20*	138.27#	138.29+
		138.31*	138.33*	138.35*	138.37*	139.08*	139.10*	139.30+
		139.32*	140.93#	140.05*	140.07*	140.09#	160.08#	100.104
		160.12*	160.14*	160.24*	160.26*	160.28*	160.30+	
108.29	02649							
108.33	02540	108.25	108.28					
108.42	02541	108.36						
109.01	02542	108.34						
109.14	02543	108.41	108.49	108.50	109.09			
109.19	02596	108.35*	109.03*					
109.23	02605	109.29	109.31	109.34				
109.33	02609	199.27						
109.35	02606	109.25						

Q9000103841 2\*\*26 V00017+100000000+09 V00018+99999999+07 V00019+80000000+01 V00028+10000000+08 V00082+64000000+02 V00083+200000000+01 V00090+37000000+02 V00091+100000000+01 V00092+100000000+02 V00093+10000000+03 V00094+100000000+04 V00095+100000000+05 V00096+100000000+06 V00097+100000000+07 V00098+10000000+08 V00099+100000000+09 V00100+100000000+10 V00101+100000000+11 V00102+100000000+12 V00103+100000000+13 V00104+100000000+14 V00105+100000000+15 V00106+100000000+16 V00107+100000000+17 V00108+100000000+18 VD0109+100000000+19 V00110+100000000+20 V00111+100000000+21

V00112+100000000+22







#### CROSS REFERENCE LISTING

PAGE BOX LABEL REFERENCES

K VALUE = 02650

110,01 02651 5.29\* 5.30\* 5.31\* 5.32\* 5.33\* 110.12 113.13\*

119.07\* 119.08\* 119.09\* 119.11\* 119.12\* 119.13\* 122.07\* 122.09\* 122.11\* 122.13\* 122.15\* 122.17\* 125.37\* 125.39\*

125.41\* 125,43\* 125.45\* 126.02\* 141.04\* 141.06\* 141.08\*

141.10\* 141.12\* 141.14\* 149.33\* 149.34\* 151.05\* 151.07\*

151.09\* 151.11\* 151.13\* 151.15\* 151.17\* 151.19\* 151.21\*

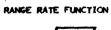
110.25 02664 110.18

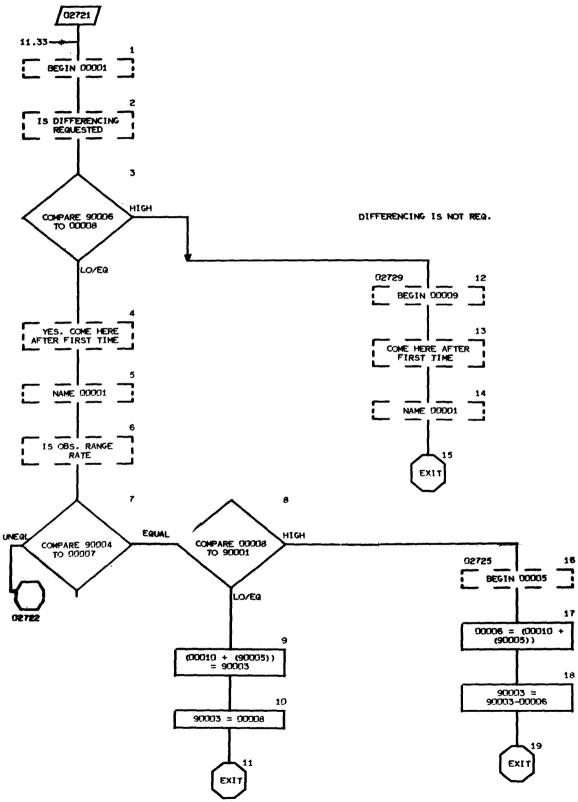
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V90005+000000000+00 V90008+10000000+01

V00011+460000000+02





Q9000100246

99000200258

9000300392

99000400253

Q9000500069

09000602098 DIFF. IND. (DIFF. IF IND.=0)

V00007+90000000+01

V00008+000000000+00

CROSS REFERENCE LISTING

PAGE BOX LABEL REFERENCES

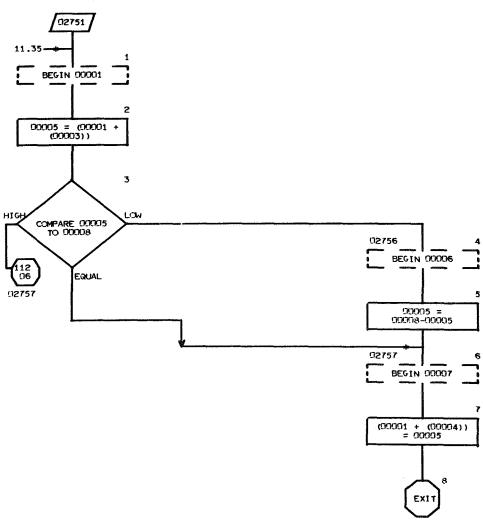
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111.01 02721 11.33\* 13.11\*

 111.12
 02729
 111.03

 111.16
 02725
 111.08

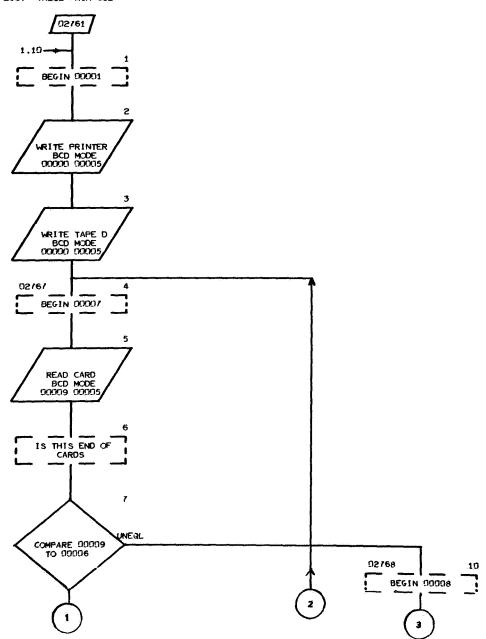
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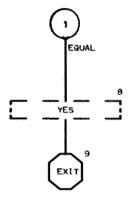


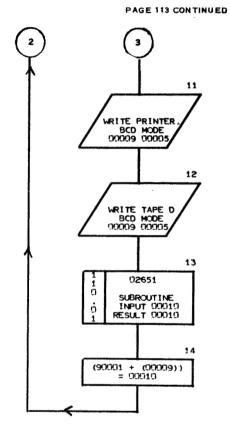
## V00008+00000000+00

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112.01	02751	11.35*	34.03*	42.20*	42.35*	43.09*	43.24*	43.39*
		44.09*	44.24*	44.39*	45.09*	45.24*	45.39*	46.09+
		46.24*	46.39*	47.09*	47.24*	47.39*	48.09*	48.24*
		48.39*	49.09*	49.26*	49.43*	50.09*	50.26*	50.43*
		51.09*	51.26*	51.43*	52.09*	52.26*	52.43*	53.09*
		53.26*	53.43*	54.09*	54.26*	54.43*	55.09*	55.26*
		102.28*	102.30*	120.51*	121.01*	158.13*	158.15*	179.13*
		179.22*	183.14*					
112.04	02756	112.03						
112.06	02757	112.03	112.03					

# ONE-WORD LOAD (PRINTS







## NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI

PAGE VIII

CROSS-RET	ERENCE	LISTING

PAGE BOX	LABEL	REF	ERENCES	ENCES		
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113.04	02767	113,14				
113.10	02768	113,07				

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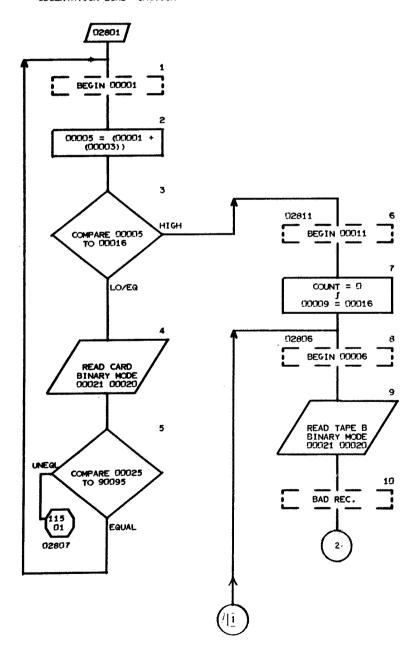
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Q9DDD2D2651 INPUT CONVERTER

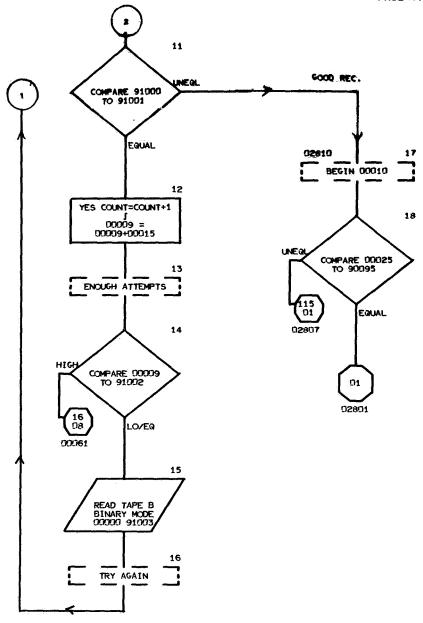
 V00005+10000000+01
 1-17 + ALPHABETIC IDENT.FROM COLS.3

 V00006+00000000+00
 OF EACH CARD LOADED. USES 19 LOCS.)

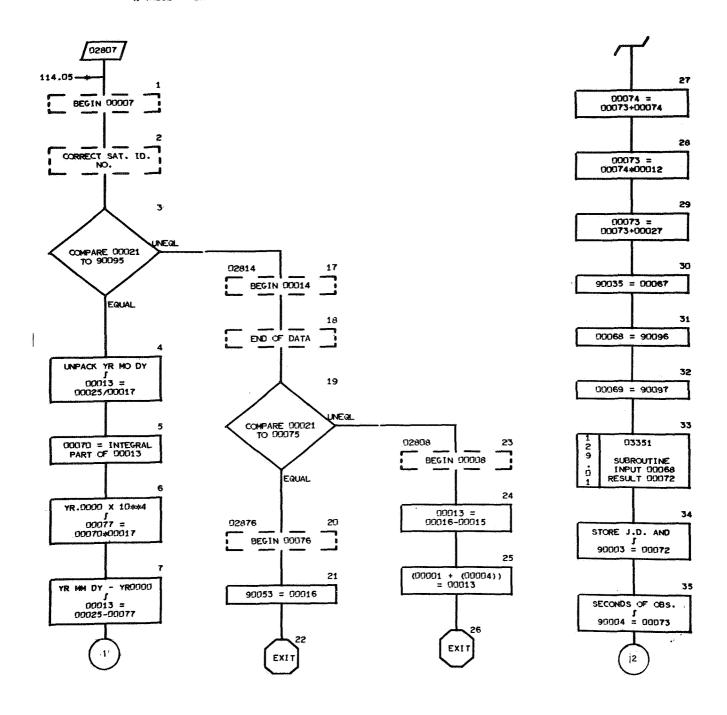
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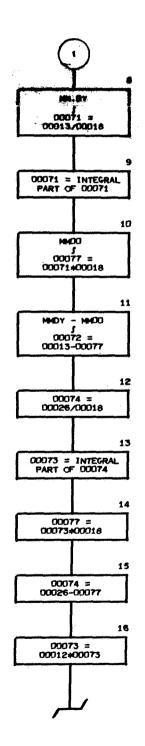


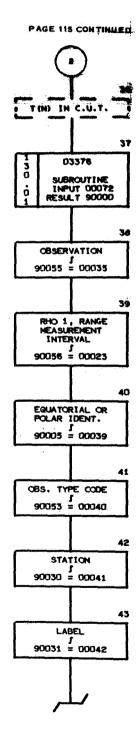




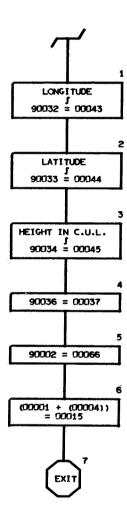
NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI K VALUE = 02800







## MASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI K VALUE = 02800



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K VAL	UE = 02800			
114.01	02801	10.07*	114.05	114,18
114.06	02811	114.03		
114.08	02806	114,16		
114.17	02810	114,11		
115.01	02807	114,05	114.18	
115.17	02814	115.03		
115.20	02876			
115.23	02808	115.19		

(N) T 0050000068

99000200245 OBSERVATION COUNTER

99000300378 J.D. TIME OF 99000400379 SECONDS OBS.

99000700007

Q9003000300 STATION LABEL

Q9003100301

 Q9003200302
 LONGITUDE

 Q9003300303
 LATITUDE

 Q9003400304
 HEIGHT

Q9003500246 RANGE RATE CODE

Q9003600258

99000500497 EQUATORIAL OR POLAR IDENT.

99005300253 OBS. TYPE CODE
99005500255 OBSERVATION

99005600396 RHO 1, RANGE MEASUREMENT INTERVAL IN

Q9009500295 SAT. ID. Q9009600296 YREF Q9009700297 DREF

Q9009803351 OBS. DATE TO DAY COUNT

Q9009903376 JULIAN DAYS-SECONDS TO C.U.T.

Q9100000000 LOC. ZERO
Q9100100459 BAD RECORD IND.
Q9100200460 NO.OF READ ATTEMPTS

Q9100303812 MINUS ONE (TO BACKSPACE ONE RECORD)
Q9001000061 TRANSFER POINT FOR BAD TAPE ON TB

V00012+60000000+02

 V00015+10000000+01
 ONE

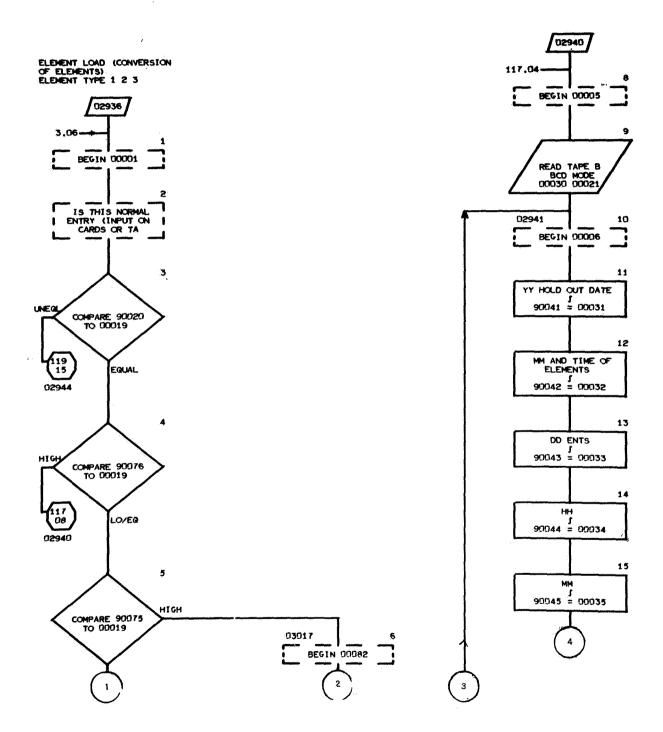
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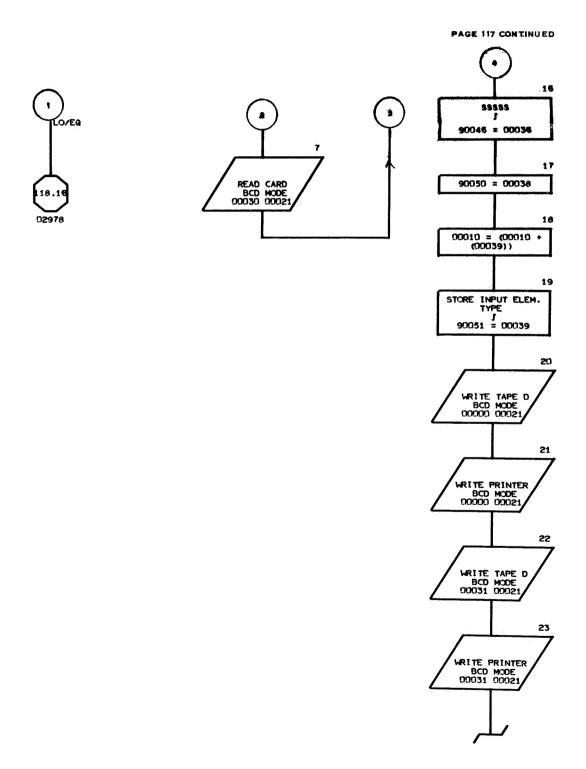
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 10\*\*\*04

 V00018+100000000+03
 10\*\*\*02

V00020+48000000+02 NO. OF INPUT WORDS

V00075+99999999+08 END SENTINEL



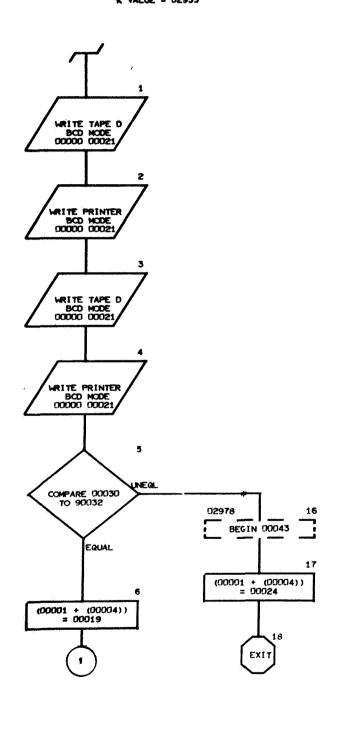


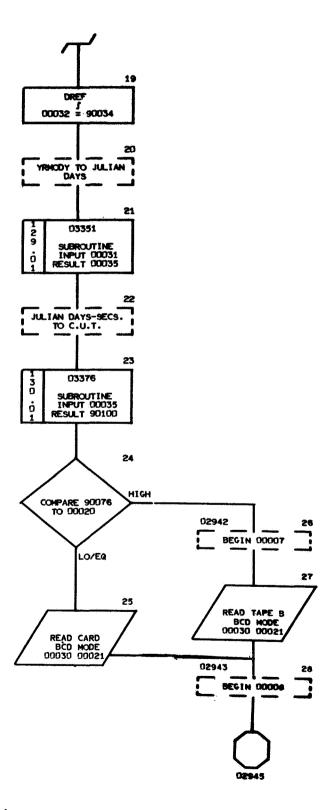
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117.06	03017	117.05		
117.08	02940	117.04		
117.10	02941	117.07		
118.16	02978	117.05	118.05	
118.26	02942	118.24		
118.28	02943	118.25		
119.01	02946			
119.15	02944	117.03		
120.32	03020	120.25		
121.04	03016	121.02		
121.06	02964	120.42		
121.09	02949	120.42	121.03	121.05
122.01	02947			
122.41	02952	126.03		
123.28	02953	123.27		
123.40	02979	123.27		
125.02	03005	120.40	121.30	
125.30	02948			
125.35	03026			

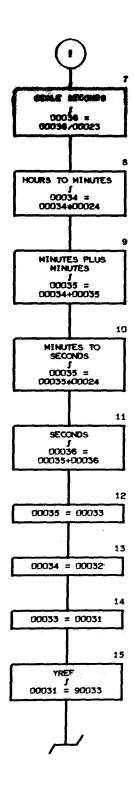
W ANDRE - TOOLS	A, SEMI-MAJOR AXIS IN C.U.L.	0
Q9000101101 Q9000201102	• • • • • • • • • • • • • • • • • • • •	0
		Ö
99000401104	R SUB I SATELLITE POSITION	
Q9000501105	R SUB J VECTOR IN C.U.L.	ı
Q9000601106	R SUB K	1
Q9000701107	V SUB I SATELLITE VELOCITY	1
Q9000801108	V SUB J VECTOR IN CUL/CUT	1
99000901109	V SUB K	I
99001001110	R D, MAG. OF R BAR	0
Q9001101111		0
Q9001201112	ANGLE DELTA IN RADIANS	0
Q9001301113	M, MEAN ANOMALY IN RADIANS	0
Q9001401114	PHI, ORBITAL AZIMUTH IN RADIANS	0
Q9001501115	SMALL OMEGA, ARG. OF PERIGEE IN RAD	o
Q9001601116	I, INCLINATION IN RADIANS	0
Q9001701117	CAP OMEGA, LONG, OF ASC, NODE IN RAD	0
Q9001801118	THETA (ELEVATION V) IN RADIANS	0
Q9001901119	N, MEAN MOTION IN RAD/C.U.T.	0
Q900200090	NORMAL OR SPECIAL ENTRY IND.	I
Q9002203852	MU	1
Q9002702651	INPUT CONVERTER	F
Q9002903864	J	1
Q9003003351	OBSERVED DATE TO J.D.	F
Q9003103376	JULIAN DAYS, SECONDS TO C.U.T.	F
Q9003200295	SATELLITE ID. NO.	1
Q9003300296	YEAR OF REFERENCE (LAST 2 DIGITS)	ī
Q9003400297	DAYS JAN.1 - DAY OF REFERENCE	1
Q9003501123	PERIOD IN C.U.T.	0
Q9003601125	HEIGHT OF APOGEE IN C.U.L.	0
99003701122	CAP OMEGA DOT IN RADIANS / CUT	0
Q9003801121	SMALL OMEGA DOT IN RADIANS/CUT	0
Q9003901124	HEIGHT OF PERIGEE IN C.U.L.	0
Q9004001120	CAP E, ECCENTRIC ANOMALY IN RAD	o,
Q9004101190	YEAR DATE AND TIME	0
Q9004201191	MONTH OF ELEMENTS	o
99004301192	DAY	0
Q9004401193	HOUR	0
99004501194	MINUTE	o

# WASA-GSFC HISSION AND TRAJECTORY AMALYSIS DIVI

	-	
<b>00</b> 004601195	(SECONDS) (1000)	0
<b>090050000065</b>	COL.69 OF TIME CARD SPERT-OPTIONS	0
49005100096	TYPE OF INPUT ELEMENTS	0
99007500044	CARD INPUT OPTION ING.	ı
99007600040	TAPE INPUT OPTION IND.	I
99007800042	ON-LINE PRINTOUT OPTION IND.	t
99007902751	ABSOLUTE VALUE	F
99008002271	ANGLE REDUCER (0 TO 2 PI)	F
99006102011	VECTOR HAGNITUDE	F
Q9008202021	VECTOR DIRECTION	F
99006302051	DOT PRODUCT	F
99008402087	VQ + 2	0
Q9008502085	VQ	0
99008602001	VECTOR MOVE	F
99008702076	SCALAR-VECTOR MULTIPLY	F
Q9008802051	CROSS PRODUCT	F
Q9DD89D2D31	VECTOR ADD	F
Q9009002041	VECTOR SUBTRACT	F
Q9009102101	SIN	F
Q9009202105	cos	F
Q9009302196	SQUARE ROOT	F
Q9009402216	KEPLER	F
Q9009502246	ARC TAN (Y/X)	F
Q9009602166	ARC COS	F
Q9009702156	ARC SIN	F
Q9010001100	T (0), TIME OF ELEMENTS IN CUT	0
Q0002203842	2 Pİ	İ
Q0007303839	Iq	Ī
V00019+00000000+00		
00+0000000+05		
V00021+100000000+01		
V09923+19000000+04		
V00024+60000000+02		
V00026+100000000+01		
V00027+000000000+00		
V00028+00000000+00		
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V00071+40000000+01		
V00072+50000000+01		

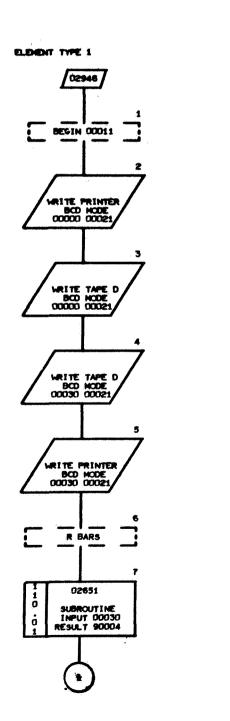


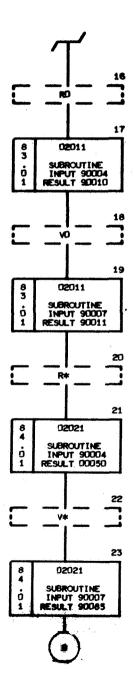


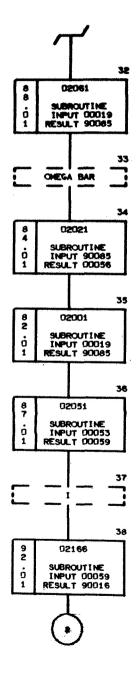


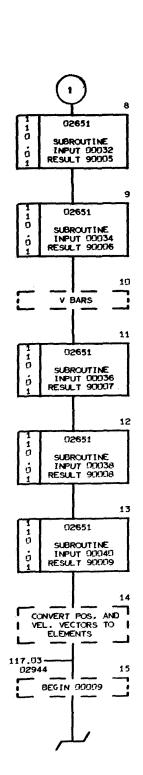
#### PAGE 119

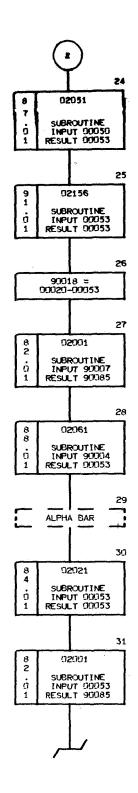
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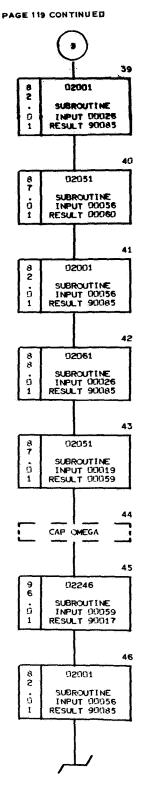


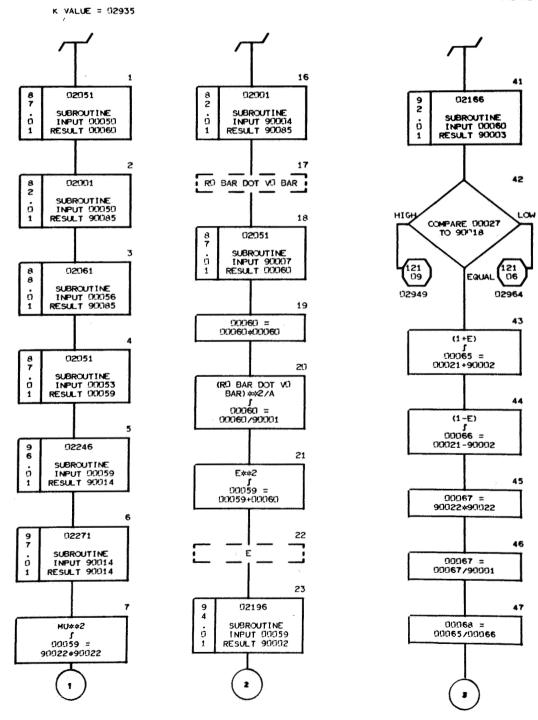




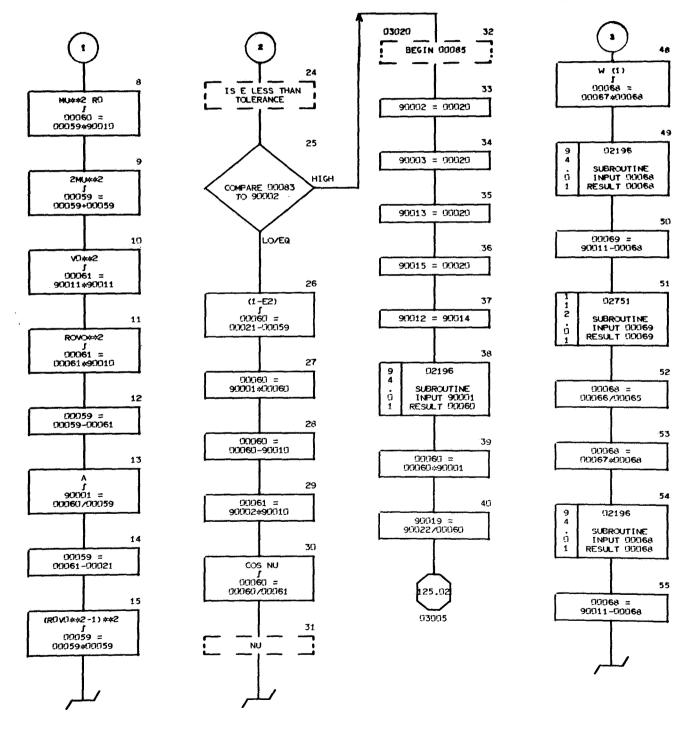


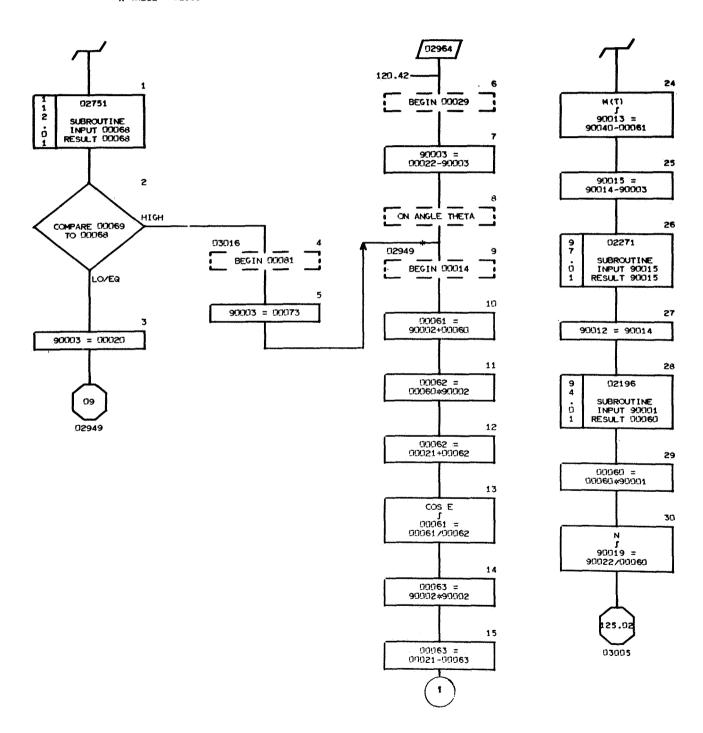


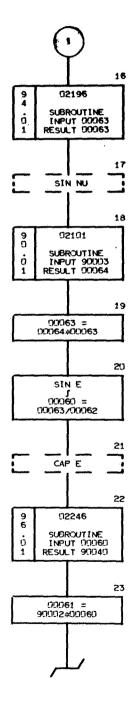




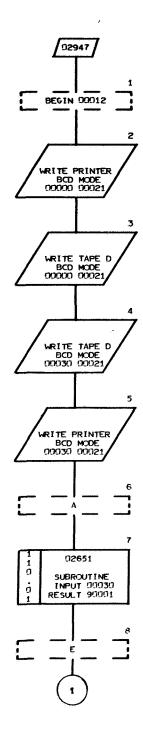
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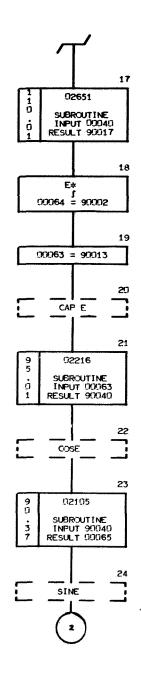


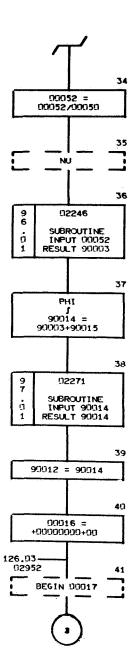


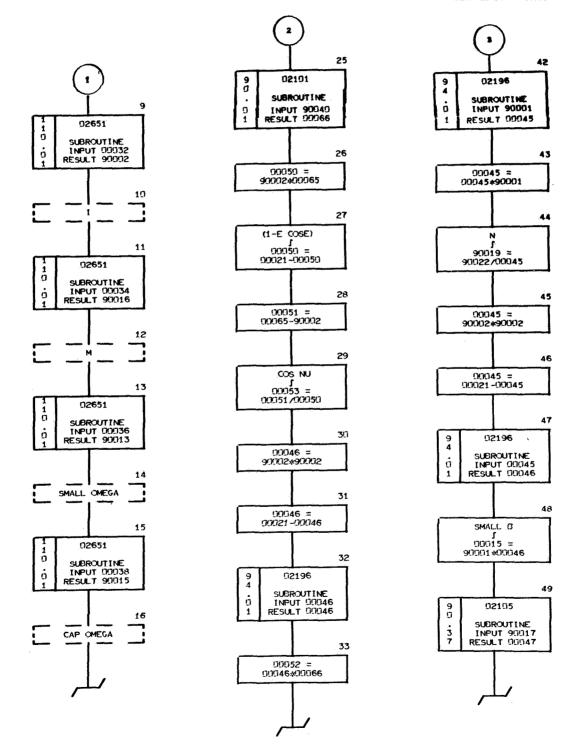


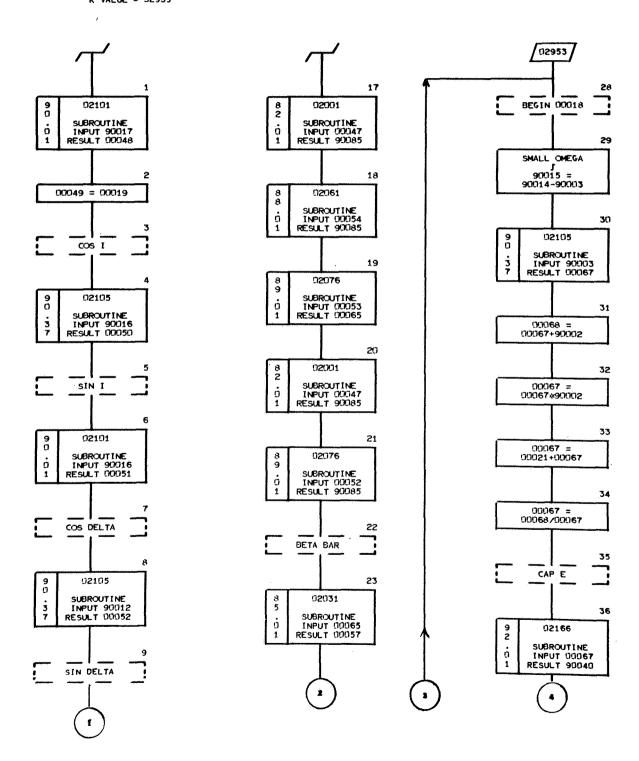
NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI K VALUE = 92935

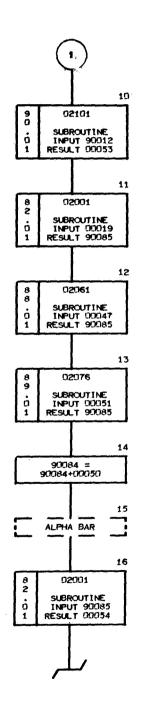


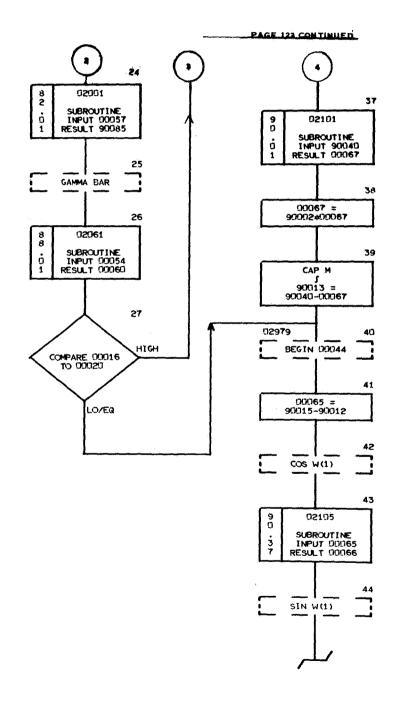


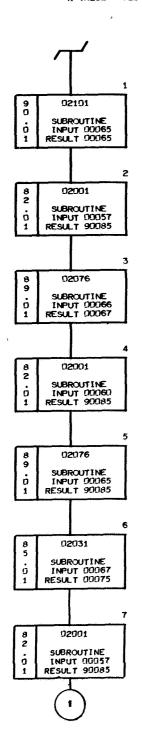


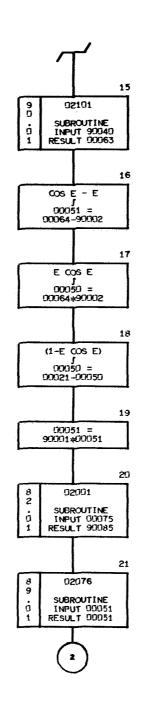


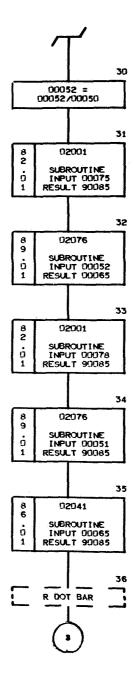




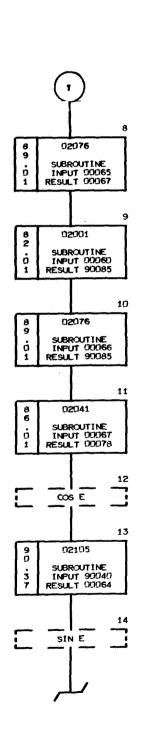


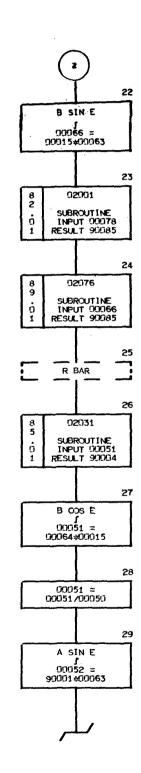


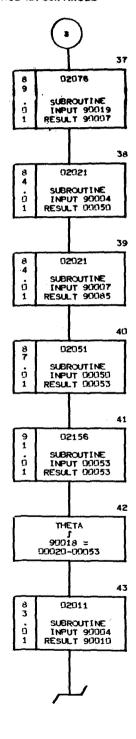




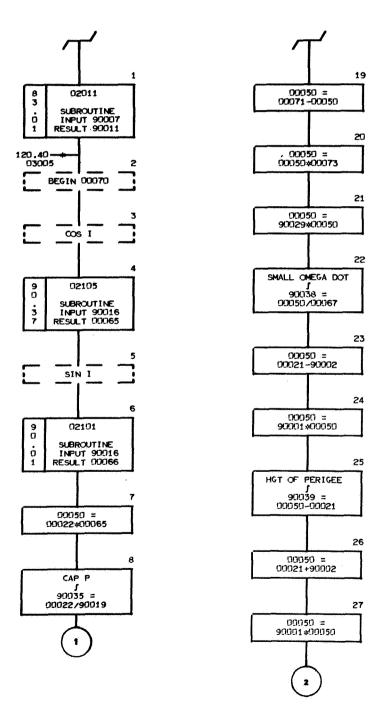
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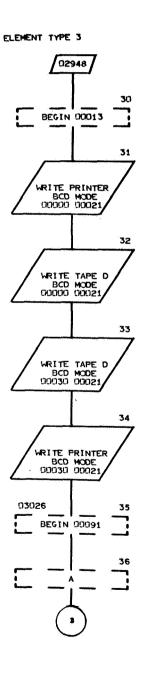


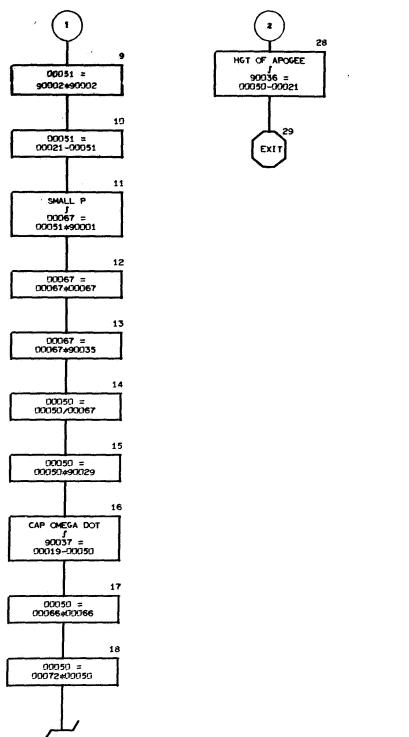


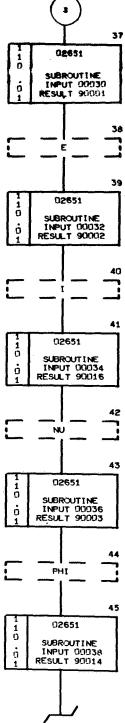


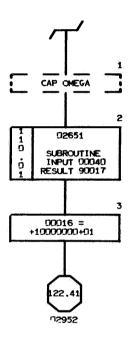
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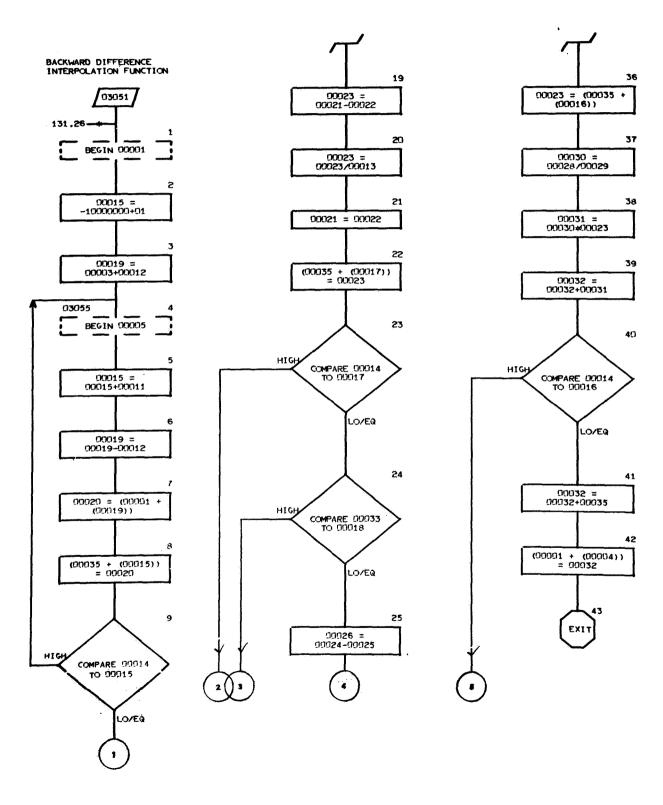


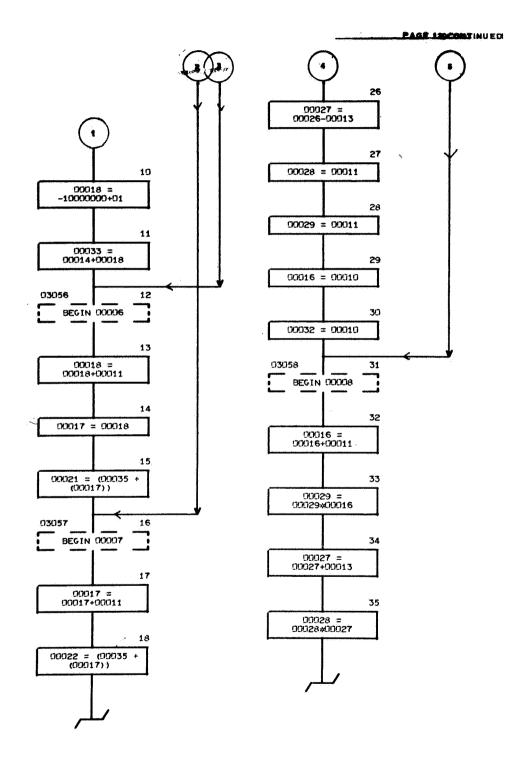












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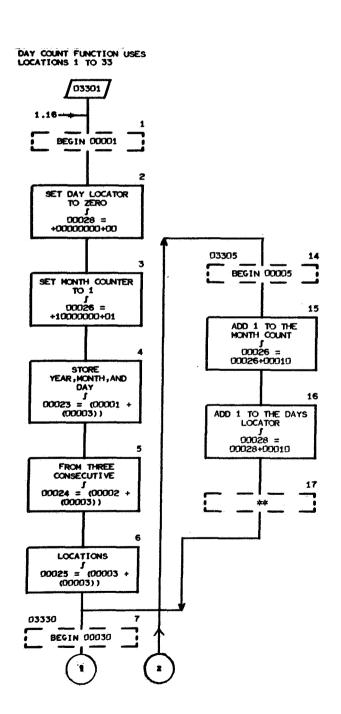
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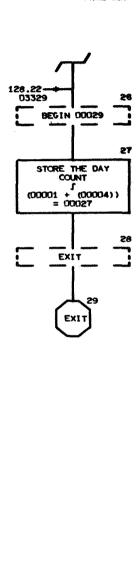
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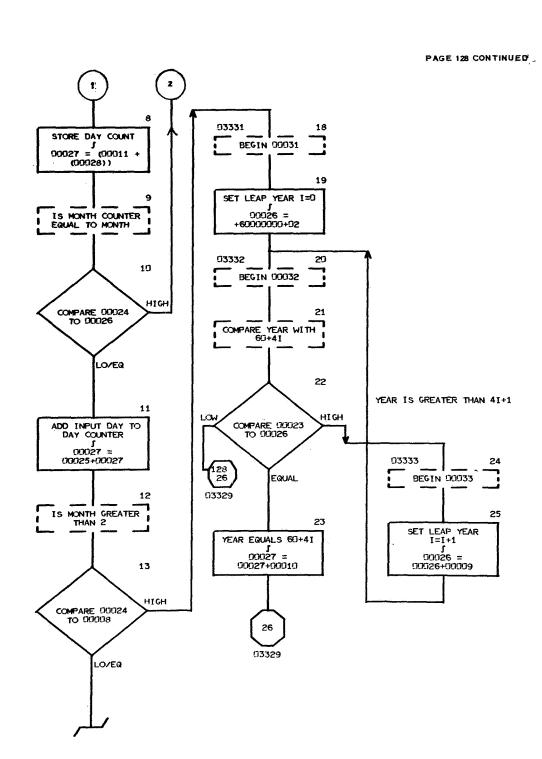
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V00012+100000000+02

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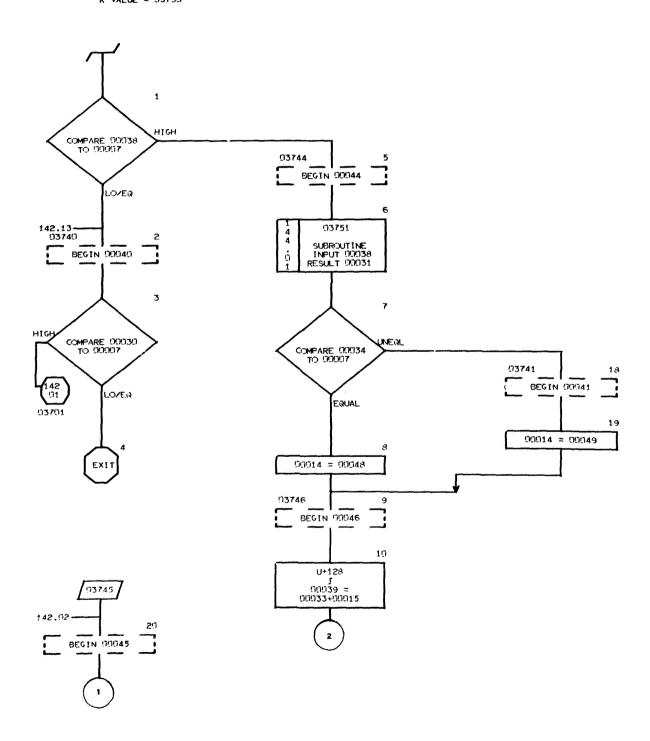




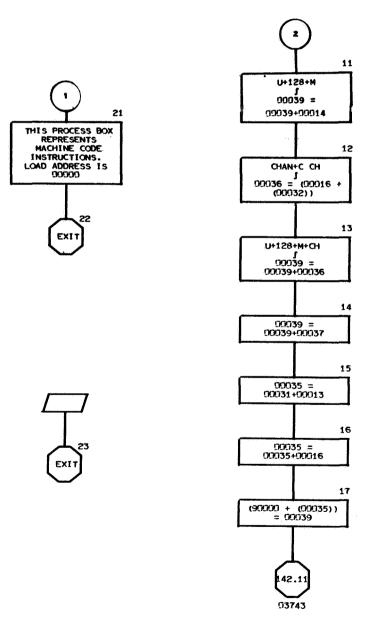
<del>-</del>	
V00008+20000000+01	ENTER WITH (Z)=YEAR, (Z+1)=HONTH, (
V00009+40000000+01	DAY. EXIT WITH (X)=NUMBER OF DAYS
V00010+10000000+01	JAN-1 OF THE GIVEN YEAR THROUGH T
V00011+00000000+00	GIVEN DATE.
V00012+31000000+02	NO. OF DAYS UP TO FEB. 1
V00013+59000000+02	NO. OF DAYS UP TO MAR. 1
V00014+900000000+02	NO. OF DAYS UP TO APR. 1
V00015+12000000+03	NO. OF DAYS UP TO MAY 1
V00016+15100000+03	NO. OF DAYS UP TO JUNE 1
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V00018+21200000+03	NO. OF DAYS UP AUG. 1
V00019+24300000+03	NO. OF DAYS UP TO SEPT. 1
V00020+27300000+03	NO. OF DAYS UP TO OCT. 1
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V00022+33400000+03	NO. OF DAYS UP TO DEC. 1

#### CROSS REFERENCE LISTING

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128.14	03305	128.10	
128.18	03331	128.13	
128.20	03332	128.25	
128.24	03333	128.22	
128.26	03329	128.22	128.23



### PAGE 143 CONTINUED



09000000000

Q9000103751

90003703841 24426

V00005+100000000+01

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V00007+000000000+00

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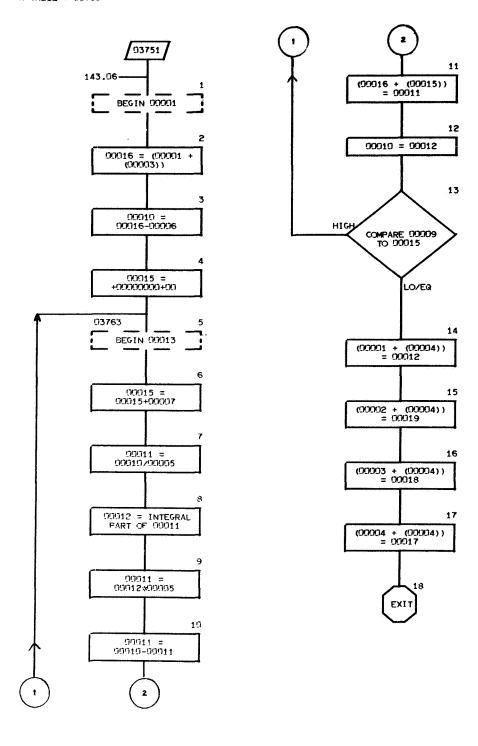
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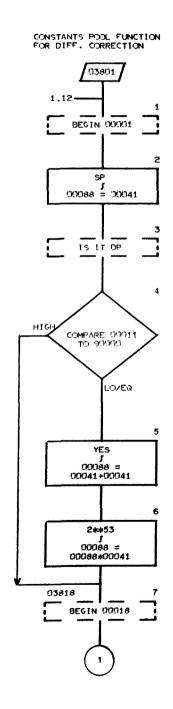
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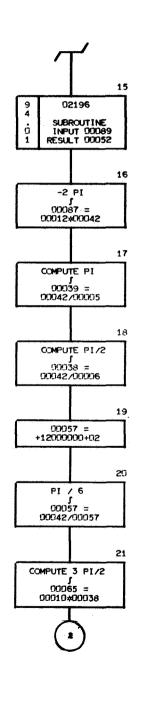
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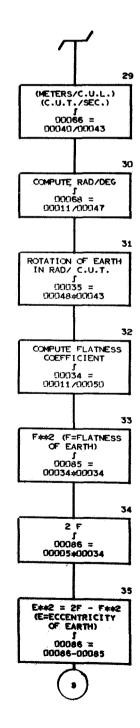
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144.13

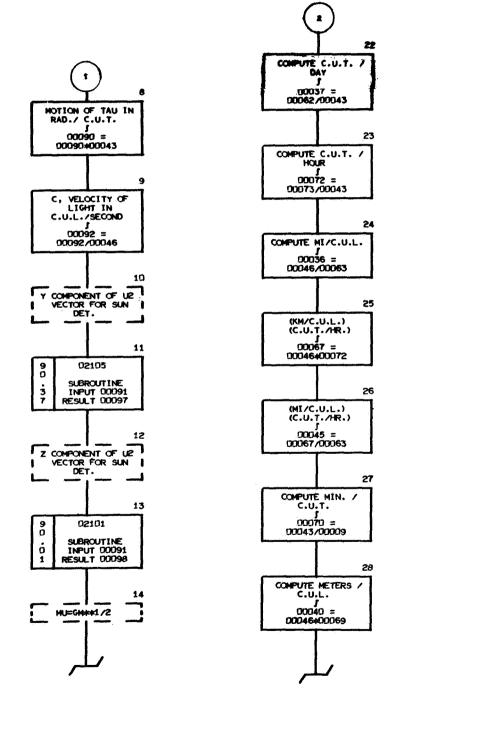
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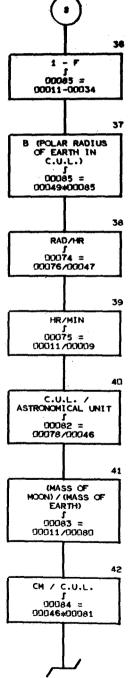




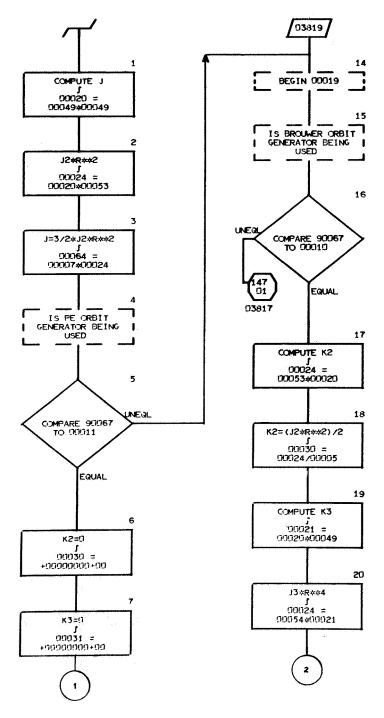


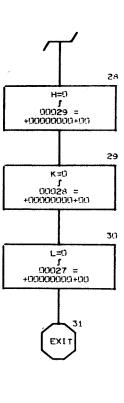
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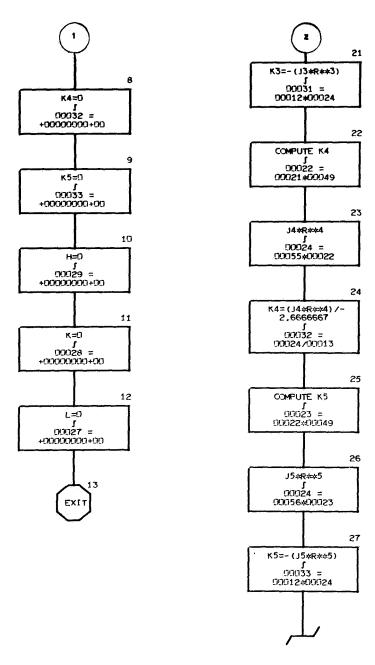


## NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI K VALUE = 93800

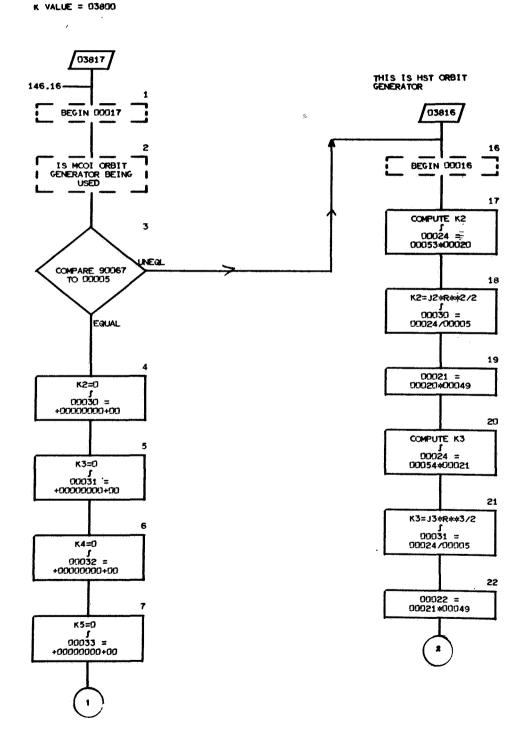




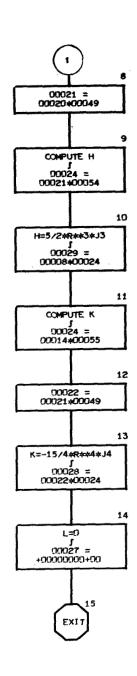
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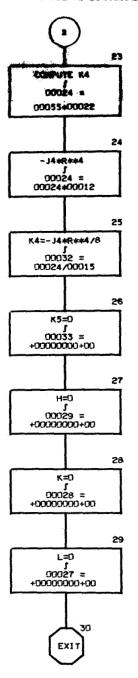


#### . . . .



#### PAGE 147 CONTINUED





#### NASA-GSEC MISSION AND TRAJECTORY ANALYSIS DIVI

# K VALUE = 93800

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<b>9</b> 9000000089	0=SP (88)=2**26, 1=DP (88)=2**53
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Q9000302101	SIN
Q9000402105	cos
V09005+20000000+01	
V00006+400000000+91	
V00007+150000000+01	3/2
vaaaa8+25aaaaa00+01	5/2
V00009-60000009+02	MIN/HR
V000117+300000000+01	
V00011+10000000+01	
V00012-10000000+01	
Vinces+3 -26666667+01	-8/3
VPRO11-375000000+01	-15/4
700015+800000000+01	

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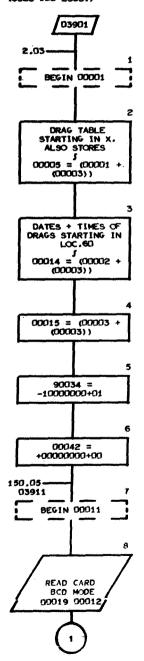
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146.05
146.16
147.03

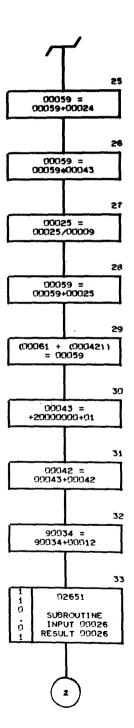
vacaaa+naaaaaaaa+aa <sup>*</sup>	T1, TOLL FOR MAGIOF (R X U) IN SUN D
V00001+00000000+00	TOL. FOR (UNIT R) DOT (U) IN SUN DET
V00016+6 <b>7</b> 108864+08	2**26
viiii017+62831853+01	2 91
V00018 90683200+03	SECONDS / C.U.T. (C.U.T. =
V00019+190000990-01	CRITICAL INCL. FOR BROUWER
V00021+63783880+04	KILOMETERS/C.U.L. (C.U.L. =
V00022+57295780+02	DEGREES/RADIAN
V00023+72921159-04	ROTATION OF EARTH IN RAD/SEC
VUOUZ4+190000000+01	RADIUS OF EARTH IN C.U.L.
V90025+29790000+03	
V00026+10000000+01	c
V00064+190000000+01	MU***2=GM
V00028+10821900-02	JS
V00029-22850000-05	J3
V00039-21230900-05	J4
V00031-23200000-06	J5
V00033+360000000+01	
V99934+31355885=91	SMALL H SUB 0
V10035+26666667+00	(4/3) (BETA)
V77736+62711769-02	(4/3) (H SUB (5)
₩99937 +36499900+95	SEC/DAY
V00038+16093472+01	KM/MI
<b>V</b> 90044+190000000+04	м/км
<b>V00046+1</b> 00000000 F01	19N SUB 17
<b>v00048+36</b> 0000000+04	SEC/HR
V90051+15000900+02	DEG/HR
V00052+45000000-04	SUNLIGHT PRESSURE IN DYNES / CM**2
V00053+14947300+09	KM/ASTRONOMICAL UNIT
V00054+33343200 ×06	(MASS OF SUN) / (MASS OF EARTH)
<b>V00055</b> +81450000+02	(MASS OF EARTH) / (MASS OF MOON)
<b>V00056</b> +100000000+96	CM/KM
V00065+19910638-06	MOTION OF TAU IN RAD. / SECOND
V00066+40915752+00	23 DEG. 26 MIN. 34./95 SEC.
V00067+29979250+06	C, VELOCITY OF LIGHT IN KM. / SECOND
V00068+100000000+01	X COMPONENT OF U1 VECTOR FOR SUN DET
V00069+0000000000+00	Y COMPONENT OF US VECTOR FOR SUN DET
yaadta+aaaaaaaa+aa	Z COMPONENT OF UI VECTOR FOR SUN DET
V99971+99999999+00	X COMPONENT OF UZ VECTOR FOR SUN DET

# CROSS REFERENCE LISTING

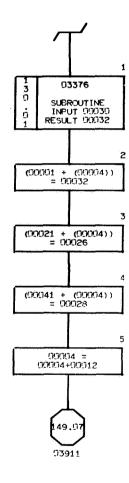
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149.07	03911	150.05
149.17	03906	149.09
149.21	03908	149.18

# LOAD DRAG DATA FUNCTION (USES 100 LOCS.)





#### PAGE 14 CONTINUED 34 02651 10 SUBROUTINE INPUT 00028 RESULT 00028 ò 9 UNE QL 35 COMPARE 00005 TO 00019 99923 = 99923#09913 03906 17 BEGIN 00006 36 EQUAL 00024 = 00024+90023 18 10 00043 = +100000000+03 NEGL 37 COMPARE 90019 TO 00007 00024 = 00024\*00013 03908 21 BEGIN 00008 11 00059 = 00020+00043 EQUAL 38 22 00025 = 00025+00024 12 19 WRITE PRINTER BCD MODE 90019 90012 00059 = 00059+00021 90001 = +00000000+00 39 00016 = 00020 13 23 40 00059 = 00059+00043 90001 = -100000000+01 EXIT 00017 = 00021 14 41 00059 = 00059+00022 99018 = 9002242 15 03351 9 5 (00060 + (00042)) = 00059 SUBROUTINE INPUT 00014 RESULT 00030 16 00059 = 00023\*00043 43 00031 = 00025



 Q9000100006
 ERROR INDICATOR

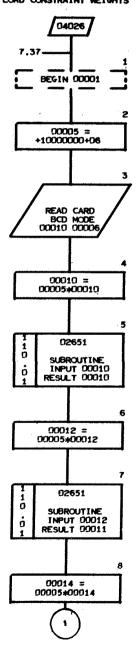
 Q9000202651
 INPUT CONVERTER F.

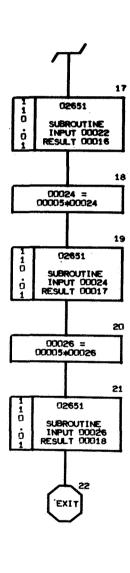
 Q90003033351
 OBS. DATE TO J.D. F.

Q9000403376 JULIAN DAYS-SECONDS TO C.U.T.

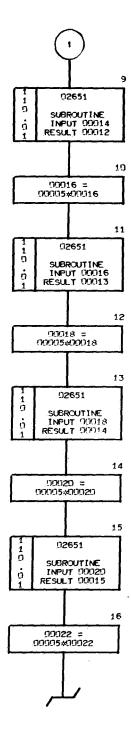
99003400002 NO. OF T(P,Q) "S - 1

### LOAD CONSTRAINT WEIGHTS





#### PAGE 151 CONTINUED



CROSS REFERENCE LISTING

PAGE BOX LABEL REFERENCES

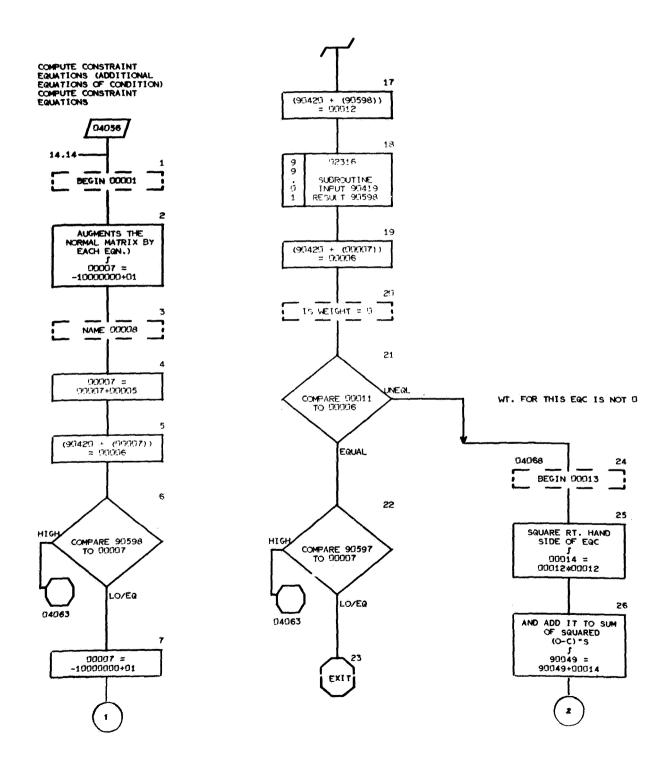
K VALUE = U4025

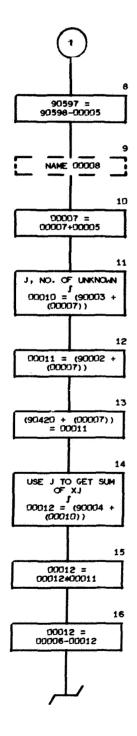
151.01 04026 7.37\*

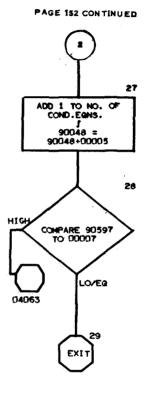
K VALUE = 04025

INPUT CONVERTER 09000102651

V00006+10000000+01







Q9059800598 M Q9059700580 M-1

99042000420

Q9000102316 AUGMENT MATRIX

Q9000204035 WEIGHTS

Q9000300400 FIRST LOC. OF UNKNOWN NO"S

Q9000404200 LOC.PRECEDING FIRST OF SUMS OF XJ

Q9041900419

Q9004800048 N, NO.OF CONDITION EGNS.
Q9004900049 SUM OF SQUARED (O-C) "S

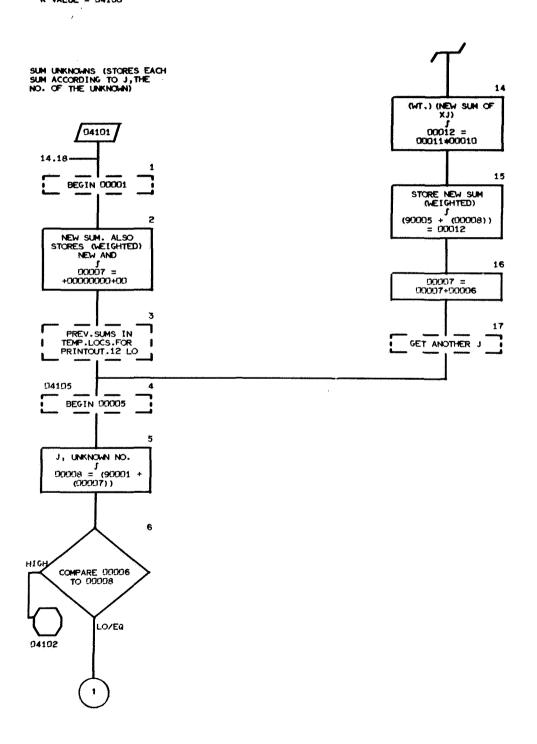
V00005+10000000+01 (COMPUTES AN EQUATION OF CONDITION V00006+000000000+00 EACH UNKNOWN BEING SOLVED FOR, AND

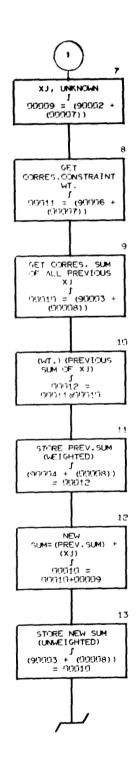
CROSS REFERENCE LISTING

PAGE BOX LABEL REFERENCES

K VALUE = 04055

152.01 04056 14.14\* 152.24 04068 152.21





99909190499 FIRST LOC. OF UNKNOWN NO"S, J
99909290441 FIRST LOC. OF UNKNOWNS, XJ
99909304299 LOC.PRECEDING (UNWEIGHTED) SUMS OF XJ

69900334200 LOC.PRECEDING (UNWEIGHTED) SUMS OF XJ
69000400700 LOC.PRECEDING (WEIGHTED) PREV.SUMS OF
69000500800 LOC.PRECEDING (WEIGHTED) SUMS OF XJ
69000604035 FIRST LOC.OF CONSTRAINT WEIGHTS

V000006+100000000+01 GET SUM OF PREV.XJ, ADDS XJ TO IT +

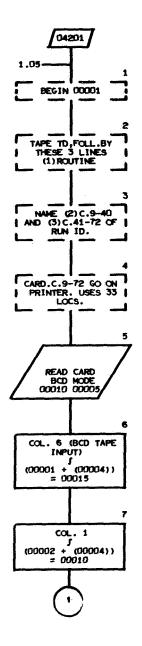
#### CROSS REFERENCE LISTING

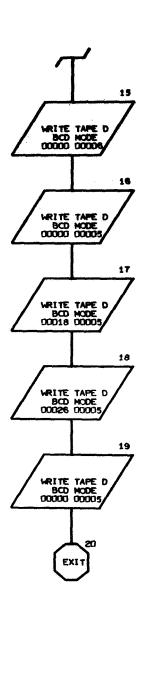
PAGE BOX LABEL REFERENCES

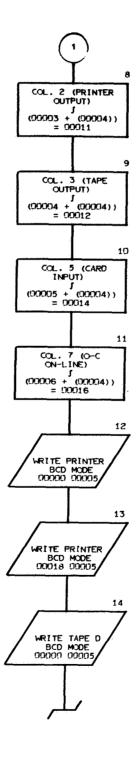
K VALUE = 94190

153.01 04101 14.18\* 153.04 04105 153.17

# RUN IDENTIFICATION LOAD AND PRINT







09000100099 ORBIT GENERATOR IDENTIFICATION

V00005+10000000+01 1,2,3,4,5,7 (ALPHABETIC) IN X...X V00006+60000000+01 LINES OF PRINT CONTROL CHAR.ARE WRI

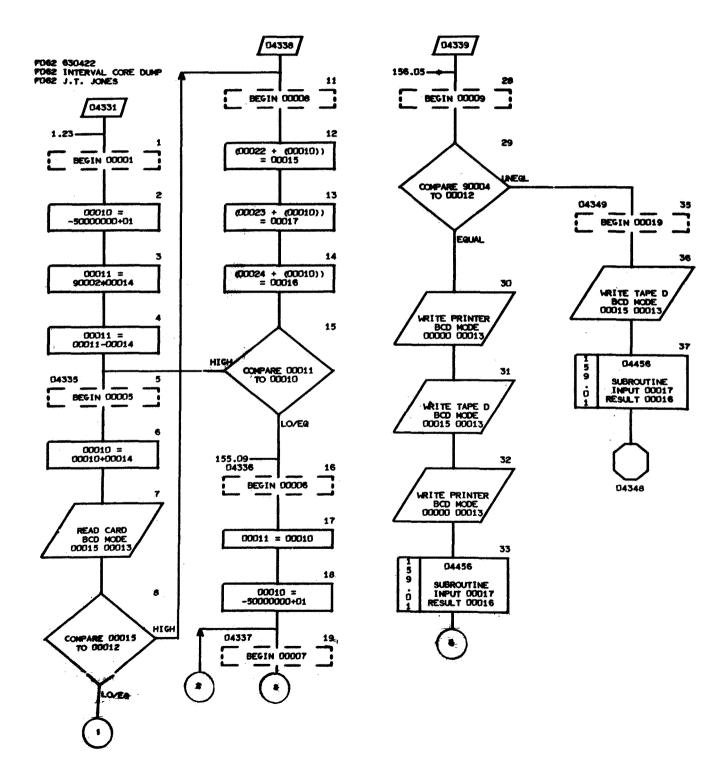
CROSS REFERENCE LISTING

PAGE BOX LABEL REFERENCES

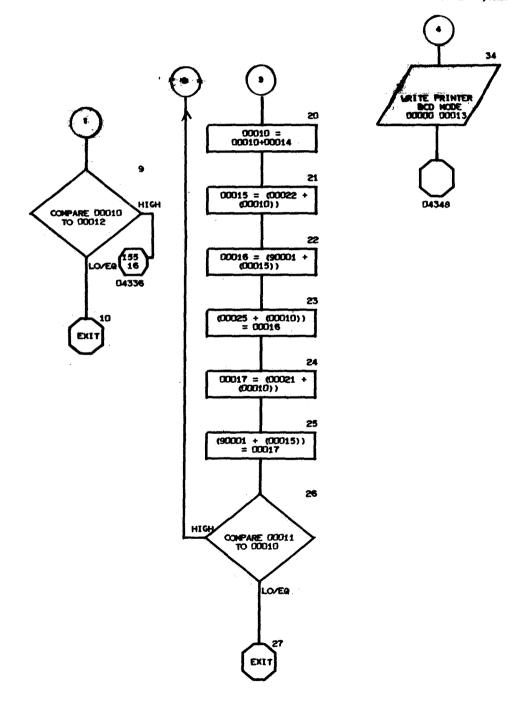
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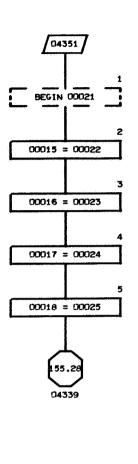
154.01 04201 1.05\*

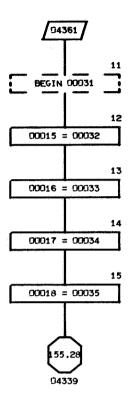
NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI K VALUE = 04330

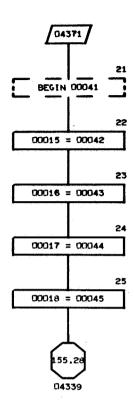


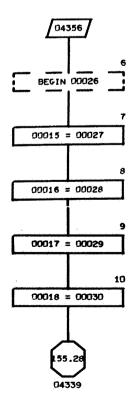


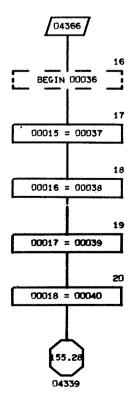


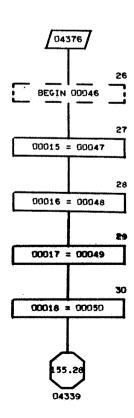


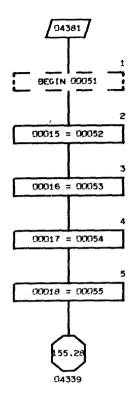


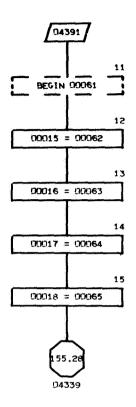


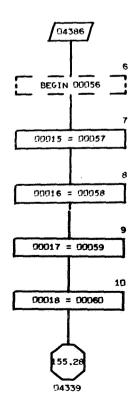


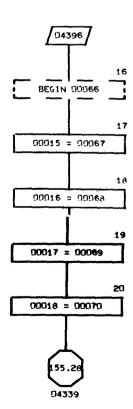






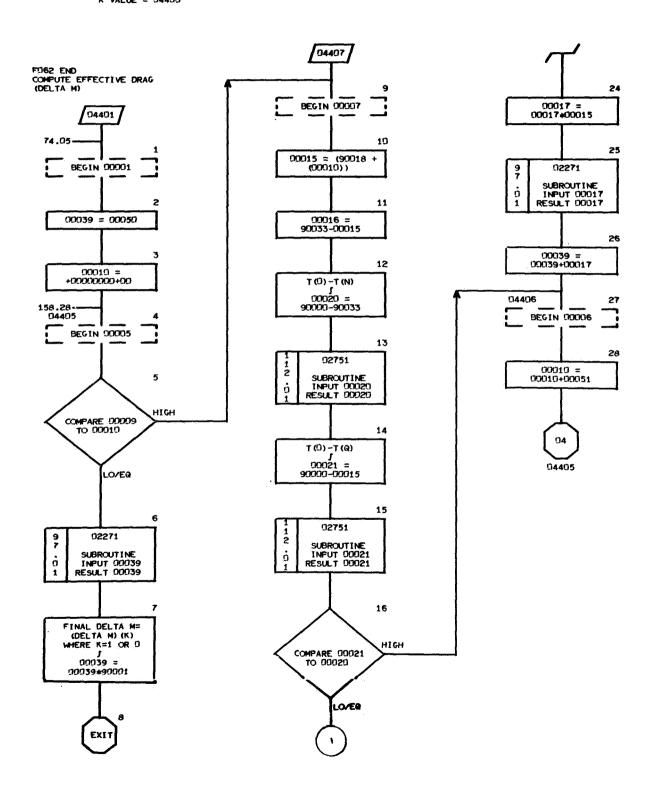




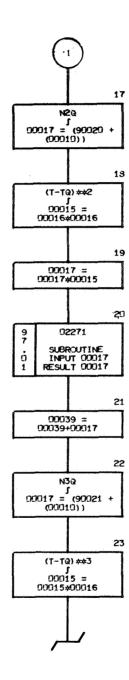


#### PAGE X NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI CROSS-REFERENCE LISTING PAGE BOX REFERENCES LABEL K VALUE = 04330 04331 155,01 1.23\* 155.15 155.05 04335 155.08 155.11 04338 155.16 04336 155.09 94337 155.26 155.19 155.28 04339 156.05 156.10 156.15 156.20 156.25 156.30 157.05 157.10 157.15 157.20 155.35 04349 155.29 156,91 04351 156.96 04356 156.11 94361 156.16 04366 156.21 94371 156.26 94376 157.91 04381 157.06 04386 157.11 04391 157.16 04396 K VALUE = 04330 99990199990 09000200465 N=MAX. OF IP DUMP REQUESTS 99000304456 09000400466 OUTPUT OPTION V90012+999999999+99 V00013+100000000+01

V99014+599099990+01



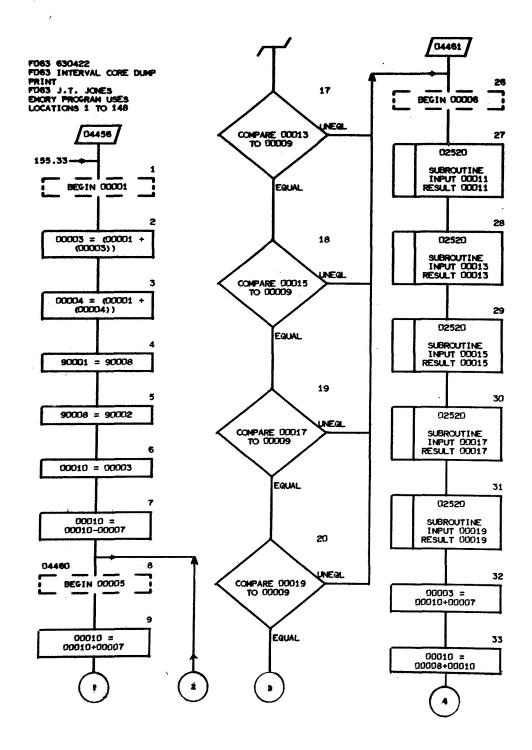
#### PAGE 158 CONTINUED

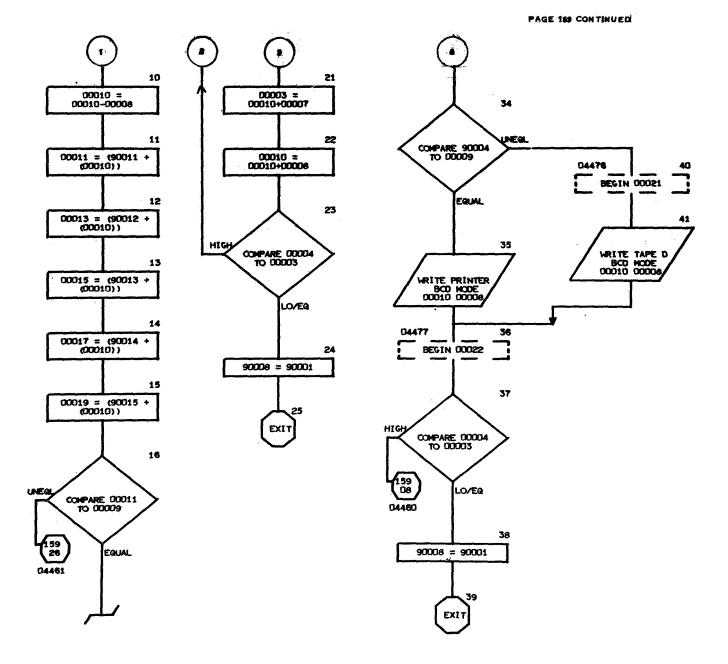


### CROSS REFERENCE LISTING

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K VAL	UE = 04400			
158.01	04401	74.05*		
158.04	D4405	158.28		
158.09	04407	158.05		
158.27	04406	158.16		

W ANTON - DAMEDO	
Q0000901197	NO. OF T(P,Q)"S
Q9000001100	T(0), EPOCH TIME IN C.U.T.
Q9000100462	K, MULTIPLIER FOR DELTA M (K= 1 OR
Q9001801130	T(P,Q) "S TIMES OF DRAGS IN C.U.T.
Q9002001150	N(2,Q) "S
Q9002101170	N(3,Q) "S
Q9003300200	OBS. TIME IN C.U.T.
Q9009002271	PRINCIPAL VALUE
99009102751	ABSOLUTE VALUE
V00050+00000000+00	
V00051+100000000+01	



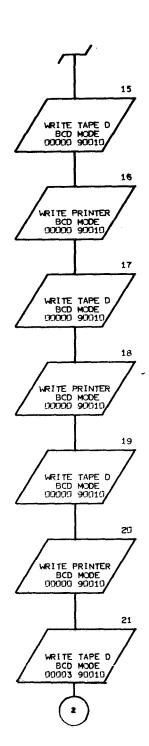


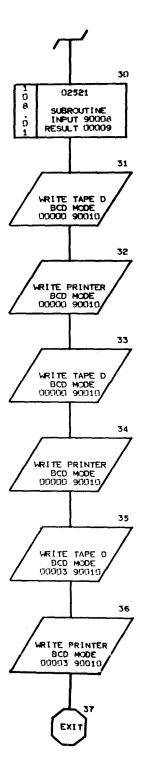
99000102650		KOF	OPS F. 4	130 ,IN	PUT CONVE	RTER
99000202649		K OF	OP5 F. 4	129		
99000302520		OUTP	UT SCALE	F.		
99000400042		OUTP	UT OPTION	4		
Q9000800089		SING	LE PRECIS	BION		
00001100001						
99001200002						
49001300003						
99001400004						
<b>49001500005</b>						
V00007+5000000	0+01					
V00008+1000000	3+01					
V00009+00000000	0+90					
CROSS REFERENCE LISTING						
PAGE BOX	LABEL	RE	FERENCES	3		
K VALU	E = 04455					
159.01	04456	155.33*	155.37*			
159.08	04460	159.23	159.37			
159.26	04461	159.16	159.17	159.18	159.19	159.20
159.36	04477	159.41				
159.40	04476	159.34				

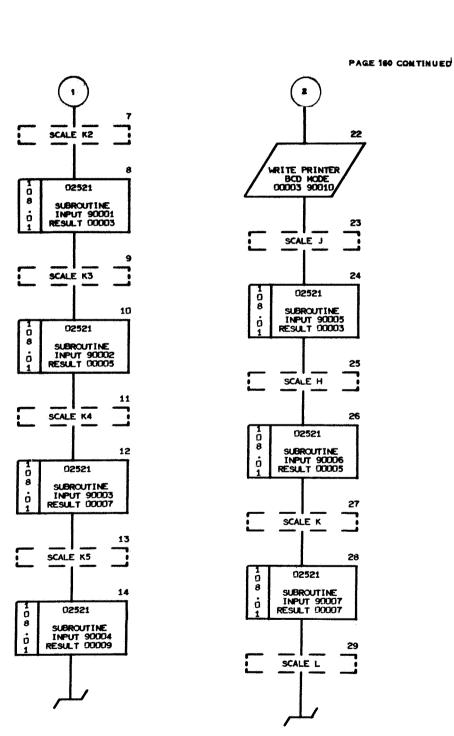
# NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI K VALUE = 04490

BROUMER 1
BROUMER ORBIT GENERATOR
SUBROUTINE FOR
DIFFERENTIAL CORRECTION ROUTINE D4491 BEGIN 00001 SETS LOOP SWITCH FOR BROUMER PROPER 00391 = +00000000+00 00386 = +9999999+08 IS PRINT IND.=0 (FOR HARMONICS PRINTED) 5 UNEG COMPARE 90009 TO 90012 EQUAL 04492 YES. SET IT=1, AND PRINT 90012 = 90010

FU63 END







Q9000103830 K 2 99000203831 К 3 99000303832 K 4 Q9000403833 K 5 09000503864 Q9000603829 H 99000703828 ĸ Q9000803827 L 99009900010 ZERO Q9001000011 ONE

Q9001102521 OUTPUT SCALE
Q9001202290 PRINT IND.
V90012+00000000+00 SET PRINT IND.

#### CROSS REFERENCE LISTING

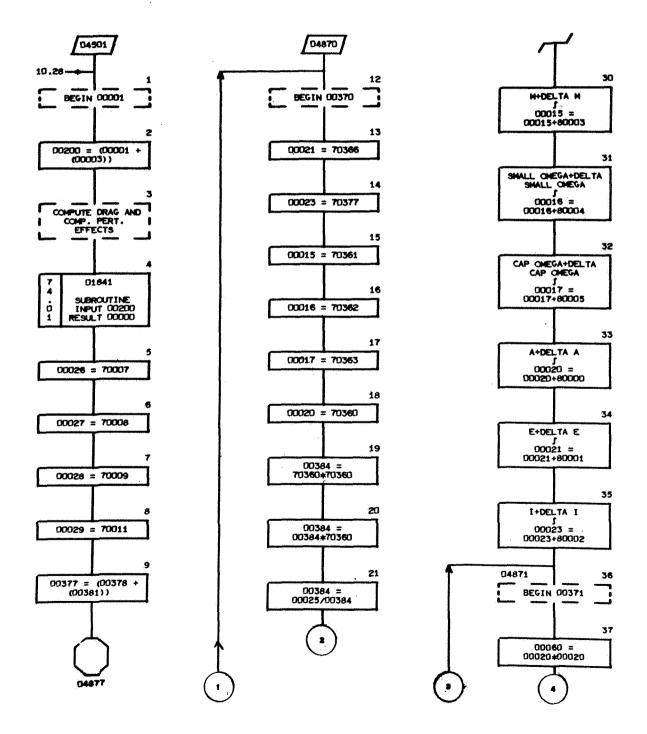
PAGE BOX LABEL REFERENCES

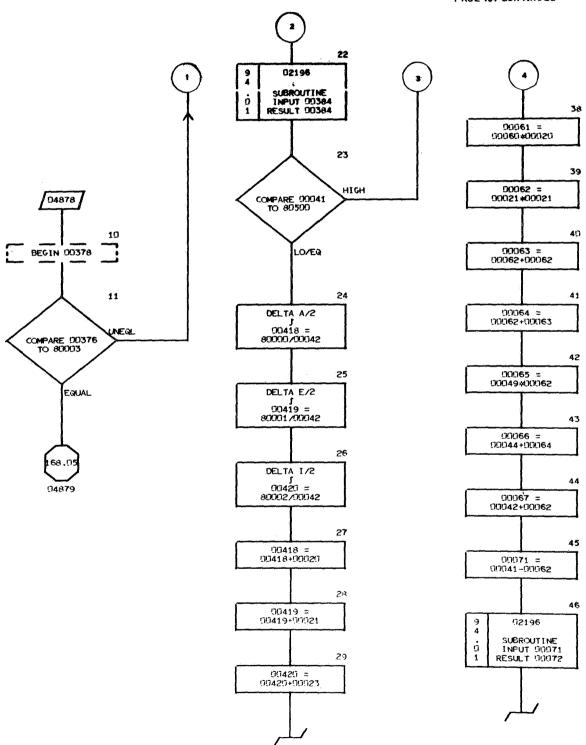
K VALUE = 04490

160.01 04491 3.31\* 10.03\*

MASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI K VALUE = 04500

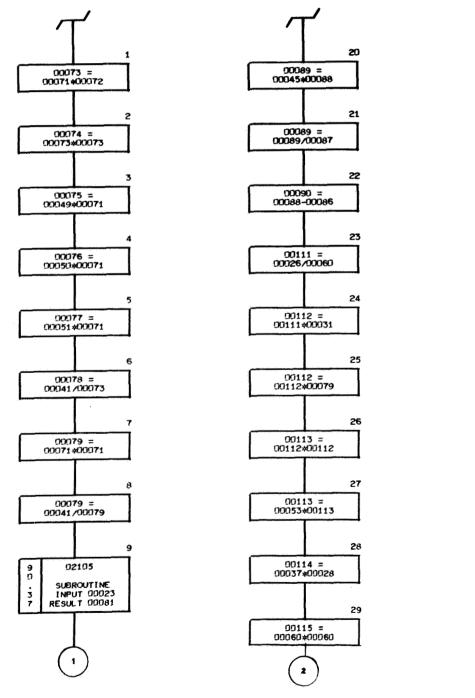
PAGE 161

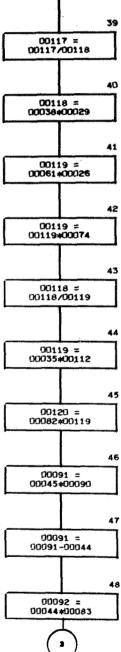


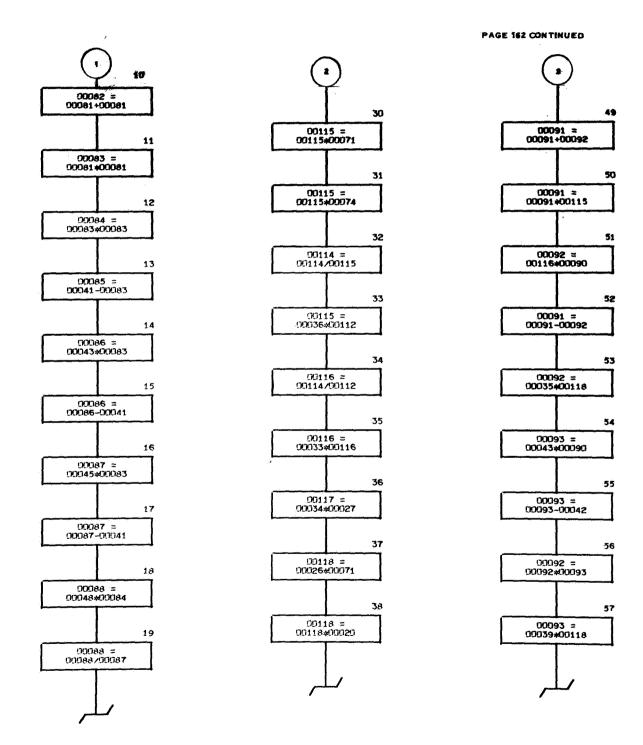


NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI K VALUE = 04500

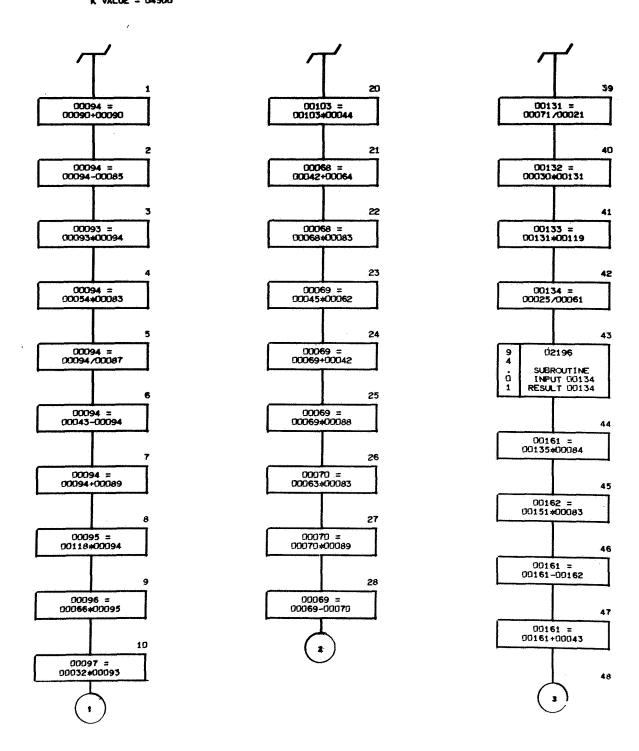
PAGE 162

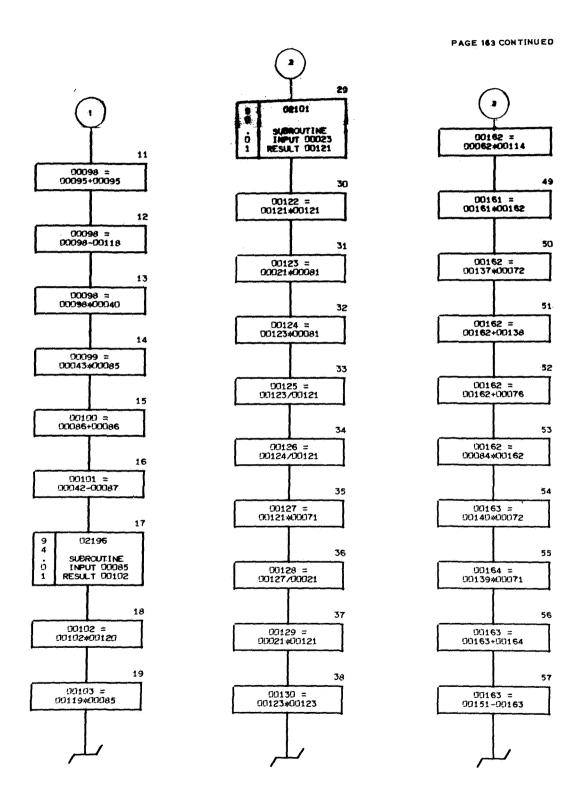


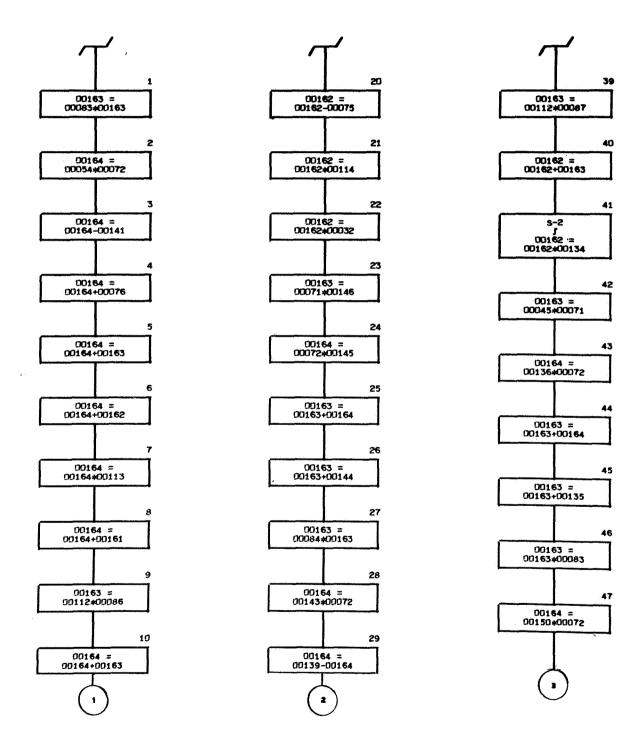


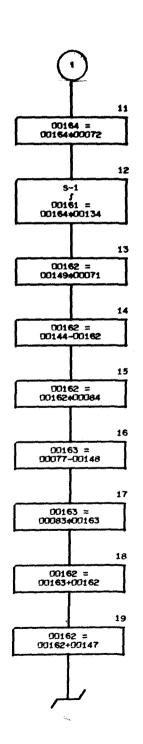


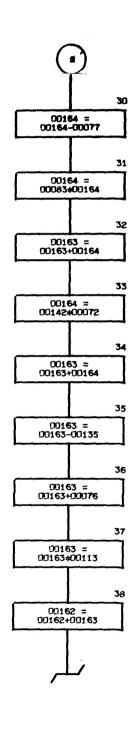
# MASA-GSFC MISSION AND TRAJECTORY AMALYSIS DIVI $\kappa$ value = 04500

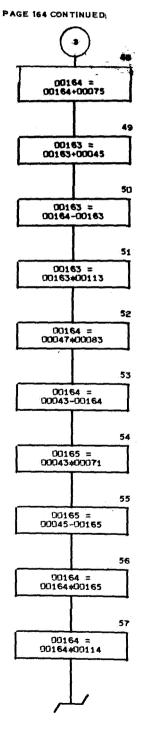


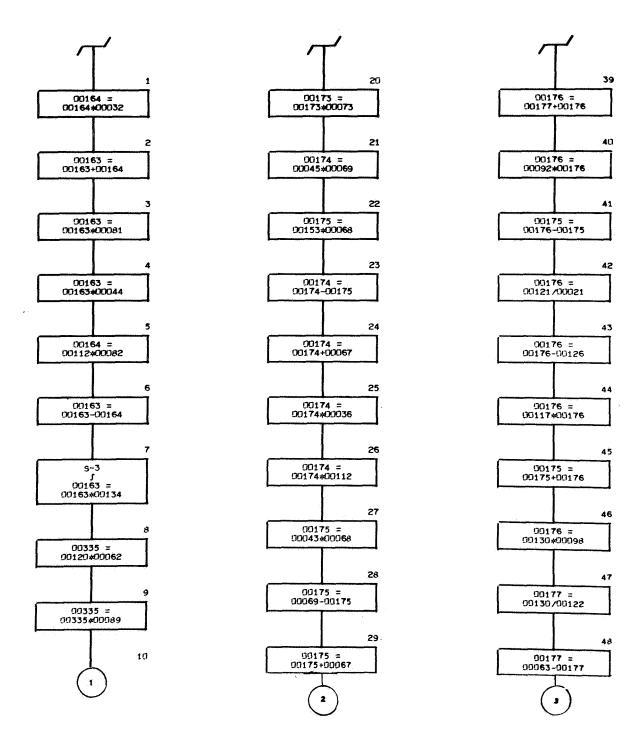


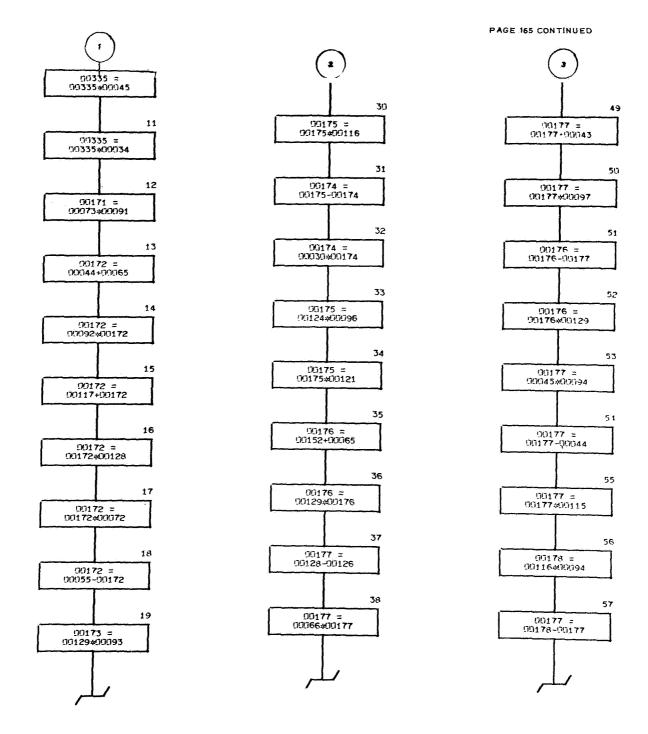




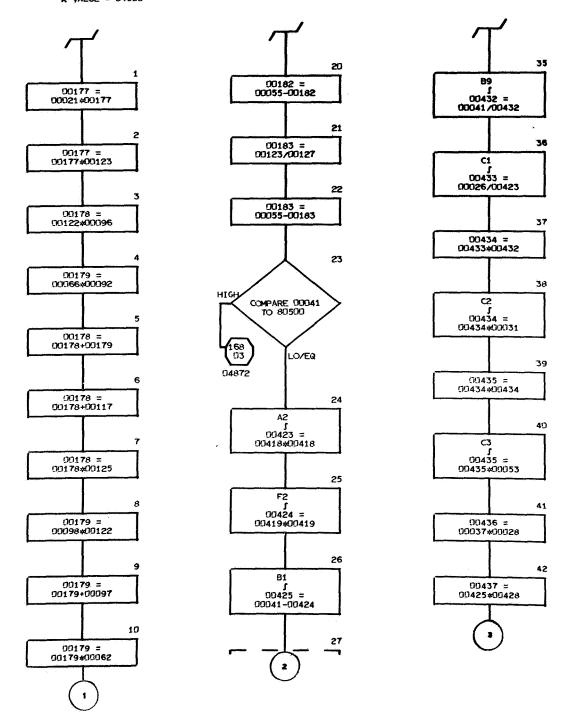


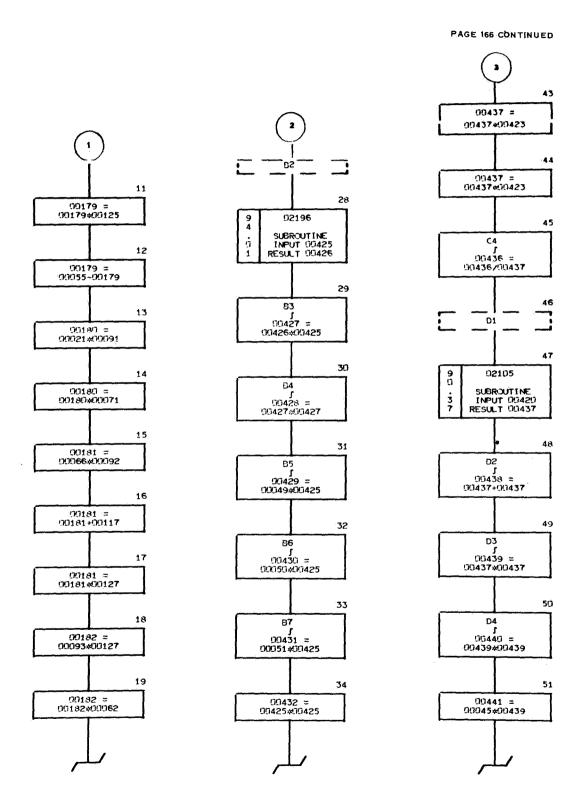






NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI  $\kappa$  yalue = 94590





# CROSS REFERENCE LISTING

LABEL	REFE	RENCES
LUE = 04500		
04501	10.28*	66.26*
04878		
04870	161.11	
04871	161.23	
04872	166.23	
04879	161.11	168.02
04713	168.23	
04853		
04836	168.34	
04837	168.41	
04656	173.01	
04657	173.01	
	04501 04501 04878 04870 04871 04872 04879 04713 04853 04836 04837	UE = 04500  04501

K VALUE = 04500 97000001100 T(D) , EPOCH TIME IN C.U.T. Q7000104947 ANGLE-GUAD: DET. 97000202196 SQUARE ROOT 97000302101 SINE 97000402105 COSINE 97000502216 KEPI FR 97000602166 ARC COSINE 97000703830 K 2 07/10/18/13/831 K 3 07000903832 K 4 97001103833 K 5 97001002271 PRINCIPAL VALUE A, SEMI-MAJOR AXIS AT THES 97036001101 M, HEAN ANOMALY AT T(0) 97036191113 Q7036201115 ARG. OF PERIGEE AT T(0) RT. ASC. OF NODE AT T(D) Q7036301117 97036601102 ECCENTRICITY AT T(0) 97037701116 I, INCLINATION AT T(D) 97040003844 K SUB C (CRITICAL INCLINATION) 06000000276 DELTA A (COMP. PERT.) 98000100277 DELTA E (COMP. PERT.) 98000200278 DELTA I (COMP. PERT.) 98000300279 DELTA M (COMP. PERT.) Q6000400280 DELTA ARG. OF PERIGEE (COMP.PERT.) 98000500281 DELTA RT. AS. OF NODE (COMP.PERT.) 080500000085 COMPLEMENTARY PERTURBATIONS IND. 08050104439 DELTA M (DRAG) A, SEMI-MAJOR AXIS AT REQUEST TIME 09000100218 09000200219 E, ECCENTRICITY AT REQUEST TIME 09000300220 I, INCLINATION AT REQUEST TIME 09000400213 M, MEAN ANOMALY AT REQUEST TIME 09000500215 ARG. OF PERIGEE AT REQUEST TIME 09000600217 RT. AS. OF NODE AT REQUEST TIME Q9000701841 COMPUTE DRAG AND COMP. PERT. EFFECTS 00001803849 METERS/C.U.L. 00001903866 (METERS/C.U.L.) (C.U.T./SEC.) Q0002403843 SECONDS/C.U.T. 00002503889 MU\*\*2=GM 00005203868 RAD/DEG 90020503842 2 PI SECONDS/C.U.T. 00020603843 V00030+500000000+00 V00031+150000000+01 V00032+333333333+00 V00033+66666667+00 V00034+250000000+00 VDDD35+16666667+DD V00036+833333333-01

V00037+93750000+00 V00038+46875000+00 Y00152+26000000+02

V00153+11000000+02

V00039+19444444+00

V00040+12962963+00

V00041+100000000+01

V00042+20000000+01

V00043+300000000+01

V90044+499999999+91

V00045+500000000+01

V00046+600000000+01

V00047+700000000+01

V00048+89009000+01

V00049+900000000+01

V00050+25000000+02 V00051+12600000+03

V00053+41666667-01

V00054+16000000+02

y00055+000000000+00

V00155+000000000+00

V00339+100000000+00

V00135+350000000+02

V00136+360000000+02

V00137+14400000+03

V00138+19599999+03

V00139+900000000+02

V00140+960000000+02

V00141+15000000+02 V00142+24000000+02

.

V00143+19200000+03

V00144+38500000+03 V00145+36000000+03

V00146+450000000+02

V90147+219090000+02

V00148+270000000+03

V00149+18900000+03

1001-97190000000000

V00150+12000000+02 V00151+30000000+02

### CROSS REFERENCE LISTING

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	LABEL		EFERENCI	<b>=5</b>			
	= 09000	22 264	97 444	27 204	92 174		
£ 11 5 5	09001	23.004	23,144	£3.6U#	67.1.1		
176.10		478 44					
176.12							
•	09070		170 03				
		176,15	179.05				
177.01 177.03		176.11	470 30	4 am ma	100.20	100 22	100 NS
177.42						100.22	106.03
*.	09005						
		178.23		211204	2.1.50		
	09071	176.13					
179.09			110,110				
179.11			179.10				
	09082						
	09053	1.0.22	100.00				
180.07	09063	180.06	181.19				
180.09		100,00	2021.23				
180.21		179.15	179.24				
	09078						
181.01	09079						
181.20	09080						
	.UE = 09001	<b>3</b>					
98002503834		F	FLATTEN	ING COEFF	ICIENT		
98002603837			SUB 2 =	C.U.T./0	AY		
96002803858		3	3.6				
Q8003003871		1	+N SUB O				
98003103859		\$	MALL H SL	<i>B</i> 0			
98003203861		•	I SUB O				
Q8003303860		ŧ	ETA .				
Q9000000204		SA	ALL R VE	CTOR, X CO	MPONENT		
99000100206		SA	ALL R VE	CTOR, Z CO	MPONENT		
99000202011		ÝE	ECTOR MAG	NITUDE FU	NCTION		
Q9000402846		A	SUB S FOR	R L,M C1 :	SUB 5 FOR	R. R DO	
Q9000502847		8	SUB S FOR	R L.M CQ/	F2 FOR R,	R DOT	
Q9000602848			ORK AREA	FOR T SU	B 0 NO. 1	OF 6 LO	
Q9000702849		T	SUB :1				
Q9000802850		T	SUB 2				
Q9000902851		T	SUB 3				
Q9001002852			SUB 4				
99001102853		,	ORK AREA	FOR T SU	B 5		
Q9001202854			ORK AREA	FOR N AU	B D NO. 1	OF # LO	
99001302855		N	SUB 1				
Q9001602858		N	SUB 4				
<b>99</b> 001702859		١	ORK AREA	FOR N SU	8 5		
99001802860			ORK AREA	FOR H SU	B 0 NO. 1	OF 6 LO	
<b>99001902861</b>		н	SUB 1				
Q9002202864		н	SUB 4				
99002302865		4	ORK AREA	FOR H SU	8 5		

### K VALUE = 04946

Q9000102156	ARC SIN
Q9000202166	ARC COS
90000503838	PI/2
90000603839	PI
Q0099793865	3 PI/2
Q0001003842	2 PI
V00008+600000000+90	.6
V90009+900000000+90	ZERO

### K VALUE = 04946

174.0	1	04947	179.96*	171.51*
174.0	6	D4961	174.15	
174.1	4	04962	174.04	
174.1	6	D4963	174.08	
174.2	6	04967	174.20	
174.3	1	04968	174.22	
174.3	4	D4969	174.27	
175.0	5	D4965	175.01	
175,0	9	04964	174.12	
175.1	4	04966	175.10	

FOR THE ANGLE-QUADRANT DETERMINATION SUBROUTINE K=04946 SEE PAGE 202 OF THE PROGRAM LISTING

09002402099 UNCORRECTED L. M. RHO OR RHO DOT

99002500200 TIME IN C.U.T.

99002602751 ENTRY TO ABSOLUTE VALUE SUBROUTINE

99002700256 COMPUTED OBSERVATIONS

 Q9002800207
 R, BAR DOT

 Q9002900375
 R, BAR STAR

 Q9003000253
 TYPE CODE

 Q9003202051
 DOT\_PRODUCT\_ENTRY

 Q9003302001
 VECTOR\_MOVE\_ENTRY

Q9003402101 SIN ENTRY Q9003502105 COS ENTRY

Q9003602085 VECTOR Q 1 NO. 1 OF 3 LOCATIONS

99003901201 LSP ENTRY

09004001851 ENTRY TO EXPONENTIAL FUNCTION

99004101202 LSP EXIT

 Q9004202196
 ENTRY TO SQ. ROOT

 Q9004303849
 EARTH RADIUS IN CUL

V00010+00000000+00 ZERO

 V00011+10000000+01
 ONE, CODE FOR RANGE

 V00047+20000000+01
 TWO, CODE FOR L

 V00048+300000000+01
 THREE, CODE FOR M

 V00049+500000000+01
 FIVE, CODE FOR EL

V00050+90000000+01 NINE, CODE FOR RANGE RATE

V00051+13000000+02 THIRTEEN, CODE FOR EL DOT

V00058+10000000-12 TOLERANCE FOR ZERO EL

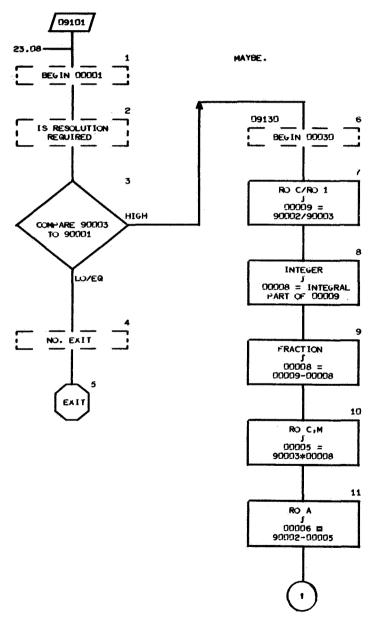
V00076+10000000+10 PSUEDO PARAMETER IF EL APPROACHES Z

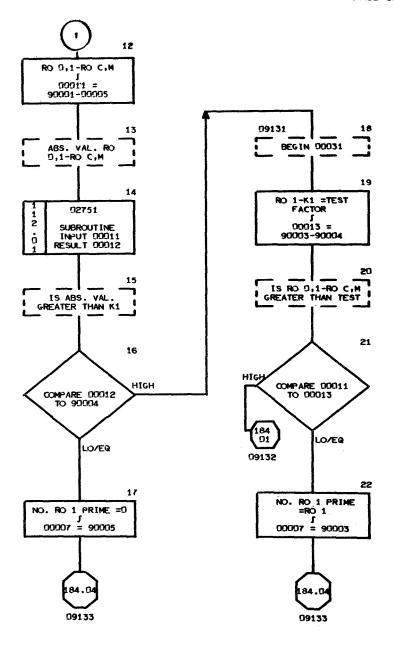
V00052+99922944+00 COS PHI SUB IM V00028+17453293+00 TEN DEGREES, B1

THE INFORMATION ON THIS PAGE IS FOR THE L, M, R, RR CORRECTOR FOR IONOSPHERIC REFRACTION SUBROUTINE K=09000. SEE PAGE 204 OF THE IPROGRAM LISTING.

K VALUE = 09100

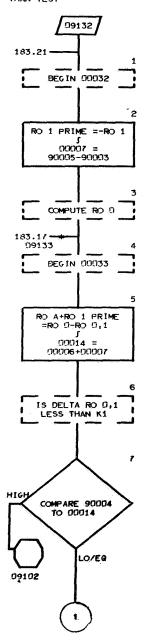
RANGE AMBIGUITY
RESOLUTION FUNCTION
RANGE AMBIGUITY
RESOLUTION F.

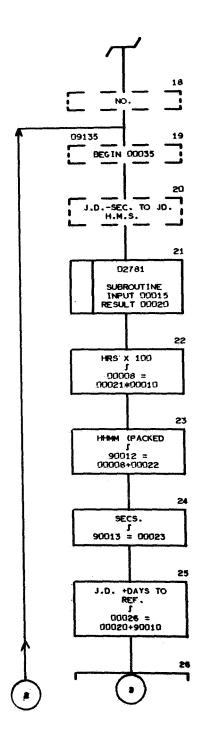


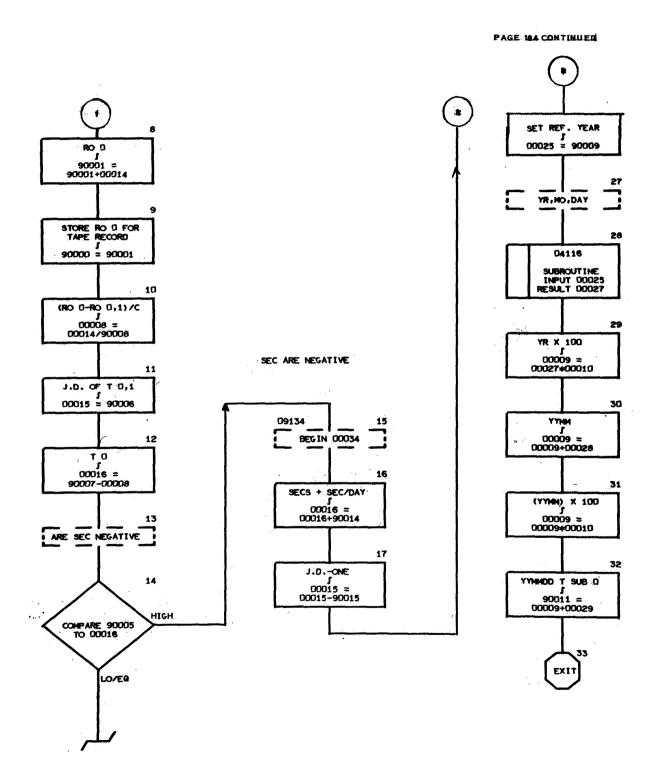


K VALUE = 09100

NO 0,1-RO C,M GREATER THAN TEST







#### PAGE 19 A

#### NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI

K VALUE = 09100

Q9000002835 CORRECTED RO SUB 0

99000100255 RO SUB 0,1 (INPUT). ROSUBD (OUTPUT)

 q9000200256
 RO SUB C

 q9000300396
 RO SUB 1

 q9000400397
 K SUB 1

 q9000500010
 ZERO

a9999699378 J.D. OF T SUB 9,1 a9999799379 SEC. OF T SUB 9,1

Q9000803892 C,VEL. OF LIGHT IN CUL/SEC

 Q9000900296
 YEAR OF REFERENCE

 Q9001000297
 DAYS JAN 1- REF DAY

 Q9001102825
 YYMMDD T SUB 0

 Q9001202826
 HH MM T SUB 0

 Q9001302827
 SECONDS T SUB 0

 Q99001403862
 SEC/DAY

 Q9001503811
 ONE

 Q9100102751
 ABSOLUTE VALUE F.

 Q9100202781
 JD.~ SEC TO J.D. HMS

 Q9100304116
 DATE FUNCTION

 V00017+000000000+00
 ROUNDING FACTOR

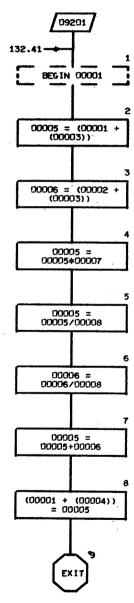
V99010+10090000+93 199

# NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI CROSS-REFERENCE LISTING

PAGE BOX	PAGE BOX LABEL		FERENCES
183.18	09131	183,16	
184.01	09132	183.21	
184.04	09133	183.17	183.22
184.15	09134	184.14	
184.19	09135	184.17	
K VAL	UE = 09100		
183.71	99101	23.03*	
183,06	99139	183,03	

K VALUE = 09200

JULIAN DAYS- SECONDS TO



### CROSS REFERENCE LISTING

PAGE BOX LABEL REFERENCES.

K VALUE = 09200

185,01 09201 132,41\* 133,38\*

K VALUE = 99200

 Q0000703862
 SECONDS/DAY

 Q0000803843
 SECONDS/C.U.T.

NASA-GSEC MISSION AND TRAJECTORY ANALYSIS DIVI

THE FOLLOWING 6 PAGES CONTAIN A LISTING OF ALL OF THE BEGIN COMMANDS OF THE D.C. PROGRAM. THE LOCATION INDICATED BY A BEGIN COMMAND IS THE EQUIVALENT OF A FORTRAN STATEMENT NUMBER OR A MAP/FAP LABEL. THEY ARE USEFUL FOR REFERENCING PURPOSES IN THE EVENT OF THE NEED FOR INTERMEDIATE OUTPUT NOT NORMALLY PRINTED BY THE PROGRAM (SEE PAGE 180 OF THE PROGRAM LISTING).

LABEL	PAGE BOX	LABEL	PAGE BOX	LABEL	PAGE BOX
00008	16.16	01218	24,07	01551	52.35
00020	1.01	01219	24.12	01552	53.01
00021	3.98	01220	24,19	01553	53.18
00022	5.12	01221	25.01	01554	53.35
00023	8.11	01222	25.08	01555	54.01
00024	8.17	01223	25.13	01556	54.18
00027	10.15	91224	25.17	01557	54.35
00028	10.10	01225	25.22	01558	55.01
00029	10.06	01226	26,01	01559	55.18
00030	13.24	01227	26,07	01601	57.01
00033	16.04	91228	26.13	01611	58.01
00036	7.15	.01229	26.18	01612	58.05
00037	7.04	01230	27,08	01613	58.07
00061	16.08	01301	28,01	01614	58.09
00062	4.22	01356	30.01	01615	58.12
00063	16.12	91365	31.05	01616	58.14
00064	16.14	01381	32.01	01617	58.17
00073	13.12	01389	32.05	01618	58.19
00074	12.09	01390	32.27	01619	59.01
00091	9.16	91391	32.09	01620	59.08
00092	9.24	.01393	33.01	01621	59.11
00093	8.04	91394	32.17	01622	59.14
00094	5.1/	91398	32.24	01623	59.19
00095	3.16	91491	34.01	01624	59.23
00101	17.91	01479	34.98	01625	60.01
90119	17.15	91494	55.35	01626	60.04
00111	18.91	<b>91495</b>	56.01	01627	60.10
00112	18.35	01501	34.10	01628	60.20
00128	18.14	91592	35.01	01629	61,07
00129	19.01	01503	36.01	01630	61.22
00130	18.25	91504	37.17	01631	62.25
00131	19.03	01505	37.23	01660	64.01
00132	17.10	91596	37,29	01668	57.23
00133	17.20	01507	38.01	01669	51.26

# NASA-GSEC MISSION AND TRAJECTORY ANALYSIS DIVI

LABEL	PAGE BOX	LABEL	PAGE BOX	LABEL	PAGE BOX
00134	17.13	01508	39.01	01701	66,01
00151	20.01	01509	40.01	01710	80.88
00158	21.03	01519	40.27	01711	68.11
00159	21.96	91511	4101	01712	68.13
00195	12.25	01519	41.11	01713	70.01
00196	13,16	01520	42.12	01714	69.05
00263	8.26	01521	42.21	01715	69.17
00264	7,26	01522	43.91	01716	70.13
00393	11.23	01523	43.16	01717	71.33
00399	11.36	01524	43.31	01718	73.01
00472	15.01	01525	44.01	01728	70.05
00473	15.98	01526	44.16	01 729	69.01
00480	9.22	01527	44.31	01730	67.21
00481	9.14	01528	45.01	91731	66.03
00482	19.17	01529	45.16	01732	67.15
00483	10.22	01530	45.31	01776	66,48
00491	10.26	91531	46.01	-01777	67.19
00492	10,32	01532	46.16	91778	68.05
00493	22.23	01533	46.31	01801	6.18
00494	22.26	<b>91534</b>	47.01	01802	6.01
00570	1.28	D1535	47.16	01804	6.06
00571	7.01	01536	47.31	01805	6.15
005/3	4.01	01537	48.01	01806	14.10
00574	4.93	01538	48.16	01807	14.01
00575	3.01	Ω15 <b>39</b>	48.31	01810	6.22
00576	2.06	01540	49.01	01812	4.13
00577	2.01	01541	49.18	91813	5.01
00578	3.25	01542	49.35	01814	2.12
01201	22.01	01543	50.01	91815	2.21
01211	23.01	01544	50.18	01816	8.22
01212	23.10	91545	50.35	01817	9.06
01213	23.16	91546	51.01	01820	2.26
01214	23.22	.01547	51.18	01831	12.13
01215	23.26	91548	51.35	01832	12.16
01216	24.01	01549	52.01	01834	13.32
01217	24.05	01550	52.18	01840	16.19

### NASA-GSEC MISSION AND TRAJECTORY ANALYSIS DIVI

		TABLE OF	BEGIN COMMAN DS		
LABEL	PAGE BOX	LABEL	PAGE BOX	LABEL	PAGE BOX
01841	74,91	92361	102.01	03026	125.35
01851	75,01	92379	102.11	03051	127.01
01875	75.19	92372	102.13	03055	127.04
01876	75.37	92374	102.15	03056	127.12
01877	76,04	923//	102.17	03057	127.16
01878	76.09	92389	102.32	03058	127.31
01883	75.08	02391	102.34	03301	128.01
01884	75.12	02396	103.07	03305	128.14
01901	77.01	02397	103.10	03329	128.26
01927	77.15	02398	102.37	03330	128.07
01928	77,06	02399	104.01	03331	128.18
01941	78.01	02400	104.03	03332	128.20
01956	79.01	02402	104.05	93333	128.24
01971	.80,01	02407	104.10	03351	129.01
01986	81.01	02413	104.20	93365	129.11
01997	81.16	02416	104.12	93366	129.14
01999	81.05	02418	105.01	03369	129.17
02001	82.01	02476	107.01	93370	129.09
02011	83.01	02484	107.16	93371	129.20
02021	84.01	02486	107.05	03376	130.01
02029	84.12	02494	107.07	03380	130.16
02031	85.01	02495	107.21	03388	130.11
02041	86.01	02521	108.01	<b>93389</b>	130.20
02051	87.01	92549	108.33	03390	130.08
02061	88.01	02541	108.42	93491	131.01
02076	89.01	92542	109.01	03405	131.03
02093	10.38	92543	109.14	03406	132.01
02094	11.05	02596	109,19	03407	133,01
02095	11.10	92695	109.23	03416	131.12
02096	11.17	92696	109.35	93417	131.21
02101	90,01	02609	109.33	03418	132.19
02105	90.37	02649	108.29	93488	133.19
02121	90.07	92651	110.91	03491	132.30
02124	90.15	02664	110.25	03492	132.39

### NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI

		TABLE OF BEGIN	COMMANDS		
LABEL	PAGE BOX	LABEL	PAGE BOX	LABEL	PAGE BOX
02126	90.34	02721	111.01	03494	133.27
02156	91.01	02725	111.16	03495	133.36
02163	91,09	02729	111.12	03496	134.01
02164	91,11	02751	112.01	03501	135.01
02166	92.01	02756	112.04	03506	135.08
02173	92.09	02757	112.06	03510	135.11
02174	92.11	02761	113.01	03513	135.18
02176	93.01	02767	113.04	D3514	135.25
02196	94.01	92768	113.10	03521	136.01
02201	94.06	02801	114.01	03530	136.04
02207	94,12	02806	114.08	03532	136.10
02216	95.01	92807	115.01	03535	136.17
02228	95.13	02808	115.23	03539	136.13
02241	95.16	02810	114.17	03551	137.01
02242	95.32	02811	114.96	03573	138.01
02243	95.09	02814	115.17	03574	138.08
02244	95.19	02876	115.20	03575	138.25
02246	96.01	02936	117.01	03576	139.15
02265	96.14	D294D	117.08	93577	140.01
02266	96.22	U2941	117.10	93578	140.16
02267	96.11	02942	118.26	03579	139.24
02271	97.01	D2943	118.28	03580	139.02
02275	97.11	02944	119.15	03601	141.01
02301	98.01	02946	119.01	03701	142.01
02314	98.08	02947	122.01	93749	143.02
02316	99.01	D2948	125.30	03741	143.18
02329	99.08	D2949	121.09	93742	142.15
02330	99.13	02952	122,41	03743	142.11
02341	100.01	02953	123.28	D3744	143.05
02353	100.06	02964	121,06	D3745	143.20
02354	100.12	02978	118.16	03746	143.09
02355	100.15	02979	123.40	03751	144.01
02356	100.17	03005	125.02	Ø3763	144,05
02357	100.22	03016	121.04	03801	145.01
02358	101.03	93917	117.06	93816	147.16
02359	101.01	03020	120.32	03817	147,01

# MASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI

LABEL	PAGE BOX	LABEL	PAGE BOX
03818	145.07	09054	180.09
03819	146.14	09059	180.21
03901	149.01	09063	180.07
03906	149.17	09067	177.03
03908	149.21	09070	176,14
03911	149.07	09071	179.01
04026	151.01	09073	176,16
04056	152.01	09078	180.23
04068	152.24	09079	181.01
04101	153.01	09080	181,20
04105	153.04	09081	179.11
04201	154.01	09082	179.16
04331	155.01	09101	183,01
04335	155.05	09130	183.06
04336	155.16	09131	183.18
04337	155.19	09132	184.01
04338	155.11	09133	184,04
04339	155.28	09134	184.15
04349	155.35	09135	184.19
04351	156.01	09201	185.01
04356	156.06		
04361	156.11		
04366	156.16		
04371	156,21		
04376	156.26		
04381	157.01		
04386	157.06		
04391	157.11		
04396	157.16		
04401	158.01		
04405	158.04		
04406	158.27		
04407	158.09		
04456	159.01		

### NASA-GSFC MISSION AND TRAJECTORY ANALYSIS DIVI

LABEL	PAGE BO
04400	159.08
04461	159.26
04476	159.40
04477	159.36
04491	160.01
D4501	161.01
04656	173.09
04657	173.17
04713	168.14
04836	168.36
04837	169.40
04853	168.20
04870	161.12
D4871	161.36
04872	168.03
04878	161,10
04879	168,05
D4947	174.01
04961	174.06
D4962	174.14
04963	174.16
D4964	175.09
04965	175.05
D4966	175.14
D4967	174,26
D4968	174.31
04969	174.34
09001	176.01
09005	177,38
09006	178.35
70000	177.12
09012	179.09
09029	176.12
09034	176.10
09045	177.01
09053	180.01

# Reference

B1. IBM 7090/7094 Autoflow System User's and Operator's Manual, Prepared under Contract No. NAS5-10021 by Applied Data Research, Inc., Washington, D. C.

# APPENDIX C

Differential Correction System Program Listing

RUN
SATELLITE
DAI COONT
ORBIT. GENERATOR
FLEMENT
MATRIX
UBSERVALIUN Seo
DRRIT
LOCAL
POSITION
PCSITION
OBSERVATION PARTIALS
AUGMENT
SOLVE
CONVERT
OAD
SEARCH
SCUARE
SIS
ARC SIN
NORWAI
NO. WORDS IN PERT.
K. MULTIPLIER FOR DRAG DELTA
NO. STURED IN LOC. 00000 IF TAPE CHECK FOUND
NO. OF TIMES
MAXIMUM
ETTING
NORMAL
MAX.NO.
SET FOR
õ
7 7 7

600	18000	00082	0000	0000	78000	2000	18000	00088	68000	06000	16000	26000	0000	#A000	20000	2000	80000	00000	00100	8 00020 00101		00103	00104	00100	00100	70100	00100	00100	01100	00111	00113	00114	00115	91100	00117	00110	00120	
LINE DUDSI	NO. OF ELEMENTS	MAX. NO. OF VARIABLES	NC. UT MUKDS/ ELEMENI MANE													RECORD ON TG	SETTING, FOR NU	S. TAPE WRITTEN UN 16	X+ WHERE K I = (RHU I) / X	TITE STAN	1 4		RE INPUT OPTION			CHANGE INITIAL CONSTANTS	CONSTANTS	SET TOL. USED IN (O-C) ANGLE RED.F.= P1/2						WERE THERE TOO MANY REQ. CARDS	INTERVAL CORE DUMP CARDS		1	
0000	V 00012 +19000000+02	00013	00015	00016	00017	V 00018 +10000000+06	V 00058 +2000000+01	000071	98000	}	00149			02091	V 02092 +80000000+01	V 00072 +48000000+02	V 00394 +000000000+00		V 00398	> 0	F 00000 90037 00000	00073	00040 90013	2000		00000	00000 90045	03838	R 00305 03849	92000 00000	F 00297 90075 00295	00200		00101 00010	00000 66006 00000	00000 00028	F 00000 90043 00000 R 01197 00010	! !

00121 00122 60122	00124			00127	00128	100		00132	00133	00134	00135	001	00137	00139	00140			00143	27 100	00145	00147	00148	00149	00150		00152	00153	00154	00155	00120	00134	00128	001	
B 00577		,	8 00576			A 01814	40						8 01815			1	8 00575				B 00021			1	N 00050								B 00578	PAGE 004
I DAN DRAG DATA	1	NO. OF T(P,Q)'S		LEM. INP	YES. REWIND TA	I DAN BECOOD DE ELEMENTS	מאט ארכטאט טד	WAS TAPE CHECK DETECTED	TO NO. OF	HAVE ENDUGH TRIES BEEN MADE	BACKSPACE AND	TRY AGAIN TO READ RECORD	CHANCE COMBITED CONSTANTS	Sold of the second			ON CARI	CHANGE COMPUTED CONSTANTS	NURMAL ELEMENT LUAD	ELEMENI LUAD					TING QUANTITIES FROM	0-1199 AND S	IN LOCS.1000-1099, TO BE SAVED AS	ORIGINAL QUANTITIES		S MCUI UKBII	TO THE THE WHO		Tarrette and the second	LINE 00160
# B 00577 # B 00577 E 01130 00072 00395	90000	01197 00002 00011	00576	01000 86000	P 00000 01800 TAB	01017	۰ ـ		01811 01811	01811 00460	L 00000 01800 TAB		* B 01815	00085 00011	00000 00011 CA	- 1	00575	F 00 000 90069 00 000	01000 06000	r 00008 90079 00010	00021	1	00014	I 00007 +99000000+02		69000 69000	00051 01100	69000	00007 00069 00050	85000 66000	17000	00567 00567 00	00578	K = 00000

	00161 00162 00163	99100	00155	00166	74100	00168	00169	00170	17100	00172	00173	00174	00175	00176	00177	001 78	00179	00180	00181	00182	00183	00184	00185	00186	00187	00198	00189	00100	00191	00192	00193	00194	00195	00196	00197	00198	001 99	00200	
PAGE UDS				,	B 00573		8 00574	1											8 01812			The second secon			:	8 01813									B 00022				PAGE 005
LINE 00161	PRINT INITIAL ELEM., DRAGS, EARTH CONSTANTS STORE ORBIT GEN.ID. IN 100-WORD RECORD	INITIALIZE OBBIT CEN. TO CET C	TABLE CENT OF STA TAB	N NEED NOT BE WELLID	יו ארבט אסן בר מסביו זו בארם סו	3 N													LOAD PERT. TAPE TITLE RECORD			ADD TO NO.	ENOUGH TRIES BEEN MADE	NO. BACKSPACE AND	AGAIN TO			DELTA T					INITIALIZE INTERPOLATION		NEXT SET OF VARIABLES	STORE K, MULT. FOR DRAG DELTA M	302030215151502		1 TNF 00200
00000 ≡ X	F 00000 90041 00000 R 01199 00099 I 00007 +5000000+02	00000	00000 11000 00000	C 00098 00010 00373	[	* 5 005/5 * 01119 01119 03851	00574	ں ا		P 00000 00011 TD	í	T COMPLEME	00 00011 TD	00000 00011	} } }	1 00069 -10000000+01	00000	0000	* 8 01812	1	C 00000 00459 01813 01813	01811 01811 00011	01811 00460	00000 01800	4	1	00000	00293 00601 03837	00601 00604	_		00294	00000 90058	89006	00022	00462	L 00075 00011 CA 0302030203020	ł	00000 × ×

00201	00202	00203	00204	1000	60700	00206	00207	00208	00200	00210	00211	00212	00213	00214	00215	00216	00217	00218	00219	00220	00221	00222	00223	-2000 	002.20	00226	00228	00229	00230	00231	00232	00233	00234	00235	00236	00237	00238	00239	00240	•
																					- 1	N 00050							8 01802			- 1	B 01804				•			
		A CT NI					SSSINNINNNNNN	SSSNNNNNNNNN		And the second s									•								TAPE TO BE WRITTEN ON TG						D FROM TB			ш	S BEEN MADE	1	AD RECORD	
		METERS SIGMA SM SA					20302030203020302	20302030203020302														$\Box$	ONKNOMNS II O				IS A CORR. OBS.	ON	YES				LOAD TITLE RECORD		TAPE CHECK I	ADD TO NO.	E ENGUGH TE	SACKSPALE	TRY AGAIN TO READ	
00011 PA	00011 TD		r.		00011 PA		00011 TD 151502030203	00011 PA 151502030203	0000+07	00075 00050			1	00083 00050	90070 00075	77000 07006	90070 00079	90070 00081	90070 00083	+99000000+02	+000000000+			-		O1800 TAB	ŀ			01800 TGB	01800 TBB	00010		<b>1</b> 88	- 1			01800 TBB		
00000	P 00000	<b> -</b>	0000		00000	<b>!</b>	00075	P 00075	000020	00075	00077		00081	00083	00075	77000		00017	F 00016		I 00069	000020		69000	20000	P 00000	00394	01801	* B 01802	00000		01803	01804		00000		01803	00000	E 01804	

* * * * * * * * * * * * * * * * * * *				8 01805	
<b>8 5 4 m 8 4 8 1 5</b>		2 TGB 0 TBB	WRITE TITLE RECORD ON TG	- 1	00241 00242 00243
ш m m m - u		0 00571 00571	WAS ELEMENT INPUT ON BINARY TAPE YES. USE SIGMA FROM TAPE RECORD	10810 8	002 <del>44</del> 00245 00246
ں بہ ھ	00178 00075	10		8 00571	00247 00248 00249
	00037 00400 00011 00011 00400	CA 0303030303030303	NEW SET OF UNKNOWNS 03030303030303030303030NNNNNNNNNNNNNN	B 00037	00250 00251 00252
00 1			SET CNTR.OF ITERATIONS USING NEW SET=0		00253 00254 00255
* 00 A	j	9 00010	SET WEIGHTS FOR ADDITIONAL CONDITION EQNS.ASSOCIATED WITH UNKNOWNS = 0	N 00050	00256 00257 00258
				N 00050	00259 00260 00261
	00069 00069 00051 00400 00051 00010	9 .00011 0 00069 0 00050			00262 00263 00264
	1 1		LOAD CONSTRAINT WEIGHTS CARD		00265 00266 00267 00268 00269
	00000 00011 T UNKNOWN 00000 00011	1 PA N NO. CONSTRAINT WT. 1 PA 1 TD		:	00271 00271 00272 00273
		19 :	GET UNKNOWN NO.	N 00050	002 74 002 75 002 76
1 ` 1	1 1	0 00093 1 PA 1 TD			002 78 002 78 002 79 002 80
		K = 00000	LINE 00280	PAGE 007	

	00281 00282 00283	00284	00286	00288 00289	00290	002 93	00294	00296	00298	00299	00301	003 02	00304	00305	00306	00308	00309	00311	00312	00314	00315	00316	00317 00318 00319	00320	
PAGE 008	B 00093				B 00023		B 00024		8 01816					8 01817						demand of the second of the se		8 00481	8 00091	i	900 2040
4.INE 00281	GET CONSTRAINT MT.		SNSNN	NSNS	RE JECT E	MERE THERE TOO MANY CARDS	NO. NEXT ITERATION		LOAD CORRECTED ELEMENTS FROM TAPE	TABE CHECK NETECTED	ADD TO NO.	ENOUGH TRIES	NO. BACKSPACE AND TRY AGAIN TO READ RECORD				30303	VEC. SET VARIABLE CONNECTOR	0 0			Z	SET VARIABLE CONNECTOR		00320
K = 00000	E 00023 * B 00093 6 00052 04035 00069	00052 90071 00052 00052	00051 00011 TD	00051		C 00262 00011 00263 00263	* B 00024	00000 01800	K 01811 00010 * B 01816	٦, د	01811 00011	01811 00460	L 00000 01800 TAB F 01816	* B 01817	¥ CO		00011 TD	C 00475 00010 00481 00481	T STATE TYMMDD HHMM SS	00000	E 00092	00481	R 01830 01832 E 00092 * B 00001		20000

00321	00322	00323	90200	#C#00		00480 00328	00327	00328	00092 00329	00330	16 600	00050 00332	00333	00334	00335	96 600	76600	00338	66600	00340	00341	00342	00343	00344		00029 00346	00347	00348	00349		25.00	00353	00354	00355	00482 00356	15600	00358		00483 00360	4 6
		722777			C	,	RW NO.		8			(O-C) SO.														<b>α</b>			G						8				8	
		NNNNNNNNNNNNNNNNNNNNNNNNNNNNNNNNNNNNNN					T 0 C 0-C 1000RW (0-C)RW					SET TO ZERO THE SUM OF WEIGHTED (C	O THE NO. OF ITEM	FOR					SET PREVIOUS OBS. TIME		⋖	1				NEXT DBSERVATION					- LM & L		1	YES. SKIP SPO		1	PREVIOU	= PRESENT STATION NAME		
00011 PA		F0503030303030303	00700			- 1	AT. YYMMDD HHMM SS	00011 PA		+40000000+02	00+0000000+		000069 00010			10	00011 00014		10	00010	00010	00010	00010	90077 00000	90081 00419				00253 00030	- 1	00242	00477 00482 00482	00482			90003 00000	00300	00301		
P 00000 C	) 	0 00400	200	000040		* B 00480	T STAT	000000 d	0000	70000	0000	0000	H 01900	69000		69000	00014	00179	00477	00478		00048	00 049		00598		00000	000010		00028	00700	00200 7	00301			00000	00478	00479	* B 00483	

PAGE 011	00401 00402 00403	40400			00400	8 00074 00408	00400	00410		8 U1831 00412	00413			00417	00418	8 00195 00419			00422	00423			00426	00427		B 00073 00429	00430	LE 700	20100	B 00196 00432	96100	B 00196	8 00196	B 00196 004 004 008 004 006	004 004 008 004 004 004 004 004	004 004 008 004 004 004 004 004 004
LINE 00401	(ABS.VALUE OF O-C)(WT.)		DOES (MT.) (O-C) = OR EXCEED (SM) (SIGMA) +SA	SET WT. FOR COND. EQN. =	STORE * FOR PRINTOUT		STORE UNWEIGHTED (O-C)	(0-C)(M1.)	B1831 OR TO	CI'S ARE IU	U-C PXIN T.	0 = TM -S8U S1	,	YES. IS CBS. TYPE = 9 (RANGE RATE)	IS IT FIRST RANGE			STORE (WEIGHTED) COND. EQN. COEFFICIENT				PUTE PARTIAL!	COMPUTE PARTIAL OF OBSERVATION/XJ	TE FUNC			3S. WT. = 0	_ 	+ ( -	MT. IS NOT 0	WT. IS NOT	WT. IS NOT 0 ENT NORMAL MATRIX WITH COND TO SUM OF (0-C)SO AND TO N	ENT NORMAL MATRIX WITH TO SUM OF (0-C)SQ AND T TO NO. OF COND. EQNS.	ENT NORMAL MATRIX WITH COND ENT NORMAL MATRIX WITH COND TO SUM OF (O-C)SQ AND TO N TO NO. OF COND. EQNS. TO SUM OF SQUARED (O-C)'S	ENT NORMAL MATRIX WITH COND TO SUM OF (O-C)SQ AND TO N TO NG. OF COND. EQNS. TO SUM OF SQUARED (O-C)'S	WT. IS NOT 0 ENT NORMAL MATRIX WITH COND TO SUM OF (O-C)SQ AND TO N TO NO. OF COND. EQNS. TO SUM OF SQUARED (O-C)'S ERECORD ON GATED OBS. TAPE
. K = 00000	M 00050 00050 00258 M 00051 00178 00077 A 00051 00051 00079	00176 +00000		00258 00010	00176			00257	ш	01831	<b>L</b> 0	* 5 01832 F 00258 00010 00195 00195	02098 00010 00029	00253 00071 00029	00246 00029	* B 00195	00392	00420 00069	69000 69000	00010 00069	00252	28006 00000	F 00000 90088 00000	9006	00195		- 1	ш	Ç	00196	8 00196 F 00598 90089	8 00196 F 00598 90089 M 00047 00257 F 00000 90047	B 00196 F 00598 90089 M 00047 00257 F 00000 90047 A 00043 00048	B 00196 F 00598 90089 M 00047 00257 F 00000 90047 A 00043 00048 A 00049 00049	B 00196 F 00598 90089 M 00047 00257 F 00000 90047 A 00048 00048 A 00049 00049	B 00196 F 00598 90089 M 00047 00257 F 00000 90047 A 00043 00048 A 00049 00049 E 00029 B 00399

00441	00444	00446	00447		00450		00453	00455	00456	00457	00459	105 004 60	00461	00463	00464	00465	00466	00400	004 68	00400	17 400	00472	00473	00474	00472		00478	004 79	00480	_
B 00030			B 01834	- 1	8 01807							B 01806														<b>1</b> 00 a				DAGE 012
Fana and	YES. REWIND PERT. TAPE		INITIALIZE INTERPOLATION	WAS TAPE WRITTEN ON TG	NO YES	WRITE END SENTINELS ON TG			Į	REWIND 16	TO IN SECUEDING TERATIONS		COMPULE AND PRINI K.M.S. TUK HACH UBS. ITTE	SOLVE NORMAL EDNS.			CC (CONVERT CORRECTIONS)		OMPUTE A	A KATIFOL		PERIOD (MINUTES)	IS MCCI ORB.GEN.BEING USED	YES. SET P DOT = 0		N - C	N / (1d	2	_ = T00	08700 HNT -
2000	-10000	00000 00069 1FB 09220 00461 TFB		00394 00010 01807 01807	01806	02821 +9999999+08	00072	02821 00072 168 01808 01800 01800	01808	_	00394 00010	1	1	00000 90049 00000	90062	01100		00010	90059	00600 01101 03846	01116	01123		01000 \$0900	00473		00004 01120 01113		00004	

PAGE 013	00481 00482 B 00473 00483	48 400	00485	00486	00487	00488	68 400	00490	16 +00	26 +00		4K-400	5	00496	00497	00498	00499	00200	00201	00502	00.00 00.00	5	00503	00.500	00508	60200	00510	11500	OF FIT 00512	00513	00514	00212	0051	0051		B 00033 00519
LINE 00481	P DOT IN MIN/DAY	HT. OF PERIGEE (KM.)	Q.	OF APDGEE (	P.	1+6	1-E	1+E/1-E	-E-1/A	F ABOVE = VE	AI PERIGEE	YAI	1-E/1+E		A B	VELOCITY AT APOGEE (KM/HR)	AT	_	ARG. OF PERIGEE (DEG.)		OMEGA 1	R.A. UF ASC. NUUE (DEG)		CAY UMEGA DUI AURGADATI	CIN CHALL DUECAL	Colo Jakin Nio	GEOC. LATITUDE OF PERIGEE		PRINT REQUESTEC QUANTITIES AND S.D.		NS	NS	HAS MAX.NO. OF ITERATIONS BEEN RUN	NG THIS SET OF UNKNOWN	JAC ANDTHER S	
00000																															1205	1205	\ \ \ \			
<b>★</b>	00604 03870 00604 03837	01124 03846	03	03	1	-4	00011 01102	1		- 1		00610 03845	- 1	00612 01101		00612 03867	١.	01113 03847	1	-	in i		8	0001/0383/	- 1	00618 00619		1	00000 19006	S	00011 PA 1.	1	4 00033			- The second second second second second second second second second second second second second second second
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0.000			
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00036	LUAU ANDINCA	8 00036	00522
T RUN TERMINATED AFTER	ITERATIONS		00523
0011 TD	SSN		00524
00011	SSN		00525
HE END			00526
00000 00011 TD			00527
00011			00528
-10000			00529
1	SET TO EOF TD		00530
00 0000 TD			00531
00032 00010 00264	DOES LOC. 00032 CONTAIN 0	and the second of the second o	00532
00032	END OF RUN		00533
00264	JUMP TO ANOTHER PROGRAM	8 00264	00534
00032		enter and the second se	00535
			00536
01		B 01810	00537
144	ORD OF BINARY OBSERVATION TAPE ON TB		00538
00000 00011 PA			00539
00010			00540
01820		8 01820	00541
APE CHECK ON BINARY	ELEMENT TAPE ON TA		00542
			00543
00010			00544
00061	HERE FROM OBSERVATION	B 00061	00545
	IF IT FINDS TAPE CHECK ON TB		00546
I TAPE CHECK ON BINARY OB	VI.		00547
011 PA			00548
1	I DAD NEXT RECORD FROM DBS. TAPE		00549
00062	AD TITLE	B 00062	00550
	TAPE READ E. IE FIND TAPE CHECK	1	00551
T TABE CUECK ON COMDIENCALTABY	DEDTIODATIONS TABE ON TE		00552
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I TAPE CHECK UN SULAK PERI	- Arr ON -r		
K = 00000	LINE 00560 P	PAGE 014	
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<u> </u>	00601	00603	0000	0000	00000	00000	60900	00610	00611	00612	00613	<b>4</b> 1900	00615	00616	11900	00618	61900	006 20	00621	00622	00623	00624	00625	00626	00627	00628	00629	006 30	00631	006 33	96900	00634	00636	16900	96900	00639	09900	
PAGE 016				,																																		PAGE 016
LINE 00601	11,	40,41,69-71, IN KUM 8 CULUMNS 2,2710:121169 21,30,33, IN ROW 9 COLUMNS 2,14,16,41-43,		can .																																		04900 HNT:
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LINE  O. OF LINES PRINTE  TR.+7  VE ALL REQ. LINES  S. SCALE SIGMA  ALE CHANGE IN SIGM  T 4 WORDS DESCRIBI  UANTITIES  C. UF Q SUB 1  SINB 1	0100  (NO. OF LINES PRINTED)+1  CNTR.+7  HAVE ALL REQ. LINES BEEN  YES. SCALE SIGMA  SCALE CHANGE IN SIGMA  SCALE CHANGE IN SIGMA  NO  GET 4 WORDS DESCRIBING  GUANTITIES  LCC. OF Q SUB 1	(NO. OF LINES PRINTED)+1   CNTR.+7   HAVE ALL REQ. LINES BEEN   YES. SCALE SIGMA   SCALE CHANGE IN SIGMA   SSS   0100 (NO. OF LINES PRINTED)+1 CNTR.+7 HAVE ALL REQ. LINES BEEN YES. SCALE SIGMA SCALE CHANGE IN SIGMA SCALE CHANGE IN SIGMA SCALE CHANGE IN SIGMA NO GET 4 WORDS DESCRIBING GUANTITIES LCC. OF Q SUB 1	
O. OF LINE VE ALL REQ S. SCALE S ALE CHANGE T 4 WORDS UANTITIES C. OF Q SU	0100 (NO. OF LINE CNTR.+7 HAVE ALL REQ YES. SCALE S SCALE CHANGE SCALE CHANGE NO GET 4 WORDS OCANTITIES CUANTITIES	1515060403050403 1515060403050403 1515060403050403 1515060403050403 1515060403050403 1515060403050403 1515060403050403 1515060403050403 1515060403050403 1515060403050403 1515060403050403	N = UOIUU  00006
	FION 151506040		000009 000009 000027 000027 000027 000009 000009 000009 000009

0404041504090304030503 SAAAASSNINNINN 0404041504090304030503 SAAAASSNINNINNIN 0404041504090304030903 SAAAASSNINNINNINNINNINNINNINNINNINNINNINNINNI	041504090304030503 SAAAASSNINNNN 041504090304030903 SAAAASSNINNNN (Q SUB 3)		00721 00722 00723	0072 <del>4</del> 00725 00726	00727 00728 00729	00730 00731 00732	00733 00734 00735	00736 00737 00738	00739 00740 00741	00742	00745	00748 00749 00750	00751 00752 00753	00754 00755 00756 00757	00758 00759 00760
100 LINE 00 404041504090304030503 404041504090304030903 4040409030403090304030903 15 0UTPUT ON PRINTER NO YES 4040409030403090304030903	K = 00100			മ	1			1							
100 LINE 404041504090304030503 404041504090304030903 4040409030403090304030903 YES YES 4040409030403090304030903	K = 00100  00006 TD 010404041504090304030503  00005 00031 00031  00002 00024 Q SUB 3  00002 00023 (C SUB 3)-(Q SUB 2)  00005 00023 (C SUB 3)-(Q SUB 2)  00005 00012 00012 IS 0UTPUT CN PRINTER  NO  NO  NO  NO  NO  NO  NO  NO  NO  N	0721	SAAAASSNNNNNN	SAAAASSNNNNNN			SAAAANNNNNNNNN	SAAAANNNNNNNN							
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			AR IDENT.											<b> </b>																			£ 00800
D.C.	- V - m.d. (Mag.	Z (	FOUNTORIAL CR POLAR	;	STATION LABEL		YR MO OY	OF CEP	DRC TVDF	OBSERVED GUANTITY	COMPUTED QUANTITY	D-C (UNWEIGHTED)	O-C (WEIGHTED)		$\Box$	CUTPUT SCALE				10 10 3						08S. TYPE (XY)/10		X+90=9X=ALPH.REP.	0×	<b>&gt;</b>	Y+90=9Y=ALPH.REP.	9 X 9 Y	
FUNCTION FOR																		red si		› <del>4</del>					Ö	3							
(0-C) PRINT	00042	60000	00493	00045	00300	00301	02825	02027	00053	00255	00256	00254	00257	00258	00245	02521		+10000000+01	00*000000+	+1000000+04	90300	90301	92825	07076	C	90253 90002	00016		- 1		91	00017 90003	>
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rate of															8 0000		8 0000																			PAGE 021
= 00150	CR PULAR IDENT	٠ سبيم		CCMPUTEL QUANTITY THRU OUTPUT SCALE		SCALE N-C (UNNEIGHTED)	į		SCALE U-C (WEIGHTED)	C 2	0.2.0 & 0.2.0	0104020705060404030403040304030704030503SAANNANNNNNNNNNNNN			<b>0000</b> 00	0.000		010402070506040463040304030704630503SAANNANNNNNNNNNNNNN																		= 00150 LINE 00840
¥	90003	90255	900C0	90256	- 1		90000	00027		90000		TO	80000		00000	2007		٧d																		×
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14800	00842	00843	00844	00845	00846	00847	00848	00849	00850	00851	00852	00853	95800	00855	95800	00857	00858	00859	09800	19800	00862	69800	00864	00865	99800	19800	89800	40800 40800	008 00	12 800	00872	00873	00874	008 75	9/800	00877	008 78	6/ 800	00880	
)ICTIONS)			WORKING STG. (BEGIN COMMAND)	WORKING STG. (BEGIN COMMAND)		(RHO BAR)*	RHO DOT BAR			BHD DOT		G I BAR		1 S S S S S S S S S S S S S S S S S S S	1			1		100	i	DOT	DOT	DOT	G 7 DOT BAR		OR MOVE	DINCT	VECTOR MAGNITUDE	SUBTRACT	ION	SCALAR-VECTOR MULTIPLY F	TAN (Y/X)			R BAR, SATELLITE POSITION VECTOR I		ш	1	
CAL STATION PREDICT								•																						-		- 1								9.9
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	00881	00882	00883		00001 00885	00886	18800	88800	00889	06800	00891	008 93	00894	00895	96800	00897	86800	00899	00600	10000	00903	40600	50900	90600 15006	10600		90058 00909	000	00011 00912		00014	00015	00916	00917		00920	3	
					) 8 9																			106 8		1	06 8		800					ď		And the second s	PAGE 023	
	COMPUTED OBSERVATION O	CY.		RANGE AMBIGUITY RESOLUTION F.	LSP.										فالمستراب المستريخ والمسترك وا						085. = T	YES. USE GI BAR	63	085. = THETA 2	G2 EAR	\$					CCMPUTE (R BAR)* FOR IGNOSPHERE F.	RANGE FOR IONOSPH	- 1				1 INF 00920	1
K = 01200	0 00000 00254		90002	90003	00001			F 00047 00089 00096	F 00050 00090 00047	F 00085 00086 00097	00089	00085 00086	00000	7 00059 00088 00047	00003 000001	00035 00031	00000	00061 00006	00026 00091	00 086 00 095	00010 00100	00035 00086	- 1	E 00027	o u	00028		6 00010 00010 00070	u i	* 8 00011 r 22250 02088 02047	90002 00090	00039 90001	90003	00002	00012	F 00099 00087 00071	01200 ± X	11

PAGE  B B B B B B B B B B B B B B B B B B	INDSPHERE  INDSPHERE  AZIMUTH  AZIMUTH  AZIMUTH  AZIMUTH  B  ELEVATION  ELEVATION  B  RIGHT ASCENSION  B  RANGE RATE  CCAL HOUR ANGLE  B  RANGE RATE  CCRPUTE (R SAR1** FOR IONOSPHERE F. CORRECT RANGE RATE  CCRPUTE (R SAR1** FOR IONOSPHERE F. CORRECT RANGE RATE FOR IONOSPHERE  L-RATE  N-RATE	INDSPHERE  INDSPHERE  AZIMUTH  AZIMUTH  AZIMUTH  B  RIGHT ASCENSION  ELEVATION  B  RIGHT ASCENSION  B  RANGE RATE  CCRPUTE (R BAR1** FOR IONOSPHERE F.  CCRRECT RANGE RATE FOR IONOSPHERE  L-RATE  RANGE RATE  CRRECT RANGE RATE FOR IONOSPHERE  L-RATE  B  M-RATE	= 01200	K = 01200         LINE 00921         PAGE           00059         INDSPHERE         B           00050         AZIMUTH         B           00072         AZIMUTH         B           00072         ELEVATION         B           00073         ELEVATION         B           00076         RIGHT ASCENSIGN         B           00077         LCCAL FOUR ANGLE         B           00074         LCCAL FOUR ANGLE         B           00005         CCRRECT RANGE RATE FOR IONDSPHERE F         B           000050         CONDTE         B           000050         CRRECT RANGE RATE FOR IONDSPHERE F         B           000050         CONDTE         B           000050         CRANGE RATE FOR IONDSPHERE F         B	00921 00922 00923	00924	00927	00928	00630	00931	00933	00934	96 600	00937	00000	00940	14600	24600	00944	00945	00946	00347	00949	00950	00951	00953	00954	00955	00920	00958	00959	09600	
ENSIGN  R ANGLE  BAR  R SAR)* FOR IONOSPHERE  ANGE RATE FOR IONOSPHER	INDSPHERE  AZIMUTH AZIMUTH AZIMUTH ELEVATION  RIGHT ASCENSION  CCAL FOUR ANGLE  CCRPUTE (R BAR)* FOR IONOSPHERE CCRRECT RANGE RATE	= 01200 INDSPHERE INDSPHERE AZIMUTH AZIMUTH AZIMUTH AZIMUTH AZIMUTH AZIMUTH CCAL FOUR ANGLE LCCAL FOUR ANGLE CCMPUTE (R SAR)* FOR IONOSPHERE CCMPUTE (R SAR)* FOR IONOSPHERE CCRRECT RANGE RATE L-RATE	K = 01200         LINE 00921           00099         INDSPHERE           00072         INDSPHERE           00072         INDSPHERE           00072         AZIMUTH           00072         ELEVATION           00073         ELEVATION           00074         RIGHT ASCENSICN           00032         CCAL FOUR ANGLE           00074         CCRPUTE (R BAR)           00075         CCRPUTE (R BAR)           00095         CCRRECT RANGE RATE           00095         CCRRECT RANGE RATE FOR IONOSPHER           00056         CCRRECT RANGE RATE FOR IONOSPHER           00056         CONPUTE           00056         M-RATE	K = 01200	1	·				000					1000	1000	i																
	01200		K =	90001 00099 90001 00099 00085 00072 90001 00099 00085 00072 00086 00073 00086 00077 00086 00077 00086 00077 00086 00077 00086 00077 00086 00076 00087 00077 00087 00077 00087 00095 00087 00095 00087 00096 00087 00078 00087 00056	INOSPHERE		INUSPHERE	AZIMUTH	P.C.R.10.11	NC W			Ø					FOUR		]			(R SAR)* FOR IONOSPHERE	RANGE RATE FOR	U <b>+ «</b> C )						M-RATE		

00 079       UINE         00 079       00 079         00 072       00 072         00 073       00 073         00 073       00 076         00 074       00 078         00 075       00 076         00 074       00 076         00 077       00 077         00 077       00 074         00 074       00 074         00 078       00 074         00 078       00 074         00 078       00 074         00 078       00 074         00 078       00 074         00 074       00 074         00 074       00 074	PAGE 025	00961 00962 00963	49600	69500 8 00022 009 <b>66</b>	19600	00968 00969	07 900	009 <b>71</b> 8 00023 009 <b>7</b> 2		52.600		8 00024 00977	62 600	08 600	18600	1.00	8 600 67000 8 70 000	00985	98 600	8 00026 <b>0</b> 0987	88600	68 600	06600	16.600 16.600	26.600°	8 000 7 000 8	00061	96600	76600	86.600	66600
K = 000079 000056 000056 000072 000072 000073 000073 000074 000074 000074 000074 000071 000074 000071 000050 0000050 000050 000050 000050 000050 0000050 000050 0000050 000050 000050 000050 000050 000050 000050 000050 000050 000050 000050 00	LINE 00961												عاميا والمرابع والمرا																		
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	01001	01003	90010 90010	0000	01007	01008	01000	01010	01011	01012	01013	1010	01010	01017	01018	01010	01020	01021	01022	01023	+2010	01025	12010 02000 B	1	01029	01030	01031	01032	01033	01034	01035	01036	. 01037	01038	01039	04010	1000
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	01041	01043	01044	01045	01046	01047	01048	01049	05010	01051	01052	01053	01054	01055	01056	01057	01058	01059	09010	01061	01062	01063	01064	01065	99010	19010	01068	01069	01070	01071	01072	01073	01074	01075	01076	01077	01078	01079	01080
PAGE 027		And the second s		, bord 3		<b>  </b>	<b></b>		pm <b>4</b> )	hout 1	<b>I</b>	O	0	0	0	H	p-ut	<b>J</b>	<b>  </b>	<b>h-set</b>	<del>jud</del>	0	0	0	0	0	0	C	0	0	Jeri	0	u.	L.		<b>l.i.</b>		ı.	u.
LINE 01041			ELL IPSE	FPOCH TIME IN C.U.T.	SEMI MAJOR AXIS AT T, TIME OF OBS.	NTRICITY AT T	ANOMALY AT	OF PERIGEE AT TIO	LONG. OF ASC. NODE AT T(0)	73		S(T)	C(T)	OF X	V (MAGNITUDE OF V BAR)	-	MEAN ANCHALY AT I	ARG. OF PERIGEE AT T	INCLINATION AT T	LONG. OF ASC. NOCE AT I	MEAN MOTION AT T(0)	ALPHA BAR	ALPHA SUB K	<b>A</b>	CANMA SAR	P BAR		BAR (POSITION	(T) RECCENTRIC	BAR (VELOCITY VEC	T, TIME OF OBS. IN C.U.T.	- 1		٥.	VECTOR MAGNITUDE	$\Box$	PRINCIPAL VALUE	SIN	SOO
K = 01300	PE (POSITION IN ELLIPSE)			01100	00218	00219	01113	01115	01117	03864	03852	00201	00202	00210	00211	01112	00213	00215	00220	00217	01119	00222	00224	00225	00228	00231	00234	00237	00240	00241	00200	02085	02001	02061	02011	02196	02271	02101	02105
		K 00000	K 01300	00006 0	0 90001		0 90003	- 1	90006	90006	90007	80006	60006	90010	90011	9001	0 90013	9001	9001		0 90019	9002	9002				90034	0 90037	0 90040	i .	0 3008 0	1	90086	90087	90088	[	60.06	0 90091	0 90092

KEPLER   CLOST   CLO		01081	01083	01084	01085	01036	01087	01088	01089	01001	01092	01093	01094	01095	01096	16010	01038	01099		01102	01103	01104	01105	01106	01107	07170	01110	01111	01112		-			01117	01118	01120	
K = 01300         LINE           02216         KEPLER ADD           02031         VECTOR ADD           02041         SCALAR MULTIPLY           +50000000+01         SCALAR MULTIPLY           +60000000+001         ZERO           +000000000+002         ZERO           +100000000+003         LP A           90001         LP A           90002         LP I           90003         LP I           90004         LP I           90005         LP I           90007         LP I           90008         90023           90009         POOZ           90009         POOZ           90009         POOZ           90009         POOZ           9009         POOZ		LL L	· U	manufactured for the control of the	10000 8									e e debe de la manufactura de la manufactura de la manufactura de la manufactura de la manufactura de la manufac			- Complete C																				
02216         02031         02041         02041         02041         02041         +500000000+01         +400000000+01         +000000000+01         +000000000+01         +000000000+01         +00000000+01         +100000000+01         9001         9002         9003         9004         9005         9007         9008         9008         9009         9008         <				1				ZERO	ZERO	₩ Q <b>-</b>		1								1 00														<b>∀</b>  -   Lu		1	
@@@@@>>>>>\\\\\\\\\\\\\\\\\\\\\\\\\\\\	0	90093 0221	90095	96006	0000	00007	00000	00049	000050	00001	00033	00035	00015 90089	00015 00015	00036 90000	00023 00034	00023 00051	00023 90089	00023 00033	00026 90091 9001	00027 00050	90085 90086	90085 90087	00015 90091	90022 90096	00015 90092	90024 90024	90085 90087	16006 51000	00020 90096	90085 90088	00015 90092	90085 90098	90025 90094	90085 90086 9002	00015 90015 9001	

V BAR R MAGNITUEE V MAGNITUEE  V MAGNITUEE	01161	01162	01164	01165	01166	01167	01168	01169	01170	01172	01173	01174	01175	01176	01177	01179	01180	01181	01182	01184	01185	01186	0110	01189	0110	01191	01192	01194	01195	0110	0110	01199	01200	
90095 500020 90095 90036 N MAGNITUE 90088 90037 V MAGNITUE 90088 90041 V MAGNITUE  10088 90041 V MAGNITUE 10088 90041 V MAGNITUE																																		030
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90095 00020 90096 00036 90088 90037 90088 90041 V MAGNITUDE  V MAGNITUDE  K = 01300																												The second secon						LINE 01200
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	01201	01203	01704	01702	90710	01207	01208	01209	01210	01211	01212	01212	01215	01216	01217	01218	01219	01220	01221	01222	01223	01224	01225	01226	01227	01228	01229	01770	01231	01233	01226	01235	01236	ıl٨	10	01239	01240	
031																																						031
PAGE				,																					The second second													PAGE
LINE 01201	CTED ELEMENTS		WHICH BIN. RECORD WR	(PREVIOUS ELEMENTS, WHICH WILL	REPLACED BY CORRECTED ELEMENTS	NU THE FUNCTION)	ANGLE DELTA	Z.	Hd :	ARG. OF PERIGEE		LONG. OF NODE	2 H	-30	2 1	HI OF TREETE		וא אחם אטם	S C C C C C C C C C C C C C C C C C C C	I CC OF BOS. + VEI	A (REPLANY PREVANT	T. OF PERIGEE	HT. 0F	N (2,Q) GR DELTA RHO SUB I	S R	PERIOD	AR (REPL. BY PREV. VECTORS	11 LUC. UF PKEV. ELEM.	J.	COMPONENT	M CAU. OF UNKNUMNS!	(	SUSTINATION OF THE PROPERTY OF		77 1N	MA) /SIGMA	2420	LINE 01240
K = 01355	AND STORE CORRECT													The second secon																							All Andrews and Andrews and Andrews and Andrews and Andrews and Andrews and Andrews and Andrews and Andrews and	K = 01355
	CUMPUTE	3	01100	1011	1102	1103	1112	1113	11114	11115	1116	11117	01119	0211	11123	1124	2711	01150	01170	11190	+011	0000	10524	0525	0545	10522	00503	0509	13851	13852	00599	00048	00049	00178	00065	00082	02196	! <b>!</b>
	1	01355	0 00006			}			1		- )			- 1					0 71006			90020 0						- 1		90031 0	- 1			, ام	0 1		90040 0	
	1	<i>L</i> ×	0	C	O	0	1.3	<b>7</b> (2)	a	Ø	3	Ø	Q' !	3	Ø	<b>3</b>	9		.87		<b>3</b> (	3 C	90	7 (3	Ø CO	3	C	C	O	a	a	œ	O :	9	3 (	œ	3 0	,

01241 01242 01243		,	01246	01247	01248	01249	01250	01251					01256	1 01257	[ 01258	B 00001 01259	01260	01261	01262	01263	<b>49710</b>	01265	01266	01267			N 00022 01270	01271	01272	01273	01274	01275			01279	01280
ANGLE RED. ABSOLUTE VALUE F) FMENT 10A0 (CONVERSION OF F1 FM.)	OF NORMAL OR SPECIAL EN	OF 11 LOC. OF PREV. ELEM.	VALUE OF (1+ RHO 1) (C SUB	ALUE	C SUB D	VECTOR MAGNITUDE F.	OR (REPL.BY CORR. 1	MAG. VEL. (REPL. BY CORR. IN	. (REPL.BY PREV. IN F.)	OC.OF 9 DELTAS (REPL.BY PREV.EL.IN	ORBIT GENERATOR INITIALIZE F	ORBIT GENERATOR	OBS. TIME IN C.U.T.	ORBIT GENERATOR IDENT.	Z pI								STORE SIGMA		S MCOI ORB. CEN.BEING USED	<₹.	CONVERT THEM TO (CORR.) ELEMENTS	POS. OR VE	DELTA COMPONENT (FROM CC)			STORE CORR. COMPONENT	. Jud addy of	A.rus. T VEL.VECTURS	YES. SIUKE PKEV. A	PREV.
90041 02271 90042 02751 90063 02936	1		- 1		90053 00267	90054 02011	1	90056 01111	90057 00510	90058 00511	90069 04491	90070 04501	90071 00200		00008 03842	1	00000 +00000000+00	00000 +100000000+01	00007 +200000000+02		00012 +10000000+03		00013 90035	•		00021 +60000000+01	00022	61006	90027	00016	000014	00014	00014		90020 90001 90030 00005 00003	000018

PAGE 032

LINE 01241

K = 01355

PAGE 033	N 00022													N 00022												B 00010												11000 N
LINE 01281	SICRE PREVICUS MAG.R, MAG. V. DELTA, MEAN ANDMALY, PHI, ARG. OF	PER. , INCLINATION, LONG. OF NODE,	ELEV. V, N, AND ECC. ANCMALY			PREV. PERICO	STORE PREV. HT. CF PERIGEE	LE PREV. HT. CF APOSCE	PECIAL ENTRY TO FL	LOAD AND COMPUTE CORRE	A CCRR. POS. +	MAX.NO.CF RHO SUB I			UH.	( )	CELTA = CO	PRFV.RHC SUB	STURE CCRR. RHO SUB I			7.	DRK, RHO SUB 1			S	<b>V</b>	PREV. A	≺1		STORE PREV.E	E + CELTA	DELTA NU (FROM CC)	SICRE PREV. NU	0 + JN			CEMPUTE AND STORE CORR.DELTA.
01355																																						
1   <b>⊻</b>	00005	90050 00014	00014 00015	00014 00006	00014 00022	90012	90013	90014	+1000000001+	90043 00000		+20000000+01	00000		90016 00014	90023 00014	1	00014 00015	00014 00016	90000 +1000	00014 00022	90051		90051 90053			90020	90001		90020 00006	00006 90002		90020 00018		ł	+900000000+	60005	
	R 00014 I 00021 N 00022	00015		41000	00021	90025		90022		00000		I 00021	000014	1	10	00016	91000		90016	00014		90052	90051	90051	E 00019	8 0001D	91000	90020	A 90001	00018	90020	80006	00016		90003	I 00001	00014	11000 N

PAGE 034				
LINE 01321	MEAN ANGMALY, PHI, ARG, OF PER.,	INCLINATION, LENG. OF NODE,	FLEV. V, N, AND ECC. ANCMALY,	
K = 01355	5 00015 90004 00014	6 00016 90058 00014	A 00016 00016 00015	H 90053 00014 00015

PAGE 034	01321 01322 01323	01324	01325	01326	01327	01328	01329	01330	01331	01332	01333	01334	01539	00000	01557 01328	01230	01010	01341	01342	01343	01344	01345	01346	01347	01348	01349	OC C TO	01351	01332	01354	0100	01356	01257	01358	01359	01360	PAGE 034
LINE 01321	MEAN ANDMALY, PHI, ARG. OF PER., INCLINATION, LONG. OF NODE, FLEV. V, N, AND ECC. ANCMALY,	ERE CORR. FLEM.=	TA CLE	STURE	OVER DELTA ELEM.		enter of the second second second second second second second second second second second second second second																														LINE 01360
K = 0.1355	\$ 00015 90064 00014 6 00016 90058 00014 A 00016 00016 00015	90059 00014	90004 00014	00014 00014	C 00021 00014 00011	•												49(																			K = 01355

	01361	01362	01364	01365	01366	01367	01368	01369	01370	01371	01372	01373	01374	01375			01378		01380	10010	01582	01384	01385	01386	01387		01389	01390	19610	01392		01394	01395	5 6	013	01398	7	
PAGE 035					· · · · · · · · · · · · · · · · · · ·				8 00001							8 00010			8 0000	1	77000 8					R 00013	; ; ;				8 00014			1000	ernon a			PAGE 035
LINE 01361			O LORG. DUANTITY IN RAD.)	COORDITION OF TAXABLE	PE	(TOLERANCE IN RAD.)			GLE REDUCER FOR ABS. VALUE OF I	POSE IS TO REDUCE COMPL	O-C) IS LARGE DUE TO DIFF.	AROUND 2 PI. ALL INPUT AND DUTPUT IS Q'D.	SES 18 LOCS.)	IS 08S. TYPE = 6 (RT. ASC.)	YES. TEST		IS OBS. TYPE = 4 (AZ.)	YES. TEST		1000 11FE = 0 1L. N. A.	VES. 1EST		TO C EDITAL TO DE LECC THAN T	CET C - C + O DI	2. 3. 3. 136 .63		I C D FOUNT TO OR LESS THAN T	ES. 1S C FOUAL TO OR MORE	SET $C = C - 2 PI$			BS. TYPE = 17 (THET	YES. IS THETA I MORE THAN PI	TO MODE TO A	1 IS MUKE	$(1HETA \ 1) - (2 PI)$		LINE 01400
K = 01380	1	K 00000	A UL 380	10000	0 90003 00253	40006	0000	00017	00001	V 00006 +6000000+01	V 00007 +40000000+01	V 00008 +80000000+01	00015 +17000000+02	C 90003 00006 00009 00010	E 00011	00010	C 90003 00007 00002 00002		60000	- 1	00011	5 00012 00003 90004	20002 20002	#0006 70006	30006	1	۵ <i>ل</i>	00012 90002	90002 90002	00002	* B 00014	90003 00015		w)	81000	S 90002 90002 00005	[	K = 01380

	01401	01402	01403	40410	C0 # 10	00+10	10410	01408	01409	01410	01411	01412	01413	01414	01415	01416	01417	01418	01419	01420	01421	01422	67410	47410	01425	01426	01427	01428	01429	01430	01431	26 4 10	01433	01434	01435	05+10	01437	01439	4	i		
PAGE 036			the matter of the second secon			,									And the second later and the s			The second secon																		- 1	8 00001				PAGE 036	
LINE 01401				CZX	ALUE FUNCTION	OF T(P,Q)'S		-	N, MEAN MOTION AT T(0)	ANDWALY AT	ECCENTRIC ANOMALY AT T	ALPHA 3AR	BETA BAR	GAMMA BAR	P BAR	BAR	(COMPUTED BY POS. IN EL	R DOT BAR (COMPUTED BY POS.IN ELLIPSE) I	, OBSERVATION T	POCH TIME IN C.U.T.	R BAR (COMPUTED BY POS.IN E.)			MULTIPLY	SUBTRACT	VECTOR ADD	UCT		SQUARE ROOT		SINE		S (T)		ARTIAL R BAR / X J	R DOT BAR / X J	POSITION PARTIALS		ADDECT FOLIAL OR EXCEEN 74	ט ויפטאן טא ויאטוינט	LINE 01440	ı
K = 01400	PP (POSITION PARTIALS)	K 00000	K 01400	90009 0025	89006	90069	90070	90071	0 90072 01119	90073	90074		90016	7006	90078	90079	90080	90081	90082	0 90083	90084	0 90085	90086	90087	90088	68006	06006	16006	02	90093 02	90094 0210	90095	0020	76006	0 90098 00380	66006	10000	S 00007 90082 90083	00000 00000	61000 60006 26000 3	K = 01400	

	01441 01442 01443	01444	01446	14410	01448	01450	01451	01452	01453	01454	01422	01456	25420	01458	01459	01460	01461	01462	01403	10110	01465	01466	01467	01468	69410	01470	01471	01472	01473	01474	01475	01476	01477	014/8	01480	
PAGE 037	B 00079	,	8 00101													`				-	8 00102											فيسكنان والمهاب البيان المباد أماروك المباوط أميار فأمير والمبادرة والمتازم والمتازم والمتازم والمتازم والمبادرة				PAGE 037
LINE 01441	YES 60 TO B (20 + J) J IS LESS THAN 74																																			LINE 01480
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TAGE 12.			B 00001			8 00007			8 00008																					PAGE 127
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		LATITUDE		05092
		HEIGHT		05093
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V 00009 +40000000+01	DAY. EXIT WITH (X)=NUMBER OF DAYS FROM	,	05726
V 00010 +1000000+01	JAN. 1 OF THE GIVEN YEAR THROUGH THE		05727
V 00011 +00000000+00	GIVEN DATE.		05728
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V 00013 +5900000000	OF DAYS UP TO		05730
1.84	NO. OF DAYS UP TO APR. 1		05731
•	NO. OF DAYS UP TO MAY 1		05732
	NO. OF DAYS UP TO JUNE 1		05733
V 00017 +1810000+03	OF DAYS		05734
•	OF		05735
	NO. OF DAYS UP TO SEPT. 1		05736
V 00020 +27300000#03	NO. OF DAYS UP TO OCT. 1		05737
	NO. OF DAYS UP TO NOV. 1		05738
V 00022 +33400000+03	١.		05739
I 00028 +00000000+00	DAY LOCATOR TO		05740
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C 00024 00026 00005	<b>FONTH COUNTER</b>		05747
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C 00024 00008 00031	IS MONTH GREATER THAN 2		05749
<b>*</b> 8 00029		8 00029	05750
H 00001 00004 00027	STORE THE DAY COUNT		05751
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K = 03300 PAGE 145	_ II			

.0000	10860	05802	05803	05804	B 00001 05805	•	90860	05807	05808	05809	05810	21000	TTOCO	05812	05813	05814	05815	B 00020 05816	-2	B 00015 05818	05819	05820	05821	05822		8 00016 05824	05826	8 00019 05827		05829	05830	B 00021 05831	05832	05833	05834	00000 00000	05837	05838	05839	05840	146	١
	DAYS FUNCTION			DAY COUNT FUNCTION	20.00	THE THEORY COMMITTIONS I TO SEL	EL F. IUSES LUCALI	*YEAR OF THE REFEREI	DATE, (Z+1)=NO. OF DAYS FROM JAN. 1 OF	H THE DAY OF REFER	THUM GV3/ - XT/ 12T/ 1	TEAC SOUTH TO THE TOTAL TO THE TOTAL	VALIEN DAIE.	THE NUMBER OF DAY	ATE THROUGH THE OBSER	DATE, (TOE. THE JULIAN DAY COUNT FROM	REFERENCE DAY.)												and the second s					***							FINE OSSAC	0,2040
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REDUCED JULIAN DAYS- SECOND 03843 03862 +10000000+001 +00000000+00 00001 00003 00010 00005 00010 00007 00003 000009 00009 00007 00009 000006 00010 00006 00010 00006 00010 00006	I I B 00001 (Z+1) = SECONDS (Z+1) = SECONDS = CUT . 15 LOCS.  DNE DAY  B 00015	05842 05843 05844 05845 05845 05847 05847 05840 05850
03843       SEC/CUT         03862       SEC/DAY         +10000000+01       ENTER WITH (2) = J.D.         +100000000+00       ENTER WITH (2) = J.D.         +00001 00003       ENTER WITH (2) = J.D.         00001 00003       LESS THAN 86,400, (X)         0001 00003       DO SEC.         00010 00005       VES.         00009 00006       J.D.         00009 00007       J.D.         00009 00006       J.D.         00010 00006       J.D.         00011 00006       J.D.         00012 00006       J.D.         00013 00010       J.D.         00014 00001       J.D.         00015 00006       J.D.         00016 00006       J.D.         00017 00006       J.D.         00018 00006       J.D.         00010 00010       J.D.	T. F. I B B +1) = SECONDS +1) = SECONDS CUT . 15 LOCS.  NE DAY  B B	05844 05846 05846 05846 05846 05846 05850
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+00000000+00 EXIT WITH (Z) = 00001 00003 LESS THAN 86,400 SEC 00003 SECONDS DO SEC. EQUAL OR YES. (SEC)-(SEC. 00010 00009 00008 J.D.+1  000003 00000 J.D.+1  000009 000007 J.D. AND REDUCED SEC. 00011 00006 O0006 O0010 SEC. 00011 O0006 O0011 O0006 O0006 O0006 O00011 O0006	= SECONDS • 15 LOCS•  DAY  B	05849 05850 05850
00001         00003         LESS THAN 86,400           00002         00003         SECONDS           00010         00007         YES. (SEC)-(SEC.           00009         00008         J.D.+1           00003         00009         J.D.+1           00009         00007         J.D. TO SEC.           00009         00006         J.D. TO SEC.           00010         00006         J.D. TO SEC.           00011         00006         J.D. TO SEC.           00011         00006         J.D. TO SEC.           00011         00006         J.D. TO SEC.	DAY B	0584 0585( 0585)
00002         00003         SECONDS           00010         00005         DG SEC. EQUAL OR           00010         00007         YES. (SEC)-(SEC.           00009         00009         J.D.+1           00003         00000         J.D. AND           00003         00010         REDUCED SEC.           00009         00006         J.D. TG SEC.           0001         00006         O0006           0001         00006         O0011           0001         00001         O0011	DAY B	0585
00010         00005         DD SEC. EQUAL OR YES. (SEC)-(SEC. 000010           00010         00007         YES. (SEC)-(SEC. 00000. 00003           00003         00009         J.D. AND REDUCED SEC. 00000. 00006           00009         00006         J.D. TO SEC. 00011           0001         00006         00011           0001         00004         00011           00010         00014         ARE SEC. ZERO OR	B B	0585
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			K = 03400	LINE 05920	PAGE 148	

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<b>K</b> ≈ 03800	CONSTANTS POOL FUNCTION		66000	00089	02196	02101	02105	+20000000+01	+40000000+01	+15000000+01	+25000000+01	+60000000+02	+30000000+01	+100000000+01		-26666667+01	-37500000+01	+800000000+01	00041	i .		00088 00041		06000	00092			- 1			00042 00006	+12000	00042 00031	1	00073	00046	00046 00072	
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PAGE 164																														01000												PAGE 164
LINE 06521	(MI/C 11 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	2	O COLLA	LE IENS	(METERS/C.U.L.) (C.U.I./SEC.)	CCMPUTE RAD/DEG	Ξ	LESS				**Z = Zr - r**Z \E-EUCHINICIII OI		AK KAUIUS UF EAKIN IN	TAD/OR UD/W#M	TIME ACTION OF A CONTRACT OF A	, L		<u> </u>	12*R**2	J=3/2+J2*R**2	IS PE DRBIT GENERATOR BEING USED	K2=0	K3=0	K4=0	K5=0	N=0	K=0	0=1			•	COMPUTE KZ	Z.I	COMPUTE K3		₽}	COMPONE K4	サード・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・	K4=(J4*R**4)/-2.6666667	COMPUTE K5	LINE 06560
K = 03800	C/000 = 1000	00045 00067	00070 00043	M 00040 00040 00009	D 00066 00040 00043	0 00068 00011 00047	00048	00034 00011	71000 H0000	00034	00086 00005	98000 98000	00011	00085 00049	00074 00076	11000	28000	11000	000084 00046	000050 00043	00000	9006 7 00011	00+00000000+ 00000		I 00032 +00000000+00	00033	000059	I 00028 +00000000+00	1	E 00002	00019	01000 19006	00024 00053	00030 00024	00021 00020	00024 00054	00012	00022 00021	00024 00055	D 00032 00024 00013	M 00023 00022 00049	K = 03800

J5*R**5     06563       K5=-(J5*R**5)     06563       H=0     06564       L=0     06564       K2=0     06569       K2=0     06569       K2=0     06569       K2=0     06572       K3=0     06572       K4=0     06572       K5=0     06572       CMPUTE H     06572       CMPUTE K     06578       K2=J2*R**2/2     06584       CMPUTE K     06584	R**5)  GRBIT GENERATOR BEING USED  H *3*4.3  K ***4*4.4  HST ORBIT GENERATOR  K3 **3.7  K4  ***4./8	###5)  GRBIT GENERATOR BEING USED  # 33*J3  KR # * * * * * J4  HST ORBIT GENERATOR  # 3 * 2 / 2  KR # * * * 4 / 8  * * * 4 / 8  * * * 4 / 8	
R**5)  ORBIT GENERATOR BEING USED  H  *3*J3  K  R**4*J4  K2  *2/2  *3/2  *3/2  *44/8	###5)  GRBIT GENERATOR BEING USED  #33*J3  K##4*J4  HST ORBIT GENERATOR  #31/2  K4  **4/8	####5)  ORBIT GENERATOR BEING USED  #3*J3  K##4#J4  HST ORBIT GENERATOR  #31/2  K4  **4/8	
DRBIT GENERATOR BEING USED  H *3*J3  K	## B 00017  #################################	GRBIT GENERATOR BEING USED  H *3*J3  K R**4*J4  HST ORBIT GENERATOR  K2 *2/2  *3/2  *4/8	
######################################	######################################	#3#J3  #3#4#J4  HST ORBIT GENERATOR  KA  *34/8  *44/8	•
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H *3*J3 K R**4*J4 HST DRBIT GENERATOR B 00016 *2/2 *2/2 *3/2 K4 **4/8	H *3*J3 K R**4*J4 HST ORBIT GENERATOR B 00016 K2 *2/2 *3/2 *4/8 **4/8	H #3*J3 K R**4*J4 HST ORBIT GENERATOR B 00016 #2/2 #3/2 K3 *3/2 K4 **4/8	
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H *3*J3 K R**4*J4 HST DRBIT GENERATOR B 00016 K2 *2/2 *3/2 K4 **4/8	H *3*J3 K R**4*J4 HST ORBIT GENERATOR B 00016 *2/2 *3/2 *4/8	H *3*J3 KR R**4*J4 HST ORBIT GENERATOR B 00016 *2/2 *3/2 *4/8	
H *3*J3  K **4*J4  HST DRBIT GENERATOR B 00016  K2 *2/2  K3 *3/2  K4  **4/8	H *3*J3 K R**4*J4 HST ORBIT GENERATOR B 00016 K2 *2/2 *2/2 K4 **4/8	H *3*J3 K R**4*J4 HST ORBIT GENERATOR B 00016 K2 *2/2 *3/2 K4 **4/8	
H *3*J3  K ***4*J4  HST DRBIT GENERATOR B 00016  K2  *2/2  *3/2  K3  *4/8	H *3*J3 K K**4*J4 HST ORBIT GENERATOR B 00016 K2 *2/2 *2/2 *4/8 **4/8	H #3*J3 K R**4*J4 HST ORBIT GENERATOR B 00016 K2 *2/2 *2/2 *4/8 **4/8	
#3*J3 K R**4*J4 HST ORBIT GENERATOR B 00016 K2 *2/2 *2/2 *3/2 K4 **4/8	#3*J3 K R**4*J4 HST ORBIT GENERATOR K2 *2/2 *3/2 *44/8	#3*J3 K R**4*J4 HST ORBIT GENERATOR B 00016 K2 *2/2 *3/2 K4 **4/8	_
K R**4*J4 HST ORBIT GENERATOR B 00016 K2 *2/2 *3/2 *3/2 *4/8	KR3	Katataja HST ORBIT GENERATOR B 00016 K2 *2/2 *3/2 *4/8	- <del></del>
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×44/8	**4/8	**4/8	-
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											The second secon																													PAGE 166
	TERMINATION CONSTANTS	FOR MAG. OF (R X U) I	TOL. FOR (UNIT R) DOT (U) IN SUN DET.	2**26	2 pI	/ C.U.T.		CRITICAL INCL. FOR BROUWER	. n.r.	CANONICAL UNIT OF LENGTH)	DEGREES/RADIAN	ON OF EARTH 1	RADIUS OF EARTH IN C.U.L.		2	MU**2=GM	32	13	74	15		SMALL H SUB 0	(4/3) (BETA)	(4/3)(H SUB 0)	SEC/DAY	KE/MI		1+N SUB 0	SEC/HR		SUNLIGHT PRESSURE IN DYNES / CM**2	TRONOMICAL UNIT	OF SUN) / (MASS OF EA	(MASS OF EARTH)/(MASS OF MOON)		TION OF TAU IN RAD./	3 DEG. 26 MIN. 34.795 SEC.	* VELOCITY OF LIGHT IN KM. /	X COMPONENT OF UI VECTOR FOR SUN DET.	LINE 06640
K 00000	03825	V 00000 +00000000+00	10000	V 00016 +67108864+08	V 00017 +62831853+01	V 00018 +80683200+03		V 00019 +10000000-01	V 00021 +63783880+04		V 00022 +57295780+02	V 00023 +72921159-04	V 00024 +10000000+01	V 00025 +29700000+03	V 00026 +10000000+01	V 00064 +10000000+01	00028 +	- 62000	V 00030 -21230000-05	V 00031 -2320000-06	00033 +		V 00035 +26666667+00	96000		00038		00046	00048	00051	00052	00053	000054		00056	00065	V 00056 +40915752+00	V 00067 +29979250+06	V 00068 +10000000+01	K = 03825

PAGE 167	06641 06642 06643	77990	06645	06647	06648	06649	06 990	15990	26990	06653	06654	06655	99990	75990	06658	69990	09990	19990	79990	06663	99990	06665	99990	19990	06668	69990	04670	1,290	06672	000 (3	9299	06675	06676	11 990	92 990	0.000	08 9 90	
LINE 06641	Y COMPONENT OF UI VECTOR FOR SUN DET. X COMPONENT OF UI VECTOR FOR SUN DET.	מחוו חודתו מו מד דבותו ימי ממו									•										e de la companya de la major de servicio de la companya de la companya de la companya de la companya de la comp						A complete the second of the s											
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				0.1	<b>L</b>		TO C.U.T. F		DATA FUNCTION (USES 100 LOCS.)	ID. NO., (Z+1)	JAN. 1-	IN CUT, (X+	N(3,Q) I.E., MITH		DRAGS STARTING IN LUC. 60					SNSNNSNNSNSNNSNS																			
e e			- 1	INDICATOR	CNVERTER F.	TE TO J.D. F.	DAYS-SECONDS	NO. OF T(P,Q)'S - 1	AD DRAG	=(Z)		EXIT WITH (X)=T(P,0)	=(0++X) *** (	STARTI	+ TIMES OF					201020201050109030903																			
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PAGE 171	4		·		w.	ţ-	**	<b></b>	10	OI	B 00001				N 00008					80000 N	1												B 00013			
LINE 06801	IONS (ADDITIONAL EQUATIONS OF CONDITION)			)	AUGMENT MATRIX	O COR BETOMBREE LO CA	TEXAL LUC. UT JURNUM MUTA	54 TO	N. NO. DE CONDITION EONS.	+ DF SQUARED (0-C) 'S	CONSTRAINT EQUATIONS	ES AN	DWN BEING SOLVED FOR, AND	AUGMENTS THE NORMAL MATRIX BY EACH EQN. !								U. NO. DE UNKNOWN		-	USE J TO GET SUM OF XJ				حاسبات ميت بيستون ويستون والمراوية و	ET CE LIT O			HIS EQC IS NOT	ARE RT. HAND SIDE OF EQC	AND ADD IT TO SUM OF SQUARED (O-C)'S	
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LINE 06881	ACCORDING TO J, THE NO. OF THE UNKNOWN)	LOC. DF	FIRST LCC. OF UNKNOWNS, XJ	RECEDING (	· SUP		INT WEIGHTS	UNKNOWNS. USES J (NO.DF UNKNOWN X	SUM OF PREV.XJ, ADDS XJ TO IT +	ALSO	REV. SUMS	C, UNKNEWN NO.	NAC NAC NAC NAC NAC NAC NAC NAC NAC NAC	CET CORREC CONSTRAINT WIT	CORRES. SUM OF ALI	.) (PREVICUS SUM OF )		NEW SUM=(PREV.SUM) + (XJ)	SUM CUN	(NEW YORK OF XC)	200	GET ANOTHER J										والمتعارض والمتع					
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(DELTA M)		D. OF T(P,Q)'S	(0), EPOCH TIME IN C.U.T.	FOR DELTA M	T(P,Q)'S TIMES OF DRAGS IN C.U.T.	(2,0)'S		OBS. TIME IN C.U.T.	RINCIPAL	ABSOLUTE VALUE								FINAL DELTA M= (DELTA M)(K) WHERE K=1 OR					T(0)-T(N)		1(0)-1(0)		N2O	(1-10) * # 2				N3Q	(1-10)**3			11NF 07120
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02196		SQUARE ROOT	
02101		SINE	F 07326
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02271		PRINCIPAL VALUE	F 07334
10110		-	1 07335
01113		MEAN ANDMALY AT T (C	I 07336
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03844		ပ	I 07341
00276			1 07342
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00215		ARG. OF PERIGEE AT REQUEST TIME	
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01841		COMPUTE DRAG AND COMP. PERT. EFFECTS	F 07356
00018 03840		METERS/CJU.L.	I 07357
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03843		SECONDS/C.W.T.	1 07359
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V 00051 +1260000+03		07392
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K = 04500 LINE (7600	PAGE 190

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PAGE 193	07681	07683	07684	20010	0010	0.76.88	01688	07690	16420	07692	07693	<b>36920</b>	07695	96910	07697	07698	07699	00770	10110	20110	50770	#0.20	20110	20110	07708	90110	01710	07711	07712	07713	07714	07715	07716	07717	07718	61//0	07750	PAGE 193
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C2	C3					C4	01	02	03	04		07			614			04(385-18981)		03(37-270)					1/3C4(24-85+03(87-270)+04(385-18981))					04(4581+36082+385)				03(90-19282-87)					
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70004 00232 COS 6			07826
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00171			07828
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00177			07840
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PAGE 197					/									A CONTRACTOR OF THE PROPERTY O		B 00337																		1				PAGE 197
LINE 07841				H-PRIME				CCS 26					518 5G	- 1	- VI - U			EC	ш	SIN		- (	(1-EC COS(E))=R/A	A/R	C++(X/F)	(オノス)***3	u XIV		CCS F		26-PRIME	26+F	2G+2F	26+3F	SIN(2G+F)	SIN(2G+3F)	CCS(26+2F)	1 TNE 07880
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	07921	07922	07923	07924	07925	07926	07927	07928	07929	07930	07931	07932	07933	07934	07935	07936	07937	07938	07939	07940	07941	07942	07943	07944	07945	01940	01947	04.040	07949	07970	16670	26610	0795	4000	07955	07956	07957	07958	07959	04620	
rage 199					,																																				PAGE 199
LINE OFFI																				9			<b></b>					THE C	m		NIS.			(1-ECCOSE)=R/A	A/R	<b>~</b>	EC**2		(1-EC**2)**1/2		LINE 07960
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LINE USUUI			Z D0T	ł						TEST FOR VIND OF OUTBILL ELEN	NING DE COLLOR	OCT SINGLE TAINE						STORE OUT UNPRIMED (CSCULATING) ELEM.		: w			OMEGA	THETA (CAP CMEGA)		STURE OLT BOUBLE PRIME ELEM.	A-DOUBLE PRIME	E-DOUBLE PRIME	PRIME	1	PRIME=SMALL OMEGA-DOUBL	H-DOUBLE PRIME=CAP OMEGA-DOUBLE PRIME									04080 ENT
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ARC SIN ARC COS P1/2 P1/2 P1/2 P1/2 P1/2 P1/2 P1/2 P1/2	E-QUADRANT	NOIN ALV HITE TIVE		08041
ARC SIN ARC COS P1/2 P1/2 P1/2 P1/2 P1/2 2 P1 3 P1/2 2 P1 3 P1/2 2 P1 3 P1/2 2 P1 3 P1/2 2 P1 6 COS A 00003 00003 000010 00010 000110 000110 000110 000110 000110 A IV, CCS A=+ GR - 00014 00010 A IN I OR IV 00014 00014 A IN I OR IV 00014 A IN I OR IV 00014 A IN I OR II 00020 00014 A IN I OR II	44 7+T=CO2	EAIL WILL A=A, A BEIMEEN U -360		08044
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																								·															PAGE 204
	IONOSPHERIC REFRACTION						F, FLATTENING COEFFICIENT	SUB 2 = C.U.T./DAY		UB O	ALL	H SUB C		SMALL R VECTOR, X COMPONENT	R VECTOR, Z	ITUDE FUNCTION	S FOR LOW CI SUB S F	S FOR LAM CO/F2 FOR RAR DOT	WORK AREA FOR T SUB 0 NO. 1 OF 6 LOC.	SUB 1	SUB		SUB 4	AREA FOR T SUB 5	WORK AREA FOR N AUB 0 NO. 1 OF 6 LOC.	SUB 1	***	AREA FOR N SUB 5	WORK AREA FOR H SUB 0 NO. 1 OF 6 LOC.	SUB	4	OR H SUB 5	ECTED L. M. RHO OR RHO DOT	I C.U.T.		MPUT ED OBSERVATIONS		R, BAR STAR	LINE 08160
	640131 L,M,R,R, CORRECTOR FOR	- 1					834	837	858	871	859	361	980	204	206	110	346	847	848	849	350	351	352	353	854	855 1	358	359	360	361	864	865	660	200	751	256	207	375	K = 09000
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08161	08162	08163	08164	08165	08166	08167	08168	08169	08170	08171	08172	08173	08174	08175	08176	08177	081 78	08179	08180	08181	08182		08184	08185	08186	08187	08188	08189	08180			081 93			081 96			081 99	200	
																						B 00001									8 00034			B 00045			8 00067			PAGE 205
<b></b>	u.	u.	LL.	ш.	0	ш.	u.	<b>Junq</b>	IJ.	-									ZERO																PARAMETER					
TYPE CODE	OUT PRODUCT ENTRY	TOR MOVE	SIN ENTRY		-	1	ENTRY 10 EXPONENTIAL FUNCTION		10 SQ.	RAD			CDE FOR L	· CCD	, CODE FOR EL	E FOR RANGE R		NCE FOR ZERO EL	EDO PARAM	PHI SUB I	TEN DEGREES, BI				UNCORRECTED L,M, OR RANGE RATE	- 1		EXIT							CORRECTED PARAMETER IS ORIGINAL P					LINE 08200
00253	02051	02001	02101	02105	02.085	01201	01851	01202	02196	03849	+000000000+	+1000000001+	+2000000401	+30000000+01	+50000000001	+99000000001	+13000000+02	+10000000-12	*100c0000*I0	+99922944+00	+17453293+00		9-1041	0.5631 00.803	0000 k	90830	00000	\$2000 \$2000	03053	00054		00010 00029 00029						- 1	10004 0000	# # # # # # # # # # # # # # # # # # #
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RESTORE EXIT 1 RESTORE TYPE CODE	RESTORE EXIT 3					TO BOOK TO STOCK OF THE COLUMN TO STOCK OF TH	CCMPUTE H							A COLD FILE ONE	1 208 0=1 208 4-1 N 618 0-N 618 A	SUB 0-1 500	COB 5-1-1	5=N SUR 1	SUB 5=H SUB										T CI18 1	1			-	SUB I	Û		11NF 08240
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94000	(	00044			08247
00018		000044			08248
81000		00037			08249
90000	i	80031	COMPUTE CAP H		08250
000019		00044			08251
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00019	90006	61000			08253
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			**				IS EL GREATER THAN TEN DEGREES			( )	CHANGE EXIL I IC EXIL ZA		DOES EL APPROACH ZERO			#										EXIT I TO EXIT 2	EXII	l	DOES EL APPROACH ZERO		EXIT 2A							EXIT 3			LINE 08320	1
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B 00054												ORDOO a			8 00012		ł	B 00078			02000	1																PAGE 209
EXIT 3A																						-1				CUC DHI CHR 2	2					CSC PHI SUB 1		CSC PHI SUB 2	100	1ST TEDM OF DELTA AND	IEKM UF UELIA	LINE 08360
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	08361	79580	0620	08364	08365	08366	19880	08368	08369	08370	08371	08372	08373	08374	08375	08376	08377	08378	08379	08380	08381	08382	08383	08384	08385	08386	08387	08388	08389	08390	1680	26880	08393	08394	08395	08396	16880	08398	08399	08400	
PAGE ZIO			0000 9																																						PAGE 210
LINE 08361			- 1		SIN EL.EL DOT							2ND TERM OF DELTA RHO DOT					1ST TERM OF DELTA RHD DOT		2 DH I-2	3RD TERM OF DELTA RHO DOT													مستها والمتباه والمتباه والمتباء والمتباء والمتباء المتباء المتباء المتباء والمتباء								LINE C8400
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			B 00032		8 00033	1									B 00035													B 00034													PAGE 212
TS BO 0.1-RO G.M GREATER THAN TEST	TO OUT THE THE T	Di Turin Du	DO O. 1-DO C.M GREATER THAN TEST	1 DOTHE LEDO 1	TRIME SOLD	מייי	A+KU I FKIER			STORE RO O FOR TAPE RECORD		T HU	c	u	3	C T C C C C C C C C C C C C C C C C C C		•	HINE (FECKED	,	+DA	SET REF. YEAR	2	XX YOU	1	O STITUTE OF THE CONTRACT OF T	202	- 1		SECS + SECZUAT	J.DUNE										LINE 08480
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K =	PAG	E LINE	NAME
00100	017	00643	PRINT REQUESTED QUANTITIES
00150	020	00763	(O-C) PRINT FUNCTION FOR D.C.
01200	022	00843	LSP (LOCAL STATION PREDICTIONS)
01300	027	01044	PE (POSITION IN ELLIPSE)
01355	031	01203	COMPUTE AND STORE CORRECTED ELEMENTS
01380	035	01363	(O-C) ANGLE REDUCER
01400	036	01403	PP (POSITION PARTIALS)
01600	062	02443	OP (OBSERVATION PARTIALS)
01700	071	02803	CC (CONVERT CORRECTIONS)
01840	080	03163	COMPUTE DRAG AND COMP. PERT. EFFECTS
01850	081	03203	EXPONENTIAL FUNCTION
01900	083	03283	SATELLITE IDENTIFICATION LOAD FUNCTION
01940	084	03323	SATELLITE IDENTIFICATION LOAD AND PRINT
01955	085	03363	ADD TO CORRES. SUM OF (O-C) SQ AND TO N FOR ONE OBS.
01970	086	03403	HOURS-MINUTES-SECONDS TO RADIANS
01985	087	03443	DEGREES-MINUTES-SECONDS TO RADIANS
02000	088	03483	VECTOR PACKAGE (VECTOR MOVE)
02010	089	03522	VECTOR PACKAGE (VECTOR MAGNITUDE)
02020	090	03562	VECTOR PACKAGE (VECTOR DIRECTION)
02030	091	03602	VECTOR PACKAGE (VECTOR ADD)
02040	092	03642	VECTOR PACKAGE (VECTOR SUBTRACT)
02050	093	03682	VECTOR PACKAGE (DOT PRODUCT)
02060	094	03722	VECTOR PACKAGE (CROSS PRODUCT)
02075	095	03762	VECTOR PACKAGE (SCALAR BY VECTOR PRODUCT)
02100	096	03803	SINE_COSINE FUNCTION
02155	098	03882	ARC SINE
02165	099	03922	ARC COSINE
02175	100	03962	TANGENT
02195	101	04002	SQUARE ROOT FUNCTION
02215	102	04046	KEPLER
02245	104	04122	ARC TAN
02270	106	04203	ANGLE REDUCTION FUNCTION
02300	107	04243	MATRIX CLEAR
02315	108	04282	FULL ROW MATRIX FUNCTION
02340	109	04323	COMPUTE AND PRINT R.M.S. FOR EACH OBSERVATION TYPE
02360	111	04403	LINEAR EQUATIONS SOLUTION FUNCTION
02417	114	04523	SPO (STATION POSITION ORIENTATION)
02475	118	04683	LOAD AND STORE PRINT REQUEST CARDS
02520	120	04763	OUTPUT SCALE
02650	124	04922	INPUT CONVERTER

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02720	125	04963	RANGE RATE FUNCTION
02750	126	05002	ABSOLUTE VALUE FUNCTION
02760	127	05043	ONE WORD LOAD
02800	128	05083	OBSERVATION LOAD FUNCTION
02935	131	05203	ELEMENT LOAD (CONVERSION OF ELEMENTS)
03050	142	05643	BACKWARD DIFFERENCE INTERPOLATION FUNCTION
03300	144	05723	DAY COUNT FUNCTION
03350	146	05083	OBSERVED DATE TO JULIAN DAYS FUNCTION
03375	147	05843	REDUCTED JULIAN DAYS-SECONDS TO CUT
03400	148	05884	PROGRAM P FOR COMPLEMENTARY PERTURBATIONS
03495	152	06043	INITIALISE PROGRAM P FOR COMP. PERT. TAPE
03500	153	06083	LOAD REJECT CARDS
03520	154	06123	OBSERVATION SEARCH AND REJECT FUNCTION
03550	155	06163	PRINT INITIAL ELEMENTS, DRAGS, AND EARTH CONSTANTS
03600	1.59	06323	LOAD AREA, MASS, DRAG DATA CARD FOR MCOI
03700	160	06366	CTMU ASSIGNMENT FUNCTION
03750	162	06442	CTMU COMPLEMENT
03800	163	06483	CONSTANTS POOL FUNCTION FOR DIFF. CORRECTION
03825	166	06602	SHADOW DETERMINATION CONSTANTS
03900	168	06683	LOAD DRAG DATA
04025	170	06763	LOAD CONSTRAIN WEIGHTS
04055	171	06803	COMPUTE CONSTRAINT EQUATIONS
04100	173	06883	SUM UNKNOWNS
04200	174	06923	RUN IDENTIFICATION LOAD AND PRINT
04330	175	06966	INTERVAL CORE DUMP
04400	178	07084	COMPUTE EFFECTIVE DRAG (DELTA M)
04455	180	07166	INTERVAL CORE DUMP DUMP
04490	182	07246	INITIALISE BROUWER ORBIT GENERATOR
04500	184	07322	BROUWER ORBIT GENERATOR
04946	202	08046	ANGLE-QUADRANT DETERMINATION
09000	204	08128	CORRECTOR FOR IONOSPHERIC REFRACTION
09100	211	08403	RANGE AMBIGUITY RESOLUTION FUNCTION
09200	213	08483	JULIAN DAYS-SECONDS TO C.U.T.

## References

- C1. Gorman, T. P., CAMEO System Description, GSFC Report X-542-64-148.
- C2. Maury, J. L., Programming in MYSTIC: A Primer on the use of CAMEO, GSFC Report X-542-64-393.